

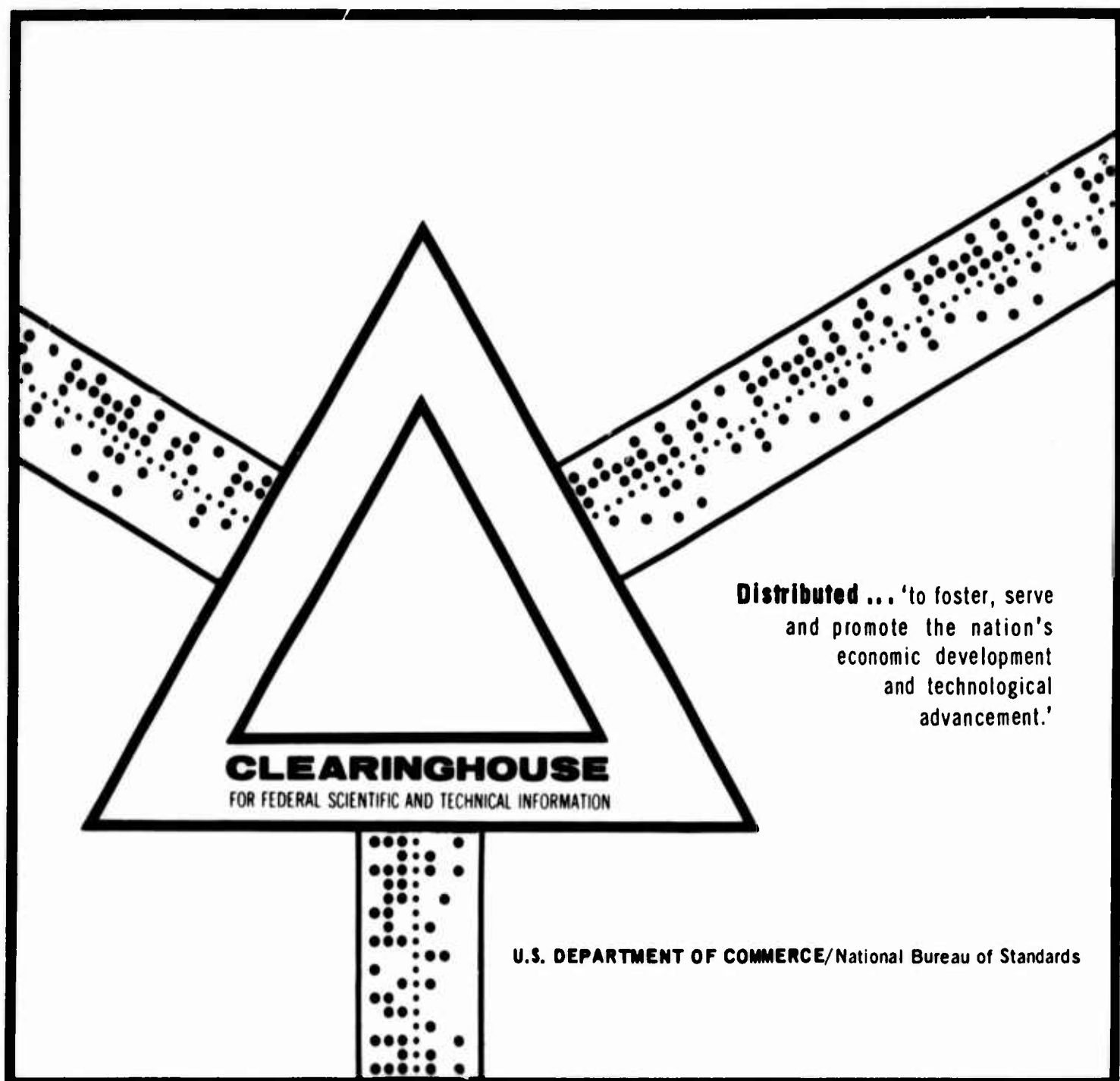
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ANALYSIS OF TRANSVERSELY ISOTROPIC LAMINATED
CYLINDERS UNDER AXISYMMETRIC MECHANICAL AND
THERMAL LOADS

Jonas A. Zukas

Ballistic Research Laboratories
Aberdeen Proving Ground, Maryland

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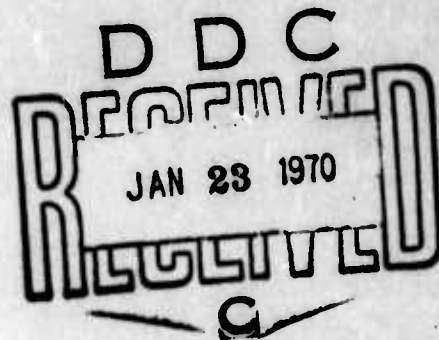
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BALLISTIC RESEARCH LABORATORIES
ABERDEEN PROVING GROUND, MARYLAND**

BALLISTIC RESEARCH LABORATORIES

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ABSTRACT

A theory for the analysis of stresses in laminated circular cylindrical shells subjected to arbitrary axisymmetric mechanical and thermal loadings has been developed. This theory, specifically for use with pyrolytic graphite type materials, differs from the classical thin shell theory in that it includes the effects of transverse shear deformation and transverse isotropy, as well as thermal expansion through the shell thickness.

Solutions in several forms are developed for the governing equations. The form taken by the solution function is governed by geometric considerations. A range in which the various solution forms occur was determined numerically.

As a sample problem, the slow cooling of pyrolytic graphite deposited onto a commercial graphite mandrel was considered. Investigation of normal and shear stress behavior at the pyrolytic graphite - mandrel interface showed that these stresses decrease in magnitude with increasing E/G_c ratio and increasing deposit to mandrel thickness (h_a/h_b) ratio. This implies that a thin mandrel and a material weak in shear are desirable to minimize the possibilities of flaking and delamination of the pyrolytic graphite.

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NOTATION

a, b	subscripts indicating upper or lower lamina
a_{ij}, b_{ij}	constants defined by equations (C.2) and (C.3)
A, B	constants defined by (C.8)
c	preferred direction in a transversely isotropic material
c_i	roots of equation (22) defined by (C.11) and (C.13)
C_i	$\frac{E_i h_i}{1 - \nu_i^2} \quad (i = a, b)$
D_i	$\frac{E_i h_i^3}{12(1 - \nu_i^2)}$
D	d/dx
e	natural base
E, E_c	Young's Modulus in the plane of isotropy and "c" direction respectively
G_c	shear modulus relating stress and strain across the plane of isotropy ($=G_{xz} = G_{\theta z} = G_{zx} = G_{z\theta}$)
g_i	constants defined by equation (C.4)
h_a, h_b	individual lamina thicknesses
h	$h_a + h_b$
i, j	subscripts
k_i	constants defined by equation (C.6)
m, n	constants defined by equation (C.7)
\bar{m}	constant defined by (11a)
M_x, M_θ	stress couples

$M_{Tx}, M_{T\theta}$	thermal couples
\bar{M}	defined by equation (41)
N_x, N_θ	stress resultants
$N_{Tx}, N_{T\theta}$	thermal resultants defined by equation (41)
P_{1i}	$\sigma_z (h_i/2) \quad (i = a, b)$
P_{2i}	$\sigma_z (-h_i/2) \quad (i = a, b)$
$p(x)$	$P_{1a} - P_{2b}$
P_j	joint normal stress $(=\sigma_z (-\frac{h_a}{2}) = \sigma_z (+\frac{h_b}{2}))$
Q_i	shear resultant
\bar{Q}	defined by equation (41)
R	Radius to shell reference surface
T	temperature measured from the stress-free temperature of the material
u_x, u_z	deflections in the "x" and "z" directions respectively
u_{0i}	axial deflection of lamina middle surface
u_i^*	virtual displacement of shell middle surface
w_i	radial displacement of lamina middle surface
\bar{w}_i	$\int_0^z (\alpha_c T)_i dz \quad (i = a, b)$
w_i^*	virtual radial displacement of lamina middle surface.
x	axial coordinate for cylindrical shell
z	cylindrical coordinate
α, α_c	thermal expansion coefficients in the "x" and "z" directions respectively

β_i	rotation of the normal to the undeformed lamina middle surface due to deformation
Γ	defined by equation (C.10)
$\lambda_1, \lambda_2, \lambda_3$	defined by equation (C.9)
β_i^*	virtual rotation
Δ_i	defined by equation (21)
ϵ_{ij}	strain component
ν	Poisson's ratio in the $x - \theta$ plane ($\nu_{x\theta} = \nu_{\theta x}$)
ν_c	Poisson's ratio ($\nu_{xz} = \nu_{\theta z}$)
ν_{ij}	Poisson's ratio defined as the negative of the ratio of the strain in the j -direction to the strain in the i -direction due to a stress in the i -direction
σ_{ij}	stress component
τ_j	joint shear stress ($= \sigma_{zx} (-h_a/2) = \sigma_{zx} (+h_b/2)$)
θ	cylindrical coordinate in the circumferential direction
θ_1, θ_2	defined by equation (C.10)
$()'$	$\frac{d}{dx} ()$

I. INTRODUCTION

The ever-expanding missile and space technology continually demands materials capable of maintaining structural integrity at very high temperatures. Of late, attention has been focused on refractory materials, their anisotropy in physical and mechanical properties making them ideally suited for a wide range of insulation and/or structural applications.

Of the many refractory materials possible, pyrolytic graphite (PG) has probably received the most attention of late although it was known to Edison (1)* in 1883 who described methods for its manufacture, the technique involving formation of carbon deposits onto substrates heated in carbon-containing gases. For structural use, pyrolytic graphite is generally deposited at temperature from 3500°F to 4000°F in a stream of hydrocarbon gas, such as methane, onto a substrate of commercial graphite maintained at temperatures of 1500°F to 5000°F. The rate at which the material is produced depends on a number of factors which include the temperature, the reaction pressure, the hydrocarbon flow rate and

*Numbers in parenthesis indicate corresponding references in the bibliography

the surface to volume ratio of the substrate surface (2), (3), (4), (5). X-ray analysis of the resultant deposit shows a well-crystallized structure having much in common with the single graphite crystal (6). Growth is always normal to the substrate surface and after a thickness of 0.1" - 0.5" is reached, the deposition process is stopped and the deposit allowed to cool for several weeks.

The result of such a formation process is a material highly anisotropic in physical properties. The PG has one plane of isotropy parallel to the mandrel surface (x, θ direction -- see figure II-1 for geometry) and a single preferred direction (z), a state commonly referred to as transverse isotropy. With thermal conductivity between 100 and 1000 times greater in the (x, θ) direction than in the (z) direction, the material acts as an excellent conductor along its surface but also as a good insulator in the thickness (z) direction. The coefficient of thermal expansion in the (z) direction is from 10 to 30 times greater than that in the plane of isotropy (x, θ) so that thermal expansion through-the-thickness must be considered in many analyses of the material's thermal behavior.

Other curious effects due to the anisotropy are manifest in the Poisson's ratio which is negative in the plane of isotropy ($\nu_{x\theta} = \nu_{\theta x} = -0.21$) but large and positive in the preferred direction

($\nu_{xz} = \nu_{\theta z} = +0.9$). Furthermore, the ratio of the elastic modulus in the isotropic plane to the shear modulus in the transverse plane (E_x/G_{xz} or $E_\theta/G_{\theta z}$) may range from 20 to 50, compared to an E/G ratio of 2.5 for an isotropic material with $\nu = 0.25$. Therefore, in an analysis of a structure composed of such material, transverse shear deformation even for thin cross-sections must be considered.

Thermal and mechanical properties can be found readily (2), (5), (7), (8), (9), (10), (11), (12). Typical properties, given in Table I-1, are taken from (10), which are reasonably close to those given in other references, discrepancies most probably being due to variations in the deposition process.

Among the earliest analyses of structures of pyrolytic materials were those of Garber (13) and Levy (14) who treated thermal stresses in cylindrical and spherical shells and also the residual stresses caused by the general anisotropy of pyrolytic graphite, but neglected transverse shear deformation and did not account for the high thermal expansion coefficient in the (z) direction. McDonough (15) has considered thermal stresses in shells of revolution of pyrolytic graphite type materials subjected to axially symmetric loads, including transverse shear deformation and thermal expansion through the thickness. He was able to show that neglect of transverse shear deformation would lead to an over-estimate of the stiffness coefficient

TABLE I-1 PHYSICAL PROPERTIES OF PYROLYTIC GRAPHITE

MECHANICAL PROPERTIES

1. Young's Modulus (PSI)

<u>TEMP</u>	<u>(x,θ) Direction</u>	<u>(z) Direction</u>
70°F	5.4×10^6	1.5×10^6
1000°F	4.3×10^6	1.29×10^6
2000°F	3.5×10^6	1.05×10^6
3000°F	2.7×10^6	0.81×10^6

2. Poisson's Ratio

70°F	$\nu_{x\theta} = 0.21$	$\nu_{xz} = 0.90$
------	------------------------	-------------------

THERMAL PROPERTIES

1. Thermal Expansion in/in - °F

70°F	0.0	13.1×10^{-6}
1000°F	0.6×10^{-6}	13.1×10^{-6}
2000°F	1.2×10^{-6}	13.1×10^{-6}
3000°F	1.7×10^{-6}	13.1×10^{-6}

2. Conductivity, BTU/hr-ft-°F

70°F	290.0	1.25
1000°F	165.0	0.82
2000°F	100.0	0.60
3000°F	60.0	0.60

for properties representative of PG while excluding thermal expansion in the (z) direction leads not only to erroneous stress predictions but that even the sign of the stress (tension or compression) may be wrong. Kliger (16), (17) has extended McDonough's work for the case of conical shells in that he derives equations for non-axially symmetric mechanical and thermal loadings. Raju (18) studied the case of shallow shells of pyrolytic graphite type materials subjected to a variety of axially symmetric and non-axially symmetric loads. Daugherty (19), (20) treated the case of non-circular cylindrical shells of pyrolytic materials.

The preceding deal with single-layer shells. Anisotropic laminated shells of revolution with elastic properties symmetric about the middle surface of the composite shell are extensively treated by Ambartsumian (21), whereas Dong (22), (23), et al (24) treat layered shells wherein the structure is assumed to be composed of an arbitrary numbers of bonded layers each of different constant thickness, different orientation of elastic axes and different anisotropic elastic properties. Since Dong does not assume elastic symmetry about the middle surface, flexural and extensional deformations are coupled and solution techniques for homogeneous shells do not carry over directly for anisotropic elastic shells. Hence, alternate methods of solution are developed.

Radkowski et al (25) considered laminated isotropic shells of revolution with variable thickness using E. Reissner's for-

formulation (26). Radkowski extended this work to include variable laminated orthotropic material properties (27). Both formulations were restricted to axisymmetric loads. In Radkowski's works and that of Sepetoski (28), the governing equations were cast in finite difference form and solved with the aid of a digital computer. The introduction to Dong's paper (23) makes interesting reading regarding the hazards of this perfectly valid technique.

Other treatments of laminated cylinders have been by Jones and Whittier (29), Tsai and Azzi (30), Paul (31), Au (32), Keeffe and Windholz (33). These, and most other references cited herein are characterized by neglect of transverse shear deformation. A recent work of great theoretical elegance, even though it neglects transverse shear deformation, is that of Zudans (34) which presents a theory for arbitrarily loaded (mechanically & thermally) shells of revolution with internal masses and ring stiffeners, derived under the Kirchhoff Hypothesis and consistent with balance of energy as well as linear and angular momentum and invariance under transformation of middle surface coordinate systems and rigid body displacements. The elegance, unfortunately, does not carry over to the computational techniques (35).

Laminated isotropic plates have been considered by Vinson (36) who treated thermal stresses in circular plates, neglecting transverse shear deformation. Summers (37) treats thick and thin isotropic and orthotropic laminated plates including transverse shear deformation.

Mehta (34) considers orthotropic and isotropic laminated as well as single-layered rectangular plates of pyrolytic graphite type materials under static mechanical and thermal loads. Wu (39), (40), and (41) treats the lateral vibrations of both small and large amplitude for rectangular plates of pyrolytic and graphite materials.

Except for those references dealing with pyrolytic graphite type materials, all the works reviewed which analyze multi-layered structures are characterized by their neglect of either transverse shear deformation or thermal expansion through-the-thickness or both.

Prompted by the absence of a definitive treatment of laminated shells applicable to pyrolytic graphite type materials, this thesis was undertaken. It is an extension of McDonough's work in that layered cylindrical shells, including the effects of transverse shear deformation and thermal expansion through-the-thickness, subjected to arbitrary axi-symmetric loading are considered.

II. DERIVATION OF GOVERNING EQUATIONS

The coordinate system used is shown in Figure (II-1). The three-dimensional equations of thermoelasticity (uncoupled) for the case of axial symmetry are given by:

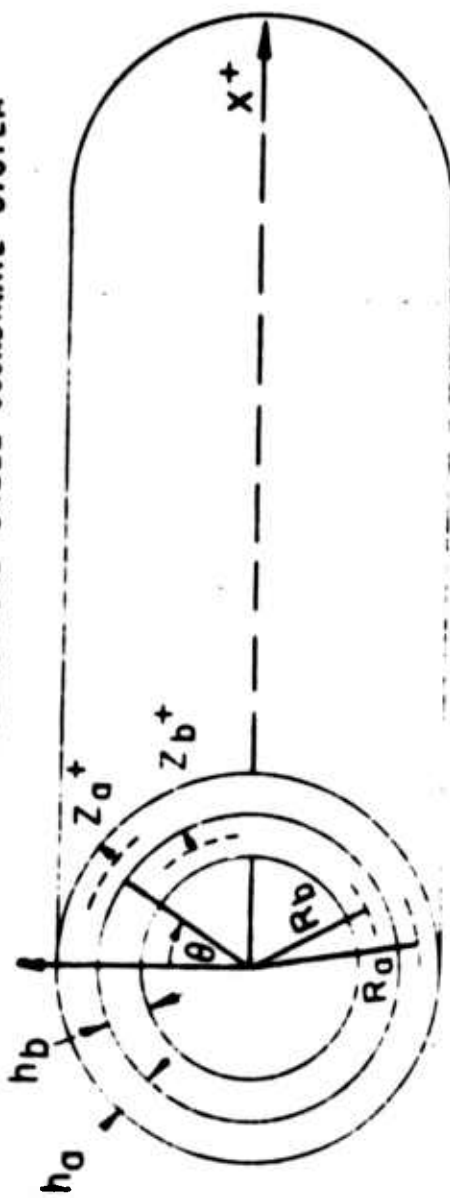
Stress-Strain Relations (transversely isotropic material)

$$\begin{aligned}\epsilon_x &= \frac{1}{E} (\sigma_x - \nu \sigma_\theta - \nu_c \sigma_z) + \alpha T \\ \epsilon_\theta &= \frac{1}{E} (\sigma_\theta - \nu \sigma_x - \nu_c \sigma_z) + \alpha T \\ \epsilon_z &= \frac{\sigma_z}{E_c} - \frac{\nu_c}{E} (\sigma_x + \sigma_\theta) + \alpha_c T \\ \epsilon_{xz} &= \frac{\sigma_{xz}}{2G_c}\end{aligned}\tag{1}$$

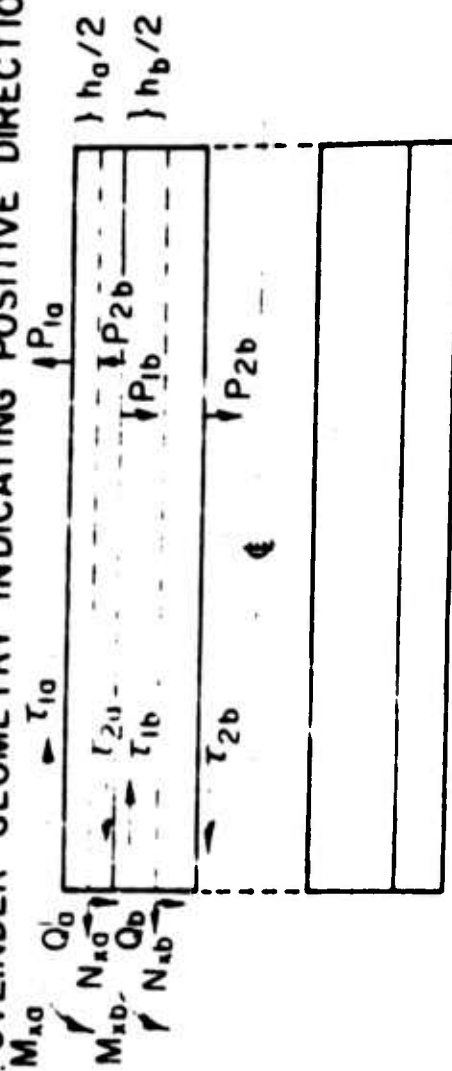
where $\epsilon_{x\theta} = \epsilon_{\theta z} = \sigma_{x\theta} = \sigma_{\theta z} = 0$ by symmetry

ϵ_{ij} and σ_{ij} are the physical components of the strain and stress tensors respectively, and for brevity $\sigma_{ij} = \sigma_i$ when $i = j$. E, E_c, ν, ν_c, G_c are five independent elastic constants where again for brevity $\nu = \nu_{x\theta} = \nu_{\theta x}$ and $\nu_c = \nu_{xz} = \nu_{\theta z}$. α, α_c are the coefficients of thermal expansion in the x and z directions of the materials. T is the temperature measured from the stress-free temperature of the material in units consistent with the α 's.

FIGURE II-1: LAYERED CYLINDRICAL SHELL COORDINATE SYSTEM



1A. CYLINDER GEOMETRY INDICATING POSITIVE DIRECTIONS



$$P(x) = P_{1o} - P_{2o}; \quad T_j = T_{2o} = T_{1i}; \quad P_j = P_{2o} = P_{1i}$$

1B. FORCES ON LAYERED CYLINDER

The preferred direction for the material is everywhere coincident with the z coordinate. This restriction is carried throughout this work.

Equilibrium Equations

The equilibrium equations when applied to the present problem become for each lamina:

$$R(1 + \frac{z}{R}) \frac{\partial \sigma_x}{\partial x} + R(1 + \frac{z}{R}) \frac{\partial \sigma_{xz}}{\partial z} + \sigma_{xz} = 0 \quad (2)$$

$$R(1 + \frac{z}{R}) \frac{\partial \sigma_z}{\partial z} + R(1 + \frac{z}{R}) \frac{\partial \sigma_{xz}}{\partial x} + \sigma_z - \sigma_\theta = 0$$

The third equation is identically zero from symmetry considerations.

Strain-Deformation Equations

The strain-deformation relations can be written correspondingly as:

$$\epsilon_x = \frac{\partial u_x}{\partial x}$$

$$\epsilon_\theta = \frac{1}{R(1 + \frac{z}{R})} u_z$$

$$\epsilon_z = \frac{\partial u_z}{\partial z} \quad (3)$$

$$\epsilon_{xz} = \frac{1}{2} \left\{ \frac{\partial u_x}{\partial z} + \frac{\partial u_z}{\partial x} \right\}$$

$$u_\theta \equiv 0 \text{ by symmetry}$$

The displacements are positive in the direction of the positive corresponding coordinate (See Figure II-1).

Assumptions:

1. The thickness of the shell is small compared with other dimensions, hence Love's First Approximation is applicable:

$$\text{i.e. } \frac{h}{R_{\min}} \ll 1 \quad (4)$$

2. The displacements are small compared to the thickness of the shell and the angles of rotation are small compared to unity.

3. The transverse normal stress is small compared with other normal stress components and is neglected in the stress-strain equations.

4. A linear element normal to the undeformed middle surface undergoes translation and rotation and remains straight, implying deformations of the form

$$u_x = u_0(x) + z\theta(x) \quad (5)$$

5. Transverse normal strain due to thermal expansion will be included; that due to mechanical (or isothermal) loads will be neglected.

This implies a deformation of the form

$$u_z(x,z) = w(x) + \bar{w}(x,z) \quad (6)$$

where

$$\bar{w}(x,z) = \int_0^z \alpha_c T dz$$

6. Material properties are constant for each lamina.

With these assumptions, the equation (1) becomes:

$$\begin{aligned}
 \epsilon_x &= \frac{1}{E} (\sigma_x - \nu \sigma_\theta) + \alpha T \\
 \epsilon_\theta &= \frac{1}{E} (\sigma_\theta - \nu \sigma_x) + \alpha T, \\
 \epsilon_z &= \alpha_c T \\
 \epsilon_{xz} &= \frac{\sigma_{xz}}{2G_c}
 \end{aligned}
 \tag{7}$$

Making use of equations (5) and (6) and neglecting z/R in comparison to unity, the equations (3) become:

$$\begin{aligned}
 \epsilon_x &= u_0' + z\beta' \\
 \epsilon_\theta &= \frac{1}{R} (w + \bar{w}) \\
 \epsilon_z &= \frac{\partial \bar{w}}{\partial z} \\
 \epsilon_{xz} &= 1/2(\beta + w')
 \end{aligned}
 \tag{8}$$

where $()' = D () = \frac{d}{dx} ()$

Integrated Equations

The stress resultants and couples are defined as follows:

$$\begin{aligned}
 N_{x_i} &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} \sigma_x dz & N_{Tx_i} &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} E_i \alpha_i T dz \\
 N_{\theta_i} &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} \sigma_\theta dz & N_{T\theta_i} &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} E_i \alpha_i T dz \\
 Q_i &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} \sigma_{xz} dz & & \\
 M_{x_i} &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} z \sigma_x dz & M_{Tx_i} &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} z E_i \alpha_i T dz \\
 M_{\theta_i} &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} z \sigma_\theta dz & M_{T\theta_i} &= \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} z E_i \alpha_i T dz
 \end{aligned} \tag{9}$$

$$i = a, b$$

The equilibrium equations in terms of stress resultants and couples are readily obtained. Integrating (2) directly yields

$$N'_{x_i} + \tau_{1i} - \tau_{2i} = 0$$

$$M'_{x_i} - Q_i + \frac{h_i}{2} \tau_{1i} + \frac{h_i}{2} \tau_{2i} = 0$$

(10)

$$Q'_i - \frac{N_{\theta i}}{R_i} + P_{1i} - P_{2i} = 0$$

$$i = a, b$$

Solving equations (7) for normal stresses, first integrating them from $-h_i/2$ to $+h_i/2$; then multiplying through the equations by z and integrating once again between the same limits, making use of the definitions of the stress resultants and couples (9) and simplifying, the following stress-strain relations result:

$$N_{x_i} = \frac{E_i h_i}{(1-\nu_i^2)} \left(u_{oi}' + \frac{\nu_i w_i}{R_i} \right) - \frac{N_{Tx_i}}{1-\nu_i} + \frac{E_i \nu_i}{R_i (1-\nu_i^2)} \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} \bar{w}_i dz$$

$$N_{\theta_i} = \frac{E_i h_i}{(1-\nu_i^2)} \left(\nu_i u_{oi}' + \frac{w_i}{R_i} \right) - \frac{N_{T\theta_i}}{1-\nu_i} + \frac{E_i}{R_i (1-\nu_i^2)} \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} \bar{w}_i dz$$

$$M_{x_i} = \frac{E_i h_i^3}{12(1-\nu_i^2)} \beta_i' - \frac{M_{Tx_i}}{1-\nu_i} + \frac{E_i \nu_i}{R_i (1-\nu_i^2)} \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} z \bar{w}_i dz \quad (11)$$

$$M_{\theta_i} = \frac{E_i h_i^3}{12(1-\nu_i^2)} \beta_i' - \frac{M_{T\theta_i}}{1-\nu_i} + \frac{E_i}{R_i (1-\nu_i^2)} \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} z \bar{w}_i dz$$

$$i = a, b$$

An integrated shear stress strain equation is required in addition to (11). The necessary expression is obtained using weighted integration, the procedure being analogous to that in reference 15, and for convenience given in Appendix D.

$$Q_i = \frac{\bar{m}_i}{6} + \frac{5}{6} h_i G_c^i (\beta_i + w_i) + \Delta_i$$

$$\text{where } \bar{m}_i = \frac{h_i}{2} (\tau_{1i} + \tau_{2i}) \quad (11A)$$

Solving for Q_a and Q_b from the second of the integrated equilibrium equations (10) and substituting into the third, making use of M_{x_i} and N_{θ_i} from (11) and the definitions

$$C_i = \frac{E_i h_i}{(1-\nu_i^2)} ; \quad D_i = \frac{E_i h_i^3}{12(1-\nu_i^2)} \quad (i = a, b)$$

$$\tau_j - \sigma_{zx} \left(-\frac{h_a}{2} \right) = \sigma_{zx} \left(+\frac{h_b}{2} \right)$$

$$p_j = \sigma_z \left(-\frac{h_a}{2} \right) = \sigma_z \left(+\frac{h_b}{2} \right)$$

two useful relations are obtained:

$$D_a D^3 \beta_a - \frac{C_a \nu_a h_a}{2R_a} D\beta_a - \frac{C_a \nu_a h_b}{2R_a} D\beta_b + \frac{h_a}{2} D\tau_j - p_j - \frac{C_a \nu_a}{R_a} Du_{o_b} - \frac{C_a}{2R_a} w_b = \alpha_{17} + \pi_1 \quad (12)$$

$$D_b D^3 \beta_b + \frac{h_b}{2} D \tau_j + p_j - \frac{C_b v_b}{R_b} D u_{o_b} - \frac{C_b}{R_b^2} w_b = \alpha_{27} + \pi_2 \quad (13)$$

Note that in the above, use has been made of the conditions that the laminae are bonded together and that no slippage occurs in the joints between laminae. The former is given by

$$u_z\left(\frac{-h_a}{2}\right) = u_z\left(\frac{h_b}{2}\right) \text{ which implies that } \bar{w}_a\left(x, \frac{-h_a}{2}\right) + w_a = \bar{w}_b\left(x, \frac{h_b}{2}\right) + w_b$$

$$\text{or } w_a = w_b + \bar{w}_b\left(x, \frac{h_b}{2}\right) - \bar{w}_a\left(x, \frac{-h_a}{2}\right)$$

$$= w_b + \hat{E} \quad (14)$$

$$\text{where } \hat{E} = \bar{w}_b\left(x, \frac{h_b}{2}\right) - \bar{w}_a\left(x, \frac{-h_a}{2}\right).$$

The latter condition is expressed by

$$u_x\left(\frac{-h_a}{2}\right) = u_x\left(\frac{h_b}{2}\right)$$

$$u_{o_a} - \frac{h_a}{2} \beta_a = u_{o_b} + \frac{h_b}{2} \beta_b$$

$$u_{o_a} = u_{o_b} + \left(\frac{h_a}{2} \beta_a + \frac{h_b}{2} \beta_b\right) \quad (15)$$

Making use of (11), (14) and (15) in the first integrated equilibrium equation, two further relations are obtained:

$$\frac{C_a h_a}{2} D^2 \beta_a + \frac{C_a h_b}{2} D^2 \beta_b - \tau_j + C_a D^2 u_{ob} + \frac{C_a v_a}{R_a} D w_b = \alpha_{37} + \pi_3 \quad (16)$$

$$\tau_j + C_b D^2 u_{ob} + \frac{C_b v_b}{R_b} D w_b = \alpha_{47} + \pi_4 \quad (17)$$

Using (11A) and the second of the integrated equilibrium equations together with (14) and (15), two final relations are obtained:

$$D_a D^2 \beta_a - \frac{5}{6} G_c^a h_a \beta_a + \frac{5}{12} h_a \tau_j - \frac{5}{6} G_c^a h_a D w_b = \alpha_{57} + \pi_5 \quad (18)$$

$$D_b D^2 \beta_b - \frac{5}{6} G_c^b h_b \beta_b + \frac{5}{12} h_b \tau_j - \frac{5}{6} G_c^b h_b D w_b = \alpha_{67} + \pi_6 \quad (19)$$

where in the above

$$\alpha_{17} = \frac{N_{T\theta a}}{R_a(1-\nu_a)} - \frac{C_a \hat{E}}{R_a^2} - \frac{E_a}{R_a^2(1-\nu_a^2)} \int_{-\frac{h_a}{2}}^{\frac{h_a}{2}} \bar{w}_a dz$$

$$+ D^2 \left\{ \frac{M_{Tx a}}{1-\nu_a} - \frac{E_a \nu_a}{R_a(1-\nu_a^2)} \int_{-\frac{h_a}{2}}^{\frac{h_a}{2}} z \bar{w}_a dz \right\}$$

$$\alpha_{27} = \frac{N_{T\theta_b}}{R_b(1-\nu_b)} - \frac{E_b}{R_b^2(1-\nu_b^2)} \int_{-\frac{h_b}{2}}^{\frac{h_b}{2}} \bar{w}_b dz$$

$$+ D^2 \left\{ \frac{M_{Tx_b}}{1-\nu_b} - \frac{E_b \nu_b}{R_b(1-\nu_b^2)} \right\} \int_{-\frac{h_b}{2}}^{\frac{h_b}{2}} z \bar{w}_b dz \}$$

$$\alpha_{37} = D \left\{ \frac{N_{Tx_a}}{1-\nu_a} - \frac{C_a \nu_a}{R_a} \hat{E} - \frac{E_a \nu_a}{R_a(1-\nu_a^2)} \right\} \int_{-\frac{h_a}{2}}^{\frac{h_a}{2}} \bar{w}_a dz \}$$

$$\alpha_{47} = D \left\{ \frac{N_{Tx_b}}{1-\nu_b} - \frac{E_b \nu_b}{R_b(1-\nu_b^2)} \right\} \int_{-\frac{h_b}{2}}^{\frac{h_b}{2}} \bar{w}_b dz \}$$

$$\alpha_{57} = \Delta_a + D \left\{ \hat{E} + \frac{M_{Tx_a}}{1-\nu_a} - \frac{E_a \nu_a}{R_a(1-\nu_a^2)} \right\} \int_{-\frac{h_a}{2}}^{\frac{h_a}{2}} z \bar{w}_a dz \}$$

$$\alpha_{67} = \Delta_b + D \left\{ \frac{M_{Tx_b}}{1-\nu_b} - \frac{E_b \nu_b}{R_b(1-\nu_b^2)} \right\} \int_{-\frac{h_b}{2}}^{\frac{h_b}{2}} z \bar{w}_b dz \}$$

(20)

with

$$\pi_1 = -p_{1a} - \frac{h_a}{2} D\tau_{1a}$$

$$\pi_2 = +p_{2b} - \frac{h_b}{2} D\tau_{2b}$$

$$\pi_3 = -\tau_{1a}$$

$$\pi_4 = +\tau_{2b}$$

$$\pi_5 = -\frac{5}{12} h_a \tau_{1a}$$

$$\pi_6 = -\frac{5}{12} h_b \tau_{2b} \quad (21)$$

$$\Delta_i = \frac{5}{4} \int_{-\frac{h_i}{2}}^{\frac{h_i}{2}} \left\{ G_c^i \bar{w}_i' - \left(\frac{2z}{h_i} \right)^2 \left[G_c^i \bar{w}_i' - \frac{1}{R} \right]_{-\frac{h_i}{2}}^z \Omega dy \right\} dz \quad (i = a, b)$$

The governing equations may be transformed to a more useful form for solution. Since they are cumbersome for manipulation in their present form, a matrix notation was used to define the coefficients of the unknowns. These are tabulated in Appendix C. After some manipulation of the governing equations, the following are obtained:

$$(g_1 D^7 + g_2 D^5 + g_3 D^3 + g_4 D) w_b = L_I(x) \quad (22)$$

$$(b_{11} D^4 + b_{12} D^2 + b_{13}) \beta_b = L_{II}(x) - (b_{14} D^3 + b_{15} D) w_b \quad (23)$$

$$a_{23} D \beta_a^2 + L_{III}(x) - (a_{21} D^2 + a_{22}) \beta_b - a_{24} D w_b \quad (24)$$

$$D^2 u_{ob} = L_{IV}(x) - k_1 D^2 \beta_a - k_2 D \beta_b - k_3 D w_b \quad (25)$$

$$\tau_j = L_V(x) - k_4 D^2 u_{ob} - k_5 D w_b \quad (26)$$

$$p_j = L_{VI}(x) - k_6 D^3 \beta_b - k_7 D \tau_j + k_8 D u_{ob} + k_9 w_b \quad (27)$$

III. SOLUTION OF GOVERNING EQUATIONS

Homogeneous Solution:

Consider first W_b . Assuming a solution of the form $W_b = e^{sx}$ and letting $y = s^2$, (22) then takes the form

$$y^3 + \frac{g_2}{g_1} y^2 + \frac{g_3}{g_1} y + \frac{g_4}{g_1} = 0 \quad (28)$$

for the homogeneous solution.

The solution of equations in the form (28) is developed in reference (42) and leads to three possibilities for the roots:

Case 1: there are two conjugate imaginary roots and one real root

Case 2: there are three real and unequal roots

Case 3: there are three real roots of which at least two are equal.

Case 1 leads to a solution of (22) which is of the form

$$W_{bH} = V_1 e^{c_1 x} + V_2 e^{-c_1 x} + e^{c_2 x} (V_3 \cos c_3 x + V_4 \sin c_3 x) + e^{-c_2 x} (V_5 \cos c_3 x + V_6 \sin c_3 x) \quad (29)$$

where $V_1 - V_6$ are constants to be evaluated through boundary conditions.

The constants $c_1 - c_3$ are defined in Appendix C.

The Case 2 solution can take on several forms depending on whether the roots of (28) are positive or negative. The final forms

for the case of one, two and three positive real roots respectively,
are:

$$W_{bh} = V_1 e^{c_4 x} + V_2 e^{-c_4 x} + V_3 \cos c_5 x + V_4 \sin c_5 x \\ + V_5 \cos c_6 x + V_6 \sin c_6 x \quad (30)$$

$$W_{bH} = V_1 e^{c_4 x} + V_2 e^{-c_4 x} + V_3 e^{c_5 x} + V_4 e^{-c_5 x} \\ + V_5 \cos c_6 x + V_6 \sin c_6 x \quad (31)$$

$$W_{bH} = V_1 e^{c_4 x} + V_2 e^{-c_4 x} + V_3 e^{c_5 x} + V_4 e^{-c_5 x} \\ + V_5 e^{c_6 x} + V_6 e^{-c_6 x} \quad (32)$$

where, again, the $V_1 - V_6$ are boundary value constants and $c_4 - c_6$ are defined in Appendix C.

Case 3 represents the degenerate forms (30) - (32) where two of the roots are equal. The equations then take the form:

$$W_{bh} = V_1 e^{c_4 x} + V_2 e^{-c_4 x} + (V_3 + V_5 x) \cos c_5 x + (V_4 + V_6 x) \sin c_5 x \quad (33)$$

$$W_{bH} = (V_1 + V_3 x) e^{c_4 x} + (V_2 + V_4 x) e^{-c_4 x} + V_5 \cos c_6 x \\ + V_6 \sin c_6 x \quad (34)$$

$$W_{bH} = (V_1 + V_3 x) e^{c_4 x} + (V_2 + V_4 x) e^{-c_4 x} + V_5 e^{c_6 x} \\ + V_6 e^{-c_6 x} \quad (35)$$

In treating single-layered cylinders, it is possible to write the $c_1 - c_6$ explicitly in terms of the physical quantities

involved and thus have a feel for the physical behavior of the shell. In the present derivation, however, these expressions are so lengthy and involved that, unless one has a specific example in mind, their explicit form for the general case is of dubious utility. A point is reached where one must decide whether to obscure the physical situation with mathematics or obscure the mathematics with the physical quantities involved. The former course was chosen to show where the mathematical formulation is analogous to that for the single-layered cylinder (Case 1) and where it diverges (Cases 2 and 3). To impart greater physical significance to the equations it becomes necessary to either consider a specific case and determine the form and constants listed in Appendix C will take or resort to numerical evaluation of the constants.

Consider next equation (25). This may be integrated directly to give

$$U_{ob} = \iint L_{IV}(x) dx dx - k_1 \beta_a - k_2 \beta_b - k_3 \int w_b dx + V_7 x + V_8 \quad (36)$$

where V_7 and V_8 are also boundary value constants. Since there are also eight boundary conditions, four at each edge, it follows that the homogeneous solutions for the remaining unknowns β_a and β_b are not required. Their particular solutions will suffice to satisfy the governing equations. This is equivalent to setting the boundary value constants of the β_a and β_b homogeneous solutions to zero.

Total Solution:

The total solution will consist of the homogeneous solutions given in the previous section together with whatever particular solutions are called for due to the mechanical and thermal loading for any given problem. Total solutions for displacements will take the form:

$$W_b(x) = W_{bH} + W_{b \text{ part}} \quad (37)$$

$$\beta_b(x) = \beta_{b \text{ part}} \quad (38)$$

$$\beta_a(x) = \beta_{a \text{ part}} \quad (39)$$

where $W_{b \text{ part}}$ is the particular solution of (22) due to $L_I(x)$, and

$\beta_{b \text{ part}}$ is the particular solution of (23) due to $L_{II}(x) - (b_{14}D^3 + b_{15}D)W_b(x)$, and

$\beta_{a \text{ part}}$ is the particular solution of (24) due to $L_{III}(x) - (a_{21}D^2 + a_{22})\beta_b(x) - a_{24}DW_b(x)$.

Once (37) - (39) are known, u_{ob} may be found from (36) and the joint shear and normal stresses from (26) and (27) respectively.

IV. BOUNDARY CONDITIONS

Boundary conditions for plates and shells are listed in many sources (21), (34), (37), (36), (37). For the axi-symmetric case, they are usually stated as:

At the edges $x = 0$ and $x = L$

either u or N prescribed

either w or Q prescribed (40)

either β or M prescribed

For multilayered problems, the same boundary conditions usually apply if N , M and Q are interpreted as resultants \bar{N} , \bar{M} , \bar{Q} . Considering a two-layered cylinder for example, laminas a and b , the resultants would be:

$$\begin{aligned}\bar{N} &= N_a + N_b \\ \bar{Q} &= Q_a + Q_b \\ \bar{M} &= M_a + M_b + \left(\frac{h_a}{2} + \frac{h_b}{2}\right)N_a\end{aligned}\tag{41}$$

However, (40), (41) do not make use of the no slip - no delamination conditions. These provide two constraints not only on displacements but on boundary conditions as well.

Appropriate boundary conditions can be derived using the principle of virtual displacements of the layer middle surfaces.

$$\begin{aligned}
& u_i^* \\
& \beta_i^* \quad (i = a, b) \\
& w_i^*
\end{aligned} \tag{42}$$

where the asterisk denotes a virtual quantity. Multiplying equations (10) by the virtual displacements (42) in the order listed, adding the products, integrating over shell length and summing over the layers, we get:

$$\begin{aligned}
& \sum_{i=a}^b \int_C \{ [N_{x_i} + (\tau_{1i} - \tau_{2i})] u_i^* + \\
& [M_{x_i} - Q_i + 1/2 h_i (\tau_{1i} + \tau_{2i})] \beta_i^* + \\
& [Q_i - (1/R) N_{\theta i} + (P_{1i} - P_{2i})] w_i^* \} dC = 0
\end{aligned} \tag{43}$$

since the virtual work done by a shell in equilibrium through a virtual displacement is equal to zero.

After integration by parts, the virtual work principle for the multilayered cylinder takes the form:

$$\begin{aligned}
& \sum_{i=a}^b \int_0^L [N_{x_i} u_i^* + M_{x_i} \beta_i^* + Q_i w_i^*]_0^L \\
& + \int_0^L [(\tau_{1i} - \tau_{2i}) u_i^* + 1/2 h_i (\tau_{1i} + \tau_{2i}) \beta_i^*
\end{aligned}$$

$$+ (p_{1i} - p_{2i}) w_i^*] dx \} = .$$

$$\sum_{i=a}^b \int_0^L [N_{x_i} u_i^* + M_{x_i} \beta_i^* + Q_i (\beta_i^* + w_i^*)] dx \quad (44)$$

$$+ \frac{N_{\theta i}}{R} w_i^*] dx$$

The first quantity in brackets on the left hand side of (44) involves terms which are specified at the boundaries, namely:

$$\sum_{i=a}^b [N_{x_i} u_i^* + M_{x_i} \beta_i^* + Q_i w_i^*]_0^L$$

Summing over layers gives

$$[N_{x_a} u_a^* + M_{x_a} \beta_a^* + Q_a w_a^* + N_{x_b} u_b^* + M_{x_b} \beta_b^* + Q_b w_b^*]_0^L$$

Since this derivation does not allow for slip or delamination, the virtual displacements must be constrained to:

$$u_a^* = u_b^* + (1/2) h_a \beta_a^* + (1/2) h_b \beta_b^* \quad (45)$$

$$w_a^* = w_b^* + (\bar{w}_b - \bar{w}_a)^*$$

Substituting into (49) and collecting terms, the quantities to be specified at $x = 0$ and $x = L$ are found to be:

either $N_{x_a} + M_{x_b}$ or u_b specified

either $1/2 h_a N_{x_a} + M_{x_a}$ or β_a specified

(46)

either $1/2 h_b N_{x_b} + M_{x_b}$ or β_b specified

either $Q_a + Q_b$ or w_b specified

$Q_a (\bar{w}_b - \bar{w}_a)$ specified

Note that the first four conditions are not a unique set.

Other possibilities are (u_a, u_b, β_a, w_b) or (u_a, u_b, β_b, w_b) specified, but the set (46) seems to be a good choice.

The last condition of (46) is the result of retaining thermal expansion through-the-thickness while dropping terms of order h/R throughout the remainder of the derivation. For cases where \bar{w} must be retained, the following procedure can be employed. For simplicity, free-free boundary conditions are considered, though the treatment is analogous for any set of conditions specified.

Since the last of (46) is a temperature dependent, $\bar{w}_b - \bar{w}_a$

is a priori specified. Hence, for free boundaries, either $Q_a = 0$ or $(\bar{w}_b - \bar{w}_a) = 0$. $Q_a = 0$ is acceptable for special cases but is not generally true. $(\bar{w}_b - \bar{w}_a)$ can be rigorously satisfied by redefining the reference surface location in layer a, i.e.,

$$\bar{w}_a(x, Z_0) = \int_0^{Z_0} \alpha_{c_a} T_a(x, z) dz \quad (47)$$

$$w_b(x, \frac{h_b}{2}) = \int_0^{h_b/2} \alpha_{c_b} T_b(x, z) dz$$

Under these conditions, the no-slip requirement becomes

$$u_{0_a} = u_{0_b} + (-Z_0 \beta_a + 1/2 h_b \beta_b) \quad (48)$$

and all other results remain the same if $1/2 h_a$ is replaced by $-Z_0$.

Note that when this approach is used, the second strain definition of equation (8) must be used in the form:

$$\epsilon_\theta = \frac{w_i + \bar{w}_i}{R_i + Z} \quad (i = a, b)$$

in order to have a zero stress state for the case of an isotropic material with the same α_c in both layers and $T = \text{constant}$.

V. SAMPLE PROBLEM

The problem selected was the slow cooling of pyrolytic graphite. Because of the difference in thermal expansion coefficients in both "a" and "c" directions of the pyrolytic graphite and mandrel material and also because of curvature effects, normal and shear stresses at the deposit-mandrel interface will be formed during the cooling process. These may be of sufficient magnitude to cause flaking or delamination. An investigation of their behavior with changing material and geometric properties is therefore of interest.

The test case considered a laminated cylinder with free-free edges and a constant temperature $T_0 = -1000^\circ\text{F}$. The material properties used were averaged in the range $3000^\circ\text{F} - 2000^\circ\text{F}$. Layer a, the top-most layer, was taken to be pyrolytic graphite, layer b commercial (ATJ) graphite. The properties used are given in Table (B.1). The calculations were performed with the aid of a CDC 6600 computer. The program tabulation is given in Appendix A. Results showing the behavior of τ_j and p_j due to variation in lamina thickness and/or E/G_c ratio are in Appendix B. From these, the following conclusions may be drawn:

1. Figures (B.1) and (B.2) indicate a decrease in normal and shear stresses at the mandrel - PG interface as the E/G_c ratio is increases, implying that a material weak in shear is desirable to minimize stresses and therefore the possibility of debonding.

2. Figures (B.3) and (B.4) show results when $h = 0.50$, $E/G_c = 20$ and $5 \geq h_a/h_b \geq 1$. Figures (B.5) and (B.6) show the case where $h_b = 0.25$, $E/G_c = 20$, $4 \geq h_a/h_b \geq 1$. All the curves indicate that a high h_a/h_b ratio is the desirable, for given material properties, for minimization of normal and shear stresses at the joint. This implies that a thin graphite mandrel is preferable to a thick mandrel. Note that when $h_a/h_b = 1$, the ultimate stress ($\sigma_{ult} = 18,000$ psi) in tension for pyrolytic graphite is exceeded.

3. To determine the behavior of the roots of equation (22) and establish a range wherein the various forms of solution (29)-(35) will occur, computations were also made for $1000 \geq h_a/h_b \geq 0.001$ with E/G_c at 50, 20 and 2.6. Results show that the Case 1 solution in the form (29) occurs whenever $h_a/h_b \geq 1$ independent of the E/G_c ratio. The constants $c_1 - c_3$ are affected by E/G_c , the constant c_1 being much more sensitive to a change in E/G_c than c_2 or c_3 .

4. The first two terms of (29) have their principal effects at the edge only. It was also observed that for $h_a/h_b \gg 1$, the value c_1 becomes so very large that the boundary value constants V_1 and V_2 tend to become very small, and as a first approximation the terms containing them can be dropped from the solution functions. In this case, (29) takes on a form similar to that for the solution for radial displacement of a single-layered cylinder, with suitably defined constants.

5. It should be kept in mind that all the above remarks are based on the numerical results and are valid in the range.

$$1000 \gg h_a/h_b \gg 0.001; 400 \gg L/h \gg 26; 0.05 \gg h/R \gg 0.0033$$

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APPENDIX A
Computer Program

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FORTRAN IV PROGRAM PGE(INPUT,OUTPUT,TAPES=INPUT,TAPE9=OUTPUT)      ***** 1
  DIMENSION ICO(6), IIM(6), IRE(6)                                  PGE 2
  DIMENSION ORQ(10), HP(10), HQ(10), COEF(5), G3ACO(5), ROOTS(2,6) PGE 3
  COMMON /SHEAR/ OA,OB,FO                                           PGE 4
  COMMON /KLP/ IC1,IC2,IC3,IC4,IC5,IC6,IM1,IM2,IM3,IM4,IM5,IM6,IR1,IPGE 5
  IR2,IR3,IR4,IR5,IR6                                             PGE 6
  COMMON DLTEMP                                                     PGE 7
  COMMON TEMP,EA,EAC,NUA,NUAC,EB,EBC,NUR,NUBC,HA,HB,R,G3A,G3B,CA,CB,PGE 8
  IDA,DB,H,CACB,CAPCB,NUNU,GAMA,GAMB,K(40),A(55),G(7),BT(30),FVTOA,FVPG 9
  2TDB,BDE(600),LOAD1,AM(50),LOAD2,EN(6,7),FNXA,FNXB,FMXA,FMXB,FNX,FMPGE 10
  3X,FYOA,FYOB,AC(3),KLIP,ELL,DELPH(20),V(6),AL(18),WM,DWM,DDWM,WB,DWPG 11
  4B,DWNB,DDWNB,WA,DWA,WU,DWU,DDWU,DDWU,TAU,DTAU,WPJ,WMA,DWMA,WUA,DWPG 12
  SUA                                                             PGE 13
  EQUIVALENCE (IC1,ICO(1)), (IM1,IIM(1)), (IR1,IRE(1))          PGE 14E
  REAL NUA,NUAC,NUB,NJBC,NUNU,K,LOAD1,LOAD2                      PGE 15
  READ (5,12) (ORQ(I),I=1,4)                                       PGE 16R
  READ (5,12) (HP(I),I=1,6)                                         PGE 17R
  READ (5,12) (HQ(M),M=1,4)                                         PGE 18R
  READ (5,11) TEMP,DLTEMP                                           PGE 19R
  READ (5,11) EA,EAC,NUA,NUAC                                       PGE 20R
  READ (5,11) EB,EBC,NUB,NUBC                                       PGE 21R
  READ (5,9) BT(1),BT(2),BT(3),BT(4)                               PGE 22R
  READ (5,10) (DELPH(J),J=1,11)                                     PGE 23R
  WRITE (9,13) (HQ(M),M=1,4)                                         PGE 24W
  WRITE (9,14) TEMP                                                 PGE 25W
  WRITE (9,15) EA,EAC,NUA,NUAC                                       PGE 26W
  WRITE (9,16) EB,EBC,NUB,NUBC                                       PGE 27W
  COEF(1)=50.                                                       PGE 28
  COEF(2)=20.                                                       PGE 29
  COEF(3)=2.6                                                       PGE 30
  DO 8 I=1,6                                                         PGE 31
  DO 8 M=1,4                                                         PGE 32
  DO 8 L=1,3                                                         PGE 33
  HA=HP(114)                                                         PGE 34
  HB=HQ(M)                                                           PGE 35
  R=30.                                                             PGE 36
  G3ACO(1)=EA/COEF(1)                                               PGE 37
  G3A=G3ACO(1)                                                       PGE 38
  G3B=EB/(2.*(1.+NUBC))                                             PGE 39
  CALL PRELIM (L,M,114)                                             PGE 40
  CALL POLYR (6,G,ROOTS,0)                                          PGE 41
  CALL RICE (KOUTS,AC,KLIP)                                         PGE 42
  CALL THERM                                                         PGE 43
  CALL MISC                                                         PGE 44
  DO 7 IJP=1,4                                                       PGE 45
  ELL=CRO(IJP)                                                       PGE 46
  IF (ELL.EQ.0.0) GO TO 7                                           PGE 47
  AM(1)=AC(2)**2-AC(3)**2                                           PGE 48
  AM(2)=2.*(AC(2)*AC(3))                                           PGE 49
  AM(3)=AC(2)*AM(1)-AC(3)*AM(2)                                     PGE 50
  AM(4)=AC(3)*AM(1)+AC(2)*AM(2)                                     PGE 51
  AM(17)=AM(1)*AC(3)-AM(2)*AC(2)                                   PGE 52
  AM(18)=AM(1)*AC(2)+AM(2)*AC(3)                                   PGE 53
  AM(25)=AM(1)*AC(2)+AM(2)*AC(3)                                   PGE 54
  AM(26)=AM(1)*AC(3)-AM(2)*AC(2)                                   PGE 55
  AM(5)=AC(2)*AM(3)-AC(3)*AM(4)                                    PGE 56
  AM(6)=AC(3)*AM(3)+AC(2)*AM(4)                                    PGE 57
  IF (KLIP=5) 1,2,3                                                 PGE 58
1 CONTINUE                                                         PGE 59
  IF (KLIP=2) 4,5,6                                                 PGE 60

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2 CALL EKLIPI5	PGE 61
GO TO 6	PGE 62
3 CALL EKLIPI6	PGE 63
GO TO 6	PGE 64
4 CALL EKLIPI1	PGE 65
GO TO 6	PGE 66
5 CALL EKLIPI2	PGE 67
6 CONTINUE	PGE 68
CALL MISC	PGE 69
7 CONTINUE	PGE 70
8 CONTINUE	PGE 71
	PGE 72
	PGE 73
	PGE 74
9 FORMAT (4E15.7)	PGE 75
10 FORMAT (6E12.5)	PGE 76
11 FORMAT (4E15.0)	PGE 77
12 FORMAT (16F4.0)	PGE 78
13 FORMAT (4H MB=12F9.3)	PGE 79
14 FORMAT (1X, SHTEMP=512.5)	PGE 80
15 FORMAT (1X, 6HAPROP=4E15.7)	PGE 81
16 FORMAT (1X, 6HMBPROP=4E15.7/)	PGE 82
END	PGE 83-
SUBROUTINE UC OFF (FNTWM, CONST, X)	***** 1
COMMON DLTEMP	2
COMMON TEMP, EA, EAC, NUA, NUAC, EB, EBC, NUB, NUBC, HA, HB, R, G3A, G3B, CA, CB,	3
IDA, DB, H, CACB, CAPCB, NUNU, GAMA, GAMB, K(40), A(55), G(7), BT(30), FNTQA, FN	4
2TOB, BDE(600), LOAD1, AM(50), LOAD2, EN(6, 7), FNXA, FNXB, FMXA, FMXB, FNX, FM	5
3X, FNQA, FNQB, AC(3), KLIP, ELL, DELPH(20), V(6), AL(18), WM, DWM, DDWM, WB, DW	6
4B, DDWB, DDDWB, WA, DWA, WU, DWU, DDWU, DDDWU, TAU, DTAU, WPJ, WWA, DWWA, WUA, DW	7
SUA	8
REAL NUA, NUAC, NUB, NUBC, NUNU, K, LOAD1, LOAD2	9
PP4=-(ELL/2.)*AC(1)	10
PP5=-(ELL/2.)*AC(2)	11
PP6=-(ELL/2.)*AC(3)	12
CONST=BUE(19)*WA+HDE(20)*WB+BDE(21)*FNTWM-BT(20)*X/CAPCB	13
RETURN	14
END	15-
SUBROUTINE PRELIM (L, M, I14)	***** 1
COMMON DLTEMP	PREL 2
COMMON TEMP, EA, EAC, NUA, NUAC, EB, EBC, NUB, NUBC, HA, HB, R, G3A, G3B, CA, CB, PREL	3
IDA, DB, H, CACB, CAPCB, NUNU, GAMA, GAMB, K(40), A(55), G(7), BT(30), FNTQA, FMPREL	4
2TOB, BDE(600), LOAD1, AM(50), LOAD2, EN(6, 7), FNXA, FNXB, FMXA, FMXB, FNX, FMPREL	5
3X, FNQA, FNQB, AC(3), KLIP, ELL, DELPH(20), V(6), AL(18), WM, DWM, DDWM, WB, DWPREL	6
4B, DDWB, DDDWB, WA, DWA, WU, DWU, DDWU, DDDWU, TAU, DTAU, WPJ, WWA, DWWA, WUA, DWPREL	7
SUA	PREL 8
REAL NUA, NUAC, NUB, NUBC, NUNU, K, LOAD1, LOAD2	PREL 9
CA=EA*HA/(1.-NUA**2)	PREL 10
CB=EB*HB/(1.-NUB**2)	PREL 11
DA=EA*(HA**3)/(12.*(1.-NUA**2))	PREL 12
DB=EB*(HB**3)/(12.*(1.-NUB**2))	PREL 13
H=HA+HB	PREL 14
CACB=CA*CB	PREL 15
CAPCB=CA*CB	PREL 16
NUNU=NUA-NUB	PREL 17
GAMA=(5./6.)*(G3A*HA)	PREL 18
GAMB=(5./6.)*(G3B*HB)	PREL 19
K(15)=1.*CB/CA	PREL 20
K(16)=1.*CA/CB	PREL 21
K(17)=HA/HB	PREL 22

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A(11)=(1./12.)*(HA**2)+H*HA/(4.*K(16))
A(12)=-HA*NUNU/(12.*R)*K(16))
A(13)=(HB*HB)*(K(15)/K(16))/12.*H*HB/(4.*K(16))
A(14)=HB*A(12)/HA
A(15)=-H/HA)*A(12)
FDUM=(CA*NUA+CA*HUR)**2
CDUM=CAPCB**2
BDUM=(CA*CAPCB)*(R**2)
A(16)=(FDUM-CDUM)/BDUM
A(21)=(HB**2)/12.+(5./24.)*(HB**2)/K(15)
A(22)=-((5./6.)*(G3A/EA)*(1.-NUB**2))
A(23)=(5./24.)*(HA*HB)/K(15)
A(24)=-((5./6.)*HB/HA)*A(12)-A(22)
A(31)=(11./12.)*(HA**2)*(1.+5./12.*K(16))
A(32)=-((5./6.)*(G3A/EA)*(1.-NUA**2))
A(33)=(K(15)/K(16))*A(23)
A(34)=(5./12.)*(HA/(R*K(16)))-GAMA/CA
      A(41) = D(11)
      A(42) = B(12)
      A(43) = B(13)
      A(44) = B(14)
      A(45) = B(15)
      A(51)=B(21)
      A(52) = A(22)
      A(53) = B(23)
      A(54) = B(24)
      A(55) = B(25)

A(41)=A(13)*A(23)-A(11)*A(21)
A(42)=A(14)*A(23)-A(11)*A(22)-A(21)*A(12)
A(43)=-A(12)*A(22)
A(44)=A(15)*A(23)-A(11)*A(24)
A(45)=A(16)*A(23)-A(12)*A(24)
A(51)=A(23)**2-A(31)*A(21)
A(52)=-A(31)*A(22)-A(32)*A(21)
A(53)=-A(32)*A(22)
A(54)=A(23)*A(34)-A(31)*A(24)
A(55)=-A(32)*A(24)
      G(1) = D(1)
      G(3) = D(2)
      G(5) = D(3)
      G(7) = D(4)
G(1)=A(51)*A(44)-A(41)*A(54)
G(3)=A(51)*A(45)+A(52)*A(44)-A(42)*A(54)-A(41)*A(55)
G(5)=A(52)*A(45)+A(53)*A(44)-A(42)*A(55)-A(43)*A(54)
G(7)=A(53)*A(45)-A(43)*A(55)
G(2)=0.0
G(4)=G(2)
G(6)=G(4)
WRITE (9,3) L,G3A,G3B
WRITE (9,4) HA,HB,R,TEMP
WRITE (9,5) (IOP,G(IOP),IOP=1,7)
WRITE (9,1) CA,CB,DA,DB
WRITE (9,6) CACH,CAPCB,GAMA,GAMB
WRITE (9,8) HA,HB,ELL,L,K(17)
WRITE (9,7)
K(1)=1.0
K(2)=G(3)/G(1)
K(3)=G(5)/G(1)
K(4)=G(7)/G(1)
FRUMP=(3.*K(3)-K(2)**2)/3.
HRMP=(2.*(K(2)**3)-9.*(K(2)*K(3))+27.*K(4))/27.
      FRUMP = SMALL A
      HRMP = SMALL B
RATIO=HA/HB

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PREL123
PREL124
PREL125
PREL126
PREL127
PREL128
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PREL162
PREL163
PREL164
PREL165
PREL166
PREL167W
PREL168W
PREL169W
PREL170W
PREL171W
PREL172W
PREL173W
PREL174
PREL175
PREL176
PREL177
PREL178
PREL179
PREL180
PREL181
PREL182

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DISC=(HRPP**2)/4.+(FRUMP**3)/27. PREL183
WRITE (9,2) HA,HB,RATIO,DISC PREL184
DISCRIMINANT.LT.0 IMPLIES KLIP = 2,3,4, OR 5 DEPENDING ON R PREL185
DISCRIMINANT.EQ.0 IMPLIES DEGENERATE CASE PREL186
DISCRIMINANT.GT.0 KLIP = 1 MODIFIED CLASSICAL SOLUTION PREL187
RETURN PREL188
PREL189
PREL190
1 FORMAT (1X,3HCA=E12.5,1X,3HCB=E12.5,1X,3HDA=E12.5,1X,3HDB=E12.5/) PREL191
2 FORMAT (1//1X,3HHA=F10.3,1X,3HMB=F10.3,5X,6HHA/MB=F15.7,5X,13HDISCRIP PREL192
1IMINANT=E15.7//) PREL193
3 FORMAT (2X,4HGA(12,2H)=E15.7,4HGB(E15.7//) PREL194
4 FORMAT (2X,4H HA=F10.2,2X,4H MB=F10.2,2X,2HR=F10.2,2X,6H TEMP=F10. PREL195
12//) PREL196
5 FORMAT (4(2X,2HG(12,2H)=E12.5)/3(2X,2HG(12,2H)=E12.5)) PREL197
6 FORMAT (2X,5HCACB=E12.5,1X,6HCAPCB=E12.5,1X,5HGAMA=E12.5,1X,5HGAMB PREL198
1=E12.5//) PREL199
7 FORMAT (20H A(1) ARE ) PREL1
8 FORMAT (F12.5,F12.5,F12.5,I3,1X,7H HA/MB=F12.5) PREL1
END PREL1
SUBROUTINE MISC ***** 1
COMMON DLTEMP 2
COMMON TEMP,EA,EAC,NUA,NUAC,EB,ERC,NUB,NUBC,HA,HB,R,G3A,G3B,CA,CB, 3
IDA,DB,H,CACB,CAPCB,NUNU,GAMA,GAMB,K(40),A(55),G(7),HT(30),FNTQA,FN 4
2TOB,BDE(600),LOAD1,AM(50),LOAD2,EN(6,7),FNXA,FNXB,FNXC,FNXD,FNX,FM 5
3X,FYQA,FYQB,AC(3),KLIP,ELL,DELPHI(20),V(6),AL(18),WW,UWW,DWW,XB,DW 6
4B,DWB,DDWB,WA,DWA,WU,DWU,DWUW,DOUWU,TAU,DTAU,WPJ,WHA,DHWA,WUA,DW 7
SUA 8
REAL NUA,NUAC,NUR,NUBC,NUNU,K,LOAD1,LOAD2 9
BDE(19)=CA*HA/(2.*CAPCB) 10
BDE(20)=(HB/HA)*BDE(19) 11
BDE(21)=(1./R)*(CA*NUA+CB*NUB)/CAPCB 12
BDE(22)=CB 13
BDE(23)=BDE(22)*NUR/R 14
BDE(24)=(HB**2)*BDE(22)/12. 15
BDE(25)=HR/2. 16
BDE(26)=BDE(22)/(R**2) 17
18
K(1)=BDE(19) 19
K(2)=BDE(20) 20
K(3)=BDE(21) 21
K(4)=BDE(22) 22
K(5)=K(8)=BDE(23) 23
K(6)=BDE(24) 24
K(7)=BDE(25) 25
K(9)=BDE(26) 26
27
28
AM(24)=AC(2)**2+AC(3)**2 29
BDE(37)=GAMA+GAMB 30
BDE(38)=BDE(37)-(H/12.)*BDE(23) 31
LOAD2=DT(11) 32
LOAD1=0.0 33
34
35
BDE(61)=BDE(19)*BDE(22) 36
BDE(62)=BDE(20)*BDE(22) 37
BDE(63)=BDE(21)*BDE(22)-BDE(23) 38
BDE(64)=BDE(24)*BDE(25)+(BDE(22)*BDE(20)) 39
BDE(65)=BDE(25)*(BDE(22)*BDE(19)) 40

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BDE(66)=BDE(25)*BDE(63).	41
BDE(67)=BDE(23)*BDE(19)	42
BDE(68)=BDE(23)*BDE(20)	43
BDE(69)=BDE(26)-BDE(23)*BDE(21)	44
AM(32)=(CA*NUA+CB*NUB)/R	45
AM(33)=CA*HA/2.	46
AM(34)=CA*HB/2.	47
AM(35)=DA+(H*HA)*CA/4.	48
AM(36)=DB+(H*HB)*CA/4.	49
AM(37)=AM(33)+AM(34)	50
AM(38)=AM(37)*NUA/R	51
AM(39)=GAMA+GAMB	52
AM(42)=DA+(CA/4.)*(HA**2)	53
AM(43)=(HA*HB)*(CA/4.)	54
BDE(81)=AM(33)*NUA/R	55
BDE(82)=AM(34)*NUA/R	56
AM(44)=AM(42)-AM(33)*BDE(19)	57
AM(45)=AM(43)-AM(33)*BDE(20)	58
AM(46)=BDE(81)-AM(33)*BDE(21)	59
AM(47)=DB+(CA/4.)*(HB**2)	60
AM(48)=AM(43)-AM(34)*BDE(19)	61
AM(49)=AM(47)-AM(34)*BDE(20)	62
AM(50)=BDE(82)-AM(34)*BDE(21)	63
BDE(83)=(H/12.)*(BDE(22)*BDE(19))	64
BDE(84)=(H/12.)*(BDE(22)*BDE(20))	65
BDE(85)=AM(39)+(H/12.)*(BDE(22)*BDE(21)-BDE(23))	66
BDE(71)=AM(33)-CA*BDE(19)	67
BDE(72)=AM(34)-CA*BDE(20)	68
BDE(73)=CA*(NUA/R-BDE(21))	69
BDE(74)=CA*(NUA*BT(4)/R+BT(20)/CAPCB)-BT(5)/(1.-NUA)	70
BDE(75)=-CA*BDE(19)	71
BDE(76)=-CB*BDE(20)	72
BDE(77)=CS*(NUB/R-BDE(21))	73
BDE(91)=AM(44)+AM(48)	74
BDE(92)=AM(45)+AM(49)	75
BDE(93)=AM(46)+AM(50)	76
RETURN	77
END	78
SUBROUTINE THERM	79
COMMON DLTMP	***** 1
COMMON TEMP,EA,EAC,NUA,NUAC,EB,EBC,NUB,NUBC,HA,HB,R,G3A,G3B,CA,CB,THERM	2
1DA,DB,H,CACB,CAPCB,NUU,GAMA,GAMB,K(40),A(55),G(7),RT(30),FNTOA,FNTHERM	3
2TOB,BDE(830),LOAD1,AM(50),LOAD2,EN(6,7),FNXA,FNXB,FNXC,FNXD,FNTE	4
3X,FNOA,FNOB,AC(3),KLIP,ELL,JELPH(20),V(6),AL(18),WW,DWW,DDWW,WB,DW	5
4B,DDWB,DDWB,WA,DWA,U,DWU,DDWU,DDWU,TAU,DTAU,WPI,WWA,DWWA,WUA,CW	6
5UA	7
REAL NUA,NUAC,NUB,NUBC,NUU,K,LOAD1,LOAD2	THERM 8
BT(1)=ALPHA(A)	THERM 9
BT(2)=ALPHA(AC)	THERM10
BT(3)=ALPHA(B)	THERM11
BT(4)=ALPHA(BC)	THERM12
BT(5)=NTRA=NTOA	THERM13
BT(6)=NTRB=NTOB	THERM14
BT(7)=WEBAR-A AT Z=-HA/2.	THERM15
BT(8)=WEBAR-B AT Z=+HB/2.	THERM16
BT(9)=E(BAR)=W(BAR-B)-W(BAR-A)	THERM17
BT(10)=ALPHA(1,7)	THERM18
BT(11)=ALPHA(2,7)	THERM19
BT(12)=MTXA	THERM20
	THERM21

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BT(13) = MTXB
BT(14) = EXPANSION IN THICKNESS DIRECTION - A
BT(15) = EXPANSION IN THICKNESS DIRECTION - B
BT(16) = MODIFIED AXIAL THERMAL RESULTANT A
BT(17) = AXIAL THERMAL RESULTANT (MODIFIED) - B
BT(18) = BT(14)
BT(19) = BT(15)
BT(5) = (EA*HA)*(TEMP*BT(1))
BT(6) = (EB*HB)*(TEMP*BT(2))
BT(7) = -BT(2)*(TEMP*HA/2.1*DLTEMP*(BT(2)*HA/8.))
BT(8) = BT(4)*(TEMP*HB/2.1*DLTEMP*(BT(4)*HB/8.))
BT(9) = BT(8) - BT(7)
BT(10) = BT(5)/(R*(1.-NUA)) - CA*BT(9)/(R*(2.1*(DA/(2.*HA))*(DLTEMP/(R*(1.21)*BT(2))
BT(11) = BT(6)/(R*(1.-NUB)) - (DB/(2.*HB))*(DLTEMP/(R*(2.1)*BT(4)
BT(12) = (TEMP*BT(1))*(EA*HA**3)/(12.*H)*(EA*HA**3)*(DLTEMP*BT(1))/(THERM37
112.*HA)
BT(13) = (TEMP*BT(3))*(EB*HB**3)/(12.*H)*(EB*HB**3)*(DLTEMP*BT(3))/(THERM39
112.*HB)
BT(14) = (DA*NUA)*(TEMP*BT(2))/R
BT(15) = (DB*NUB)*(TEMP*BT(4))/R
BT(16) = BT(5)/(1.-NUA) - CA*(NUA*BT(9))/R - (DA*DLTEMP/(2.*HA))*(NUA*BT(12)/R)
BT(17) = BT(6)/(1.-NUB) - (DB*DLTEMP/(2.*HB))*(NUB*BT(4))/R
BT(18) = (DA*NUA/R)*(BT(2)*TEMP)
BT(19) = (DB*NUB/R)*(BT(4)*TEMP)
BT(20) = BT(16) + BT(17)
BT(21) = (DA*(DLTEMP*BT(2))/(2.*HA))*(NUA/R)
BT(22) = (DB*(DLTEMP*BT(4))/(2.*HB))*(NUB/R)
PP3 = A(23)*A(53)
PP10 = A(32)*A(43) - A(12)*A(53)
ADUM = (PP3/CA)*(BT(10) + BT(11))
BDUM = (CA*NUA + CB*NUB)/(CA*(R*CAPCB))
CDUM = PP3*(BT(16) + BT(17))
DDUM = BDUM*CDUM
EDUM = ADUM + DDUM
FDUM = CA*HB/(CB*CAPCB)
GDUM = (FDUM*(5./12.))*(PP10*BT(5)/(1.-NUA))
HDUM = (5./12.)*(HB*PP10/CAPCB)*BT(16)
PDUM = EDUM - GDUM*HDUM
QDUM = (PP10/CB)*(BT(13) - BT(18))
RDUM = PDUM + QDUM
BDE(49) = RDUM/G(7)
WRITE (9,1)
WRITE (9,4) ADUM,BDUM,CDUM,DDUM,EDUM,FDUM,GDUM,HDUM,PDUM,RDUM,BDE(
149)
WRITE (9,2)
WRITE (9,3) (BT(I),I=1,30)
RETURN

1 FORMAT (1X,35HCOMPONENTS OF BDE(49) ARE /)
2 FORMAT (1X,10H BT ARE )
3 FORMAT (1X,6E12.5)
4 FORMAT (1X,4E15.7)
END
SUMRGUTIME EKLIP1
DIMENSION JIP(11)
COMMON /SHEAR/ QA,QB,FQ
COMMON DLTEMP
COMMON TEMP,EA,EAC,NUA,NUAC,EB,EBC,NUB,NUBC,HA,HB,R,G3A,G3B,CA,CB,EKLIP 5

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THERM22
THERM23
THERM24
THERM25
THERM26
THERM27
THERM28
THERM29
THERM30
THERM31
THERM32
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THERM36
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THERM60
THERM61
THERM62
THERM63
THERM64
THERM65W
THERM66W
THERM67W
THERM68W
THERM69W
THERM70
THERM71
THERM72
THERM73
THERM74
THERM75
THERM76-
00000 1
EKLIP 2
EKLIP 3
EKLIP 4

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1DA,DR,M,CACB,CAPCB,NUNU,GAMA,GAMH,K(40),A(55),G(7),RT(30),FNTOA,FNEKLIP 6
2TOB,BDE(600),LOAD1,AM(50),LOAD2,EN(6,7),FNXA,FNXB,FNXC,FNXD,FNX,EKLI 7
3X,FNDA,FNOB,AC(3),KLIP,ELL,DELPH(20),V(6),AL(14),WW,DWW,DDWW,WB,DWEKLIP 8
4R,DDWB,DDDBB,WA,DWA,WU,DWU,DDWU,DDDBU,TAU,DTAU,WPJ,MWA,UMWA,MUA,DWEKLIP 9
SUA
REAL NUA,NUAC,NUB,MIBC,NUNU,K,LOAD1,LOAD2
REAL JIP
P4=AC(1)*ELL
P5=AC(2)*ELL
P6=AC(3)*ELL
BDE(1)=A(44)*(AC(1)**3)+A(45)*AC(1)
BDE(2)=A(41)*(AC(1)**4)+A(42)*(AC(1)**2)+A(43)
AM(30)=BDE(1)/BDE(2)
BDE(3)=AM(5)*A(41)+AM(1)*A(42)+A(43)
BDE(4)=AM(6)*A(41)+AM(2)*A(42)
BDE(5)=AM(3)*A(44)+AC(2)*A(45)
BDE(6)=AM(4)*A(44)+AC(3)*A(45)
BDE(7)=BDE(5)*BDE(3)+BDE(6)*BDE(4)
BDE(8)=BDE(6)*BDE(3)-BDE(5)*BDE(4)
BDE(9)=BDE(3)**2+BDE(4)**2
BDE(10)=BDE(7)/BDE(4)
BDE(11)=BDE(8)/BDE(4)
AM(7)=-A(24)*AC(1)+AM(30)*(A(21)*(AC(1)**2)+A(22))
AM(8)=A(23)*(AC(1)**2)
BDE(12)=AM(7)/AM(8)
AM(19)=A(21)*AM(1)+A(22)
AM(20)=A(21)*AM(2)
AM(21)=A(24)*AC(2)
AM(22)=A(24)*AC(3)
BDE(13)=AM(19)*BDE(10)-AM(20)*BDE(11)-AM(21)
BDE(14)=AM(19)*BDE(11)+AM(20)*BDE(13)-AM(22)
BDE(15)=BDE(13)/A(23)
BDE(16)=BDE(14)/A(23)
AM(9)=AM(1)**2+AM(2)**2
AM(10)=AM(1)*BDE(15)+AM(2)*BDE(16)
AM(11)=AM(1)*BDE(16)-AM(2)*BDE(15)
BDE(17)=AM(13)/AM(9)
BDE(18)=AM(11)/AM(9)
AM(13)=-BDE(14)*BDE(12)+BDE(20)*AM(30)-BDE(21)/AC(1)
BDE(19)=BDE(19)*(AM(2)*BDE(18)-AM(1)*BDE(17))+BDE(20)*(AM(1)*BDE(17)
10)-AM(2)*BDE(11))-BDE(21)*AC(2)
BDE(28)=-BDE(19)*(AM(2)*BDE(17)+AM(1)*BDE(18))+BDE(20)*(AM(1)*BDE(17)
11)+AM(2)*BDE(10))-BDE(21)*AC(3)
BDE(29)=AM(1)*BDE(27)+AM(2)*BDE(28)
BDE(30)=AM(1)*BDE(28)-AM(2)*BDE(27)
AM(14)=BDE(29)/AM(9)
AM(15)=BDE(30)/AM(9)
BDE(31)=EXP(P5)*COS(P6)
BDE(32)=EXP(P5)*SIN(P6)
BDE(33)=EXP(-P5)*COS(P6)
BDE(34)=EXP(-P5)*SIN(P6)
BDE(51)=AC(1)*(HA*BDE(12)-HB*AM(30))
BDE(52)=-BDE(16)*HA+BDE(11)*HB
BDE(53)=BDE(17)*HA-BDE(10)*HB
BDE(54)=AC(1)*BDE(52)+AC(2)*BDE(53)
BDE(55)=AC(2)*BDE(52)-AC(3)*BDE(53)
AM(12)=BT(20)/CAPCB-BDE(21)*BT(9)

BDE(70)=LOAD2*BDE(23)*RT(20)/CAPCB

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EKLIP10
EKLIP11
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EKLIP64
EKLIP65

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[illegible]


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1(5)*BDE(141)*AL(6)*RDE(151) EKLIP
DDWB=AL(1)*BDE(102)+AL(2)*BDE(112)+AL(3)*BDE(122)+AL(4)*BDE(132)+AEKLIP
1L(5)*RDE(142)+AL(6)*BDE(152) EKLIP
DDWB=AL(1)*BDE(103)+AL(2)*BDE(113)+AL(3)*BDE(123)+AL(4)*BDE(133)+EKLIP
1AL(5)*BDE(143)+AL(6)*BDE(153) EKLIP
WA=AL(7)*EXP(P1)+AL(8)*EXP(-P1)+EXP(P2)*(AL(9)*COS(P3)+AL(10)*SIN(P3)) EKLIP
1P3)+EXP(-P2)*(AL(11)*COS(P3)+AL(12)*SIN(P3)) EKLIP
DWA=AL(7)*BDE(101)+AL(8)*BDE(111)+AL(9)*BDE(121)+AL(10)*BDE(131)+AEKLIP
1L(11)*BDE(141)+AL(12)*BDE(151) EKLIP
DDWA=AL(7)*BDE(102)+AL(8)*BDE(112)+AL(9)*BDE(122)+AL(10)*BDE(132)+EKLIP
1AL(11)*BDE(142)+AL(12)*BDE(152) EKLIP
DDWA=AL(7)*BDE(103)+AL(8)*BDE(113)+AL(9)*BDE(123)+AL(10)*BDE(133)+EKLIP
1+AL(11)*BDE(143)+AL(12)*BDE(153) EKLIP
FNTW=(1./AC(1))*(V(1)*EXP(P1)-V(2)*EXP(-P1))+(EXP(P2)/AM(24))*(V( EKLIP
13)*AC(2)*COS(P3)+AC(3)*SIN(P3))+V(4)*(AC(2)*SIN(P3)-AC(3)*COS(P3)) EKLIP
2)+(EXP(-P2)/AM(24))*(V(5)*(-AC(2)*COS(P3)+AC(3)*SIN(P3))-V(6)*(ACEKLIP
3(2)*SIN(P3)+AC(3)*COS(P3))+BDE(49)*X EKLIP
IF (KF.EQ.1) CALL UC0FF (FNTW,CONST,7) EKLIP
MU=-BDE(19)*WA-BDE(20)*WR-BDE(21)*FNTW+1RT(20)/CAPCB)*X+CONST EKLIP
DWU=-BDE(19)*DWA-BDE(20)*DWR-BDE(21)*WW+BT(20)/CAPCB EKLIP
DDMU=-BDE(19)*DDWA-BDE(20)*DDWB-BDE(21)*DDW EKLIP
DDDMU=-BDE(19)*DDWA-BDE(20)*DDWB-BDE(21)*DDW EKLIP
TAU=LOAD1-BDE(22)*DDMU-BDE(23)*DDW EKLIP
DTAU=-BDE(22)*DDMU-BDE(23)*DDW EKLIP
WPU=LOAD2-BDE(24)*DDMU-BDE(25)*DTAU+BDE(23)*DWU+BDE(26)*WW EKLIP
WUA=WW+HA*WA/2.+HB*WR/2. EKLIP
DWUA=DWU+HA*DWA/2.+HB*DWR/2. EKLIP
DDWUA=DDMU+HA*DDWA/2.+HB*DDWB/2. EKLIP
CALL RSLT (X) EKLIP
CALL PRINT (X) EKLIP
2 CONTINUE EKLIP
RETURN EKLIP

3 FORMAT (1X,9H BDE ARE /(10E12.5)) EKLIP
4 FORMAT (10E12.5) EKLIP
5 FORMAT (20H AM(J) ARE ) EKLIP
6 FORMAT (1X,7E11.4) EKLIP
7 FORMAT (4(1X,2HV(11,2H)=E12.5)/2(1X,2HV(11,2H)=E12.5)) EKLIP
8 FORMAT (1X,2CH AL(I) ARE ) EKLIP
9 FORMAT (6E12.5) EKLIP
10 FORMAT (21H EN(I,J)BY ROWS ARE ) EKLIP
END EKLIP -
SUBROUTINE EKLIP2 EKLIP 1
DIMENSION JIP(11) EKLIP 2
COMMON /SHEAR/ QA,QR,FQ EKLIP 3
COMMON DLTEMP EKLIP 4
COMMON TEMP,EA,EAC,NUA,NUAC,EB,ERC,NUB,NUBC,HA,HB,R,G3A,G3B,CA,CB,EKLIP 5
IDA,DR,H,CACB,CAPCB,MUNU,GAMA,GAMB,K(40),A(55),G(7),HT(30),FNTOA,FNEKLIP 6
2TOB,BDE(600),LOAD1,AM(50),LOAD2,ZN(6,7),FNXA,FNXB,FNXC,FNX,FMEXKLIP 7
3X,FNCA,FNCB,AC(3),KLIP,ELL,DELPH(20),V(6),AL(18),WW,DWW,DDWW,WR,DWEKLIP 8
4B,DDWB,DDWB,WA,DWA,WU,DWU,DDMU,DDMU,TAU,DTAU,WPU,WWA,UWUA,WUA,DWEKLIP 9
5UA EKLIP 10
REAL NUA,NUAC,NUB,NUBC,MUNU,K,LOAD1,LOAD2 EKLIP 11
REAL JIP EKLIP 12
P4=AC(1)*ELL EKLIP 13
P5=AC(2)*ELL EKLIP 14
P6=AC(3)*ELL EKLIP 15
BDE(1)=A(44)*(AC(1)*P3)-A(45)*AC(1) EKLIP 16

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RDE(2)=A(41)*(AC(1)**4)-A(42)*(AC(1)**2)+A(43)	EKLIP17
AM(30)=BDE(11)/BDE(2)	EKLIP18
BDE(3)=AM(5)*A(41)+AM(1)*A(42)+A(43)	EKLIP19
BDE(4)=AM(6)*A(41)+AM(2)*A(42)	EKLIP20
BDE(5)=AM(3)*A(44)+AC(2)*A(45)	EKLIP21
BDE(6)=AM(4)*A(44)+AC(3)*A(45)	EKLIP22
BDE(7)=RDE(5)*RDE(3)+RDE(6)*BDE(4)	EKLIP23
BDE(8)=RDE(6)*BDE(3)-BDE(5)*RDE(4)	EKLIP24
BDE(9)=BDE(3)**2+RDE(4)**2	EKLIP25
BDE(10)=RDE(7)/RDE(9)	EKLIP26
BDE(11)=BDE(8)/BDE(9)	EKLIP27
AM(7)=-A(24)*AC(1)+AM(30)*(A(21)*(AC(1)**2)-A(22))	EKLIP28
AM(9)=A(23)*(AC(1)**2)	EKLIP29
BDE(12)=AM(7)/AM(8)	EKLIP30
AM(19)=A(21)*AM(1)+A(22)	EKLIP31
AM(20)=A(21)*AM(2)	EKLIP32
AM(21)=A(24)*AC(2)	EKLIP33
AM(22)=A(24)*AC(3)	EKLIP34
BDE(13)=AM(19)*BDE(10)-AM(20)*BDE(11)-AM(21)	EKLIP35
BDE(14)=AM(19)*BDE(11)+AM(20)*BDE(10)-AM(22)	EKLIP36
BDE(15)=BDE(13)/A(23)	EKLIP37
BDE(16)=RDE(14)/A(23)	EKLIP38
AM(9)=AM(1)**2+AM(2)**2	EKLIP39
AM(10)=AM(1)*RDE(15)+AM(2)*BDE(16)	EKLIP40
AM(11)=AM(1)*BDE(16)-AM(2)*BDE(15)	EKLIP41
BDE(17)=AM(10)/AM(9)	EKLIP42
BDE(18)=AM(11)/AM(9)	EKLIP43
AM(13)=-BDE(19)*BDE(12)+RDE(20)*AM(30)-BDE(21)/AC(1)	EKLIP44
BDE(27)=RDE(19)*(AM(2)*RDE(18)-AM(1)*RDE(17))+BDE(20)*(AM(1)*BDE(1	EKLIP45
10)-AM(2)*BDE(11))-BDE(21)*AC(2)	EKLIP46
BDE(28)=-RDE(19)*(AM(2)*RDE(17)+AM(1)*BDE(18))+BDE(20)*(AM(1)*BDE(1	EKLIP47
11)+AM(2)*RDE(10))-RDE(21)*AC(3)	EKLIP48
BDE(29)=AM(1)*BDE(27)+AM(2)*BDE(28)	EKLIP49
BDE(30)=AM(1)*BDE(29)-AM(2)*BDE(27)	EKLIP50
AM(14)=RDE(29)/AM(9)	EKLIP51
AM(15)=BDE(30)/AM(9)	EKLIP52
BDE(31)=EXP(P5)*COS(P6)	EKLIP53
BDE(32)=EXP(P5)*SIN(P6)	EKLIP54
BDE(33)=EXP(-P5)*COS(P6)	EKLIP55
BDE(34)=EXP(-P5)*SIN(P6)	EKLIP56
BDE(51)=AC(1)*(HA*RDE(12)-HB*AM(30))	EKLIP57
BDE(52)=-BDE(18)*HA+CDE(11)*HB	EKLIP58
BDE(53)=BDE(17)*HA-RDE(10)*HB	EKLIP59
BDE(54)=AC(3)*BDE(52)+AC(2)*RDE(53)	EKLIP60
BDE(55)=AC(2)*RDE(52)-AC(3)*RDE(53)	EKLIP61
AM(12)=BT(20)/CAPCB-BDE(21)*BT(9)	EKLIP62
	EKLIP63
	EKLIP64
	EKLIP65
AM(32)=(CA*NUA+CR*NUB)/R	EKLIP66
AM(33)=CA*HA/2.	EKLIP67
AM(34)=CA*HB/2.	EKLIP68
AM(35)=DA+(H*HA)*CA/4.	EKLIP69
AM(36)=DB+(H*HB)*CA/4.	EKLIP70
AM(37)=AM(33)+AM(34)	EKLIP71
AM(38)=AM(37)*NUA/R	EKLIP72
AM(39)=GAMA*GAMB	EKLIP73
AM(42)=DA*(CA/4.)*(HA**2)	EKLIP74
AM(43)=(HA*HB)*(CA/4.)	EKLIP75
BDE(81)=AM(33)*NUA/R	EKLIP76
BDE(82)=AM(34)*NUA/R	

AM(44)=AM(42)-AM(33)*BDE(19)	EKLIP77
AM(45)=AM(43)-AM(33)*BDE(20)	EKLIP78
AM(46)=BDE(81)-AM(33)*BDE(21)	EKLIP79
AM(47)=DR*(CA/4.)*(HB**2)	EKLIP80
AM(48)=AM(43)-AM(34)*BDE(19)	EKLIP81
AM(49)=AM(47)-AM(34)*BDE(20)	EKLIP82
AM(50)=BDE(82)-AM(34)*BDE(21)	EKLIP83
BDE(83)=(H/12.)*(BDE(22)*BDE(19))	EKLIP84
BDE(84)=(H/12.)*(BDE(22)*BDE(20))	EKLIP85
BDE(85)=AM(39)*(H/12.)*(BDE(22)*BDE(21)-BDE(23))	EKLIP86
	EKLIP87
DO 1 I=1,6	EKLIP88
V(I)=0.0	EKLIP89
DO 1 J=1,7	EKLIP90
EN(I,J)=0.0	EKLIP91
1 CONTINUE	EKLIP92
EN(1,1)=AC(1)*(AM(44)*BDE(12)-AM(45)*AM(30))+AM(46)	EKLIP93
EN(1,2)=C.0	EKLIP94
EN(1,3)=AM(44)*(AC(2)*BDE(17)-AC(3)*BDE(18))+AM(45)*(AC(3)*BDE(11))	EKLIP95
1-AC(2)*BDE(10))+AM(46)	EKLIP96
EN(1,4)=AM(44)*(AC(3)*BDE(17)+AC(2)*BDE(18))-AM(45)*(AC(3)*BDE(10))	EKLIP97
1+AC(2)*BDE(11))	EKLIP98
EN(1,5)=EN(1,3)	EKLIP99
EN(1,6)=-EN(1,4)	EKLIP
EN(2,1)=AC(1)*(BDE(12)*AM(48)-AM(30)*AM(49))+AM(50)	EKLIP
EN(2,2)=0.0	EKLIP
EN(2,2)=EN(2,1)	EKLIP
EN(2,3)=AM(46)*(-AC(3)*BDE(18)+AC(2)*BDE(17))+AM(49)*(AC(3)*BDE(11))	EKLIP
1-AC(2)*BDE(10))+AM(50)	EKLIP
EN(2,4)=AM(46)*(AC(3)*BDE(17)+AC(2)*BDE(18))-AM(49)*(AC(3)*BDE(10))	EKLIP
1+AC(2)*BDE(11))	EKLIP
EN(2,5)=EN(2,3)	EKLIP
EN(2,6)=-EN(2,4)	EKLIP
EN(3,1)=0.0	EKLIP
EN(3,2)=BDE(85)*AC(1)+BDE(12)*(BDE(83)*AC(1)**2-GAMA)-AM(30)*(BDE(184)*AC(1)**2-GAMB)	EKLIP
EN(3,3)=BDE(85)*AC(2)-BDE(10)*(GAMB+BDE(84)*AM(1))+BDE(11)*(BDE(84)*AM(2))+BDE(17)*(GAMA+AM(1)*BDE(83))-BDE(18)*(BDE(83)*AM(2))	EKLIP
EN(3,4)=AC(3)*BDE(85)-BDE(11)*(GAMB+BDE(84)*AM(1))-BDE(10)*(BDE(84)*AM(2))+BDE(18)*(GAMA+AM(1)*BDE(83))+BDE(17)*(BDE(83)*AM(2))	EKLIP
EN(3,5)=-EN(3,3)	EKLIP
EN(3,6)=EN(3,4)	EKLIP
EN(4,1)=EN(1,1)*COS(P6)	EKLIP
EN(4,2)=EN(1,1)*SIN(P6)	EKLIP
EN(4,3)=EN(1,3)*BDE(31)-EN(1,4)*BDE(32)	EKLIP
EN(4,4)=EN(1,3)*BDE(32)+EN(1,4)*BDE(31)	EKLIP
EN(4,5)=EN(1,5)*BDE(33)-EN(1,6)*BDE(34)	EKLIP
EN(4,6)=EN(1,5)*BDE(34)+EN(1,6)*BDE(33)	EKLIP
EN(5,1)=EN(2,1)*COS(P6)	EKLIP
EN(5,2)=EN(2,1)*SIN(P6)	EKLIP
EN(5,3)=EN(2,3)*BDE(31)-EN(2,4)*BDE(32)	EKLIP
EN(5,4)=EN(2,3)*BDE(32)+EN(2,4)*BDE(31)	EKLIP
EN(5,5)=EN(2,5)*BDE(33)-EN(2,6)*BDE(34)	EKLIP
EN(5,6)=EN(2,5)*BDE(34)+EN(2,6)*BDE(33)	EKLIP
EN(6,1)=-EN(3,1)*SIN(P6)	EKLIP
EN(6,2)=EN(3,1)*COS(P6)	EKLIP
EN(6,3)=EN(3,3)*BDE(31)-EN(3,4)*BDE(32)	EKLIP
EN(6,4)=EN(3,3)*BDE(32)+EN(3,4)*BDE(31)	EKLIP
EN(6,5)=EN(3,5)*BDE(33)-EN(3,6)*BDE(34)	EKLIP
EN(6,6)=EN(3,5)*BDE(34)+EN(3,6)*BDE(33)	EKLIP

EN(1,7)=AM(33)*BT(20)/CAPCB*BDE(81)*BT(9)*RT(18)-(HA/2.)*(BT(5)/(1	EKLIP	
1.-NUA))-BT(12)/(1.-YUA)*(HA/2.)*RT(21)*AM(46)*RDE(49)	EKLIP	
EN(2,7)=AM(34)*BT(20)/CAPCB*RDE(82)*BT(9)*RT(19)-(HA/2.)*(BT(5)/(1	EKLIP	
1.-NUA))-PT(13)/(1.-NUR)*(HB/2.)*BT(22)*AM(50)*BDE(49)	EKLIP	
EN(1,7)=-EN(1,7)	EKLIP	
EN(2,7)=-EN(2,7)	EKLIP	
EN(3,7)=LOAD1	EKLIP	
EN(4,7)=EN(1,7)	EKLIP	
EN(5,7)=EN(2,7)	EKLIP	
EN(6,7)=EN(3,7)	EKLIP	
WRITE (9,3) (BDE(I),I=1,100)	EKLIP	W
WRITE (9,5)	EKLIP	W
WRITE (9,4) (AM(J),J=1,50)	EKLIP	W
WRITE (9,12)	EKLIP	W
WRITE (9,6) ((EN(I,J),J=1,7),I=1,6)	EKLIP	W
CALL FINGLE (EN,V)	EKLIP	
WRITE (9,7) (I,V(I),I=1,6)	EKLIP	W
	EKLIP	
AM(12)=BT(20)/CAPCB-BDE(21)*BDE(49)	EKLIP	
PP1=AC(1)*ELL/2.	EKLIP	
PP2=AC(2)*ELL/2.	EKLIP	
PP3=AC(3)*ELL/2.	EKLIP	
CALL CHECK2 (V,AC,PP1,PP2,PP3)	EKLIP	
WRITE (9,7) (I,V(I),I=1,6)	EKLIP	W
	EKLIP	
AL(1)=AM(30)*V(1)	EKLIP	
AL(2)=-AM(30)*V(2)	EKLIP	
AL(3)=-BDE(10)*V(3)-BDE(11)*V(4)	EKLIP	
AL(4)=BDE(11)*V(3)-RDE(10)*V(4)	EKLIP	
AL(5)=BDE(10)*V(5)-RDE(11)*V(6)	EKLIP	
AL(6)=BDE(11)*V(5)+RDE(10)*V(6)	EKLIP	
AL(7)=-BDE(12)*V(1)	EKLIP	
AL(8)=BDE(12)*V(2)	EKLIP	
AL(7)=BDE(12)*V(1)	EKLIP	
AL(8)=-RDE(12)*V(2)	EKLIP	
AL(9)=BDE(17)*V(3)+BDE(18)*V(4)	EKLIP	
AL(10)=-RDE(19)*V(3)+BDE(17)*V(4)	EKLIP	
AL(11)=-RDE(17)*V(5)+RDE(18)*V(6)	EKLIP	
AL(12)=-BDE(18)*V(5)-BDE(17)*V(6)	EKLIP	
WRITE (9,8)	EKLIP	W
WRITE (9,9) (AL(I),I=1,12)	EKLIP	W
	EKLIP	
DO 2 KF=1,11	EKLIP	
X=DELPH(KF)	EKLIP	
IF (X.GT.ELL) GO TO 2	EKLIP	
P1=AC(1)*X	EKLIP	
P2=AC(2)*X	EKLIP	
P3=AC(3)*X	EKLIP	
	EKLIP	
CALL DIFF (P1,P2,P3)	EKLIP	
	EKLIP	
WW=V(1)*COS(P1)+V(2)*SIN(P1)+EXP(P2)*(V(3)*COS(P3)+V(4)*SIN(P3))+E	EKLIP	
XP(-P2)*(V(5)*COS(P3)+V(6)*SIN(P3))+RDE(49)	EKLIP	
DWW=V(1)*RDE(181)+V(2)*RDE(191)+V(3)*RDE(121)+V(4)*BDE(131)+V(5)*B	EKLIP	


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COMMON TEMP,EA,EAC,NUA,NUAC,EB,EBG,NUB,NUBC,HA,HB,R,G3A,G3B,CA,CB,EKLIP 4
1DA,DB,H,CACH,CAPCB,NUNU,GAMA,GAMB,K(40),A(55),G(7),B(30),FNTOA,FNEKLIP 5
2TOB,RDE(600),LOAD1,AM(5),LOAD2,EN(6,7),FNXA,FNXH,FMXA,FMXB,FNX,FMEKLIP 6
3X,FYDA,FYOB,AC(3),KLIP,ELL,DELPH(20),V(6),AL(18),WW,DWW,DDWW,WA,DWEKLIP 7
4B,DDWB,DDWB,WA,DWA,WU,DWU,DDWU,TAU,DTAU,WPJ,WPA,DWPA,WUA,DWEKLIP 8
SUA EKLIP 9
REAL NUA,NUAC,NUB,NUBC,NUNU,K,LOAD1,LOAD2 EKLIP10
BDE(1)=A(44)*(AC(1)**3)+A(45)*AC(1) EKLIP11
BDE(2)=A(41)*(AC(1)**4)+A(42)*(AC(1)**2)+A(43) EKLIP12
BDE(3)=BDE(1)/BDE(2) EKLIP13
EKLIP14
BDE(4)=A(44)*(AC(2)**3)-A(45)*AC(2) EKLIP15
BDE(5)=A(41)*(AC(2)**4)-A(42)*(AC(2)**2)+A(43) EKLIP16
BDE(6)=BDE(4)/BDE(5) EKLIP17
EKLIP18
BDE(7)=A(44)*(AC(3)**3)-A(45)*AC(3) EKLIP19
BDE(8)=A(41)*(AC(3)**4)-A(42)*(AC(3)**2)+A(43) EKLIP20
BDE(9)=BDE(7)/BDE(8) EKLIP21
EKLIP22
BDE(10)=BDE(3)*(A(21)*(AC(1)**2)+A(22))-A(24)*AC(1) EKLIP23
BDE(11)=A(23)*(AC(1)**2) EKLIP24
BDE(12)=BDE(10)/BDE(11) EKLIP25
EKLIP26
BDE(13)=BDE(6)*(A(21)*(AC(2)**2)-A(22))-A(24)*AC(2) EKLIP27
BDE(14)=A(23)*(AC(2)**2) EKLIP28
BDE(15)=BDE(13)/BDE(14) EKLIP29
EKLIP30
BDE(16)=BDE(9)*(A(21)*(AC(3)**2)-A(22))-A(24)*AC(3) EKLIP31
BDE(17)=A(23)*(AC(3)**2) EKLIP32
BDE(18)=BDE(16)/BDE(17) EKLIP33
EKLIP34
P4=AC(1)*ELL EKLIP35
P5=AC(2)*ELL EKLIP36
P6=AC(3)*ELL EKLIP37
EKLIP38
EN(1,1)=(AM(44)*BDE(12)-AM(45)*BDE(3))*AC(1)+AM(46) EKLIP39
EN(1,2)=EN(1,1) EKLIP40
EN(1,3)=(AM(44)*BDE(15)-AM(45)*BDE(6))*AC(2)+AM(46) EKLIP41
EN(1,4)=0.0 EKLIP42
EN(1,5)=(AM(44)*BDE(18)-AM(45)*BDE(9))*AC(3)+AM(46) EKLIP43
EN(1,6)=0.0 EKLIP44
EN(4,1)=EN(1,1)*EXP(P4) EKLIP45
EN(4,2)=EN(1,2)*EXP(-P4) EKLIP46
EN(4,3)=EN(1,3)*COS(P5) EKLIP47
EN(4,4)=EN(1,3)*SIN(P5) EKLIP48
EN(4,5)=EN(1,5)*COS(P6) EKLIP49
EN(4,6)=EN(1,5)*SIN(P6) EKLIP50
EKLIP51
EN(2,1)=(AM(44)*BDE(12)-AM(49)*BDE(3))*AC(1)+AM(50) EKLIP52
EN(2,2)=EN(2,1) EKLIP53
EN(2,3)=(AM(44)*BDE(15)-AM(49)*BDE(6))*AC(2)+AM(50) EKLIP54
EN(2,4)=0.0 EKLIP55
EN(2,5)=(AM(44)*BDE(18)-AM(49)*BDE(9))*AC(3)+AM(50) EKLIP56
EN(2,6)=0.0 EKLIP57
EN(5,1)=EN(2,1)*EXP(P4) EKLIP58
EN(5,2)=EN(2,2)*EXP(-P4) EKLIP59
EN(5,3)=EN(2,3)*COS(P5) EKLIP60
EN(5,4)=EN(2,3)*SIN(P5) EKLIP61
EN(5,5)=EN(2,5)*COS(P6) EKLIP62
EN(5,6)=EN(2,5)*SIN(P6) EKLIP63

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1(5)*BDE(221)+AL(6)*BDE(231) EKLIP
DDWB=AL(1)*BDE(102)+AL(2)*BDE(112)+AL(3)*BDE(202)+AL(4)*BDE(212)+AEKLIP
1L(5)*BDE(222)+AL(6)*BDE(232) EKLIP
DDWB=AL(1)*BDE(103)+AL(2)*BDE(113)+AL(3)*BDE(203)+AL(4)*BDE(213)+EKLIP
1AL(5)*BDE(223)+AL(6)*BDE(233) EKLIP
WA=AL(7)*EXP(P1)+AL(8)*EXP(-P1)+AL(9)*COS(P2)+AL(10)*SIN(P2)+AL(11)EKLIP
1)*COS(P3)+AL(12)*SIN(P3) EKLIP
DWA=AL(7)*BDE(101)+AL(8)*BDE(111)+AL(9)*BDE(201)+AL(10)*BDE(211)+AEKLIP
1L(11)*BDE(221)+AL(12)*BDE(231) EKLIP
DDWA=AL(7)*BDE(102)+AL(8)*BDE(112)+AL(9)*BDE(202)+AL(10)*BDE(212)+EKLIP
1AL(11)*BDE(222)+AL(12)*BDE(232) EKLIP
DDWA=AL(7)*BDE(103)+AL(8)*BDE(113)+AL(9)*BDE(203)+AL(10)*BDE(213)+EKLIP
1AL(11)*BDE(223)+AL(12)*BDE(233) EKLIP
FNTW=(V(1)*EXP(P1)-V(2)*EXP(-P1))/AC(1)+(V(3)*COS(P2)-V(4)*SIN(P2)EKLIP
1)/AC(2)+(V(5)*SIN(P3)-V(6)*COS(P3))/AC(3)+BDE(49)*X EKLIP
IF (KF.EQ.1) CALL UC OFF (FNTW,CONST,0) EKLIP
WU=-BDE(19)*WA-BDE(20)*WB-BDE(21)*FNTW+IRT(20)/CAPCB)*X+CONST EKLIP
DWU=-BDE(19)*DWA-BDE(20)*DWB-BDE(21)*WW+HT(20)/CAPCH EKLIP
DDWU=-BDE(19)*DDWA-BDE(20)*DDWB-BDE(21)*DDWW EKLIP
DDWU=-BDE(19)*DDWA-BDE(20)*DDWB-BDE(21)*DDWW EKLIP
TAU=LOAD1-BDE(22)*DDWU-BDE(23)*LWW EKLIP
DTAU=-BDE(22)*DDWU-BDE(23)*DDWW EKLIP
WPJ=LCAG2-BDE(24)*DDWB-BDE(25)*DTAU+BDE(23)*DWU+BDE(26)*WW EKLIP
CALL RSLT (X) EKLIP
CALL PRINT (X) EKLIP
IF (X.EQ.ABS(ELL/2.)) CALL VINSON (X,EA,EB,HA,HB,G3A,G3B,FNXB) EKLIP
1 CONTINUE EKLIP
RETURN EKLIP
2 FORMAT (1X,9HBDE ARE / (10E12.5)) EKLIP
3 FORMAT (1X,20H EN(1,J) ARE ) EKLIP
4 FORMAT (1X,7E11.4) EKLIP
5 FORMAT (4(1X,2HV(11,2H)=E12.5)/2(1X,2HV(11,2H)=E12.5)) EKLIP
6 FORMAT (1X,20H AL(1) ARE ) EKLIP
7 FORMAT (6E12.5) EKLIP
END EKLIP
SUBROUTINE EKLIPS EKLIP -
DIMENSION JIP(11) EKLIP 1
COMMON DLTEMP EKLIP 2
COMMON TEMP,EA,EAC,NUA,NUAC,EB,EBC,NUB,NUBC,HA,HB,R,G3A,G3B,CA,CC,EKLIP 3
1DA,DB,H,CACB,CAPCB,PIUW,GAMA,GAMB,K(40),A(55),G(7),HT(30),FNTCA,FNEKLIP 4
2TOB,BDE(600),LOAD1,AM(50),LOAD2,EN(6,7),FNXA,FNXB,FNXA,FNXB,FNA,FNEKLIP 5
3X,FNOA,FNCB,AC(3),KLIP,ELL,DELPH(20),V(6),AL(14),WW,DWW,DDWW,HB,DWEKLIP 6
4B,DWB,DDWB,WA,DWA,WU,DWU,DDWU,TAU,DTAU,WPJ,HA,DWA,WUA,DWEKLIP 7
5UA EKLIP 8
READ NUA,NUAC,NUB,NUBC,NUN,K,LOAD1,LOAD2 EKLIP 9
BDE(1)=A(44)*(AC(1)*03)+A(45)*AC(1) EKLIP10
BDE(2)=A(41)*(AC(1)*04)+A(42)*(AC(1)*02)+A(43) EKLIP11
BDE(3)=BDE(1)/BDE(2) EKLIP12
BDE(4)=A(44)*(AC(2)*03)+A(45)*AC(2) EKLIP13
BDE(5)=A(41)*(AC(2)*04)+A(42)*(AC(2)*02)+A(43) EKLIP14
BDE(6)=BDE(4)/BDE(5) EKLIP15
BDE(7)=A(44)*(AC(3)*03)+A(45)*AC(3) EKLIP16
BDE(8)=A(41)*(AC(3)*04)+A(42)*(AC(3)*02)+A(43) EKLIP17
BDE(9)=BDE(7)/BDE(8) EKLIP18
EKLIP19
EKLIP20
EKLIP21
EKLIP22

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BDE(10)=BDE(3)*(A(21)*(AC(1)**2)+A(22))-A(24)*AC(1)
 BDE(11)=A(23)*(AC(1)**2)
 BDE(12)=BDE(10)/BDE(11)

BDE(13)=BDE(6)*(A(21)*(AC(2)**2)+A(22))-A(24)*AC(2)
 BDE(14)=A(23)*(AC(2)**2)
 BDE(15)=BDE(13)/BDE(14)

BDE(16)=BDE(4)*(A(21)*(AC(3)**2)+A(22))-A(24)*AC(3)
 BDE(17)=A(23)*(AC(3)**2)
 BDE(18)=BDE(16)/BDE(17)
 P4=AC(1)*ELL
 P5=AC(2)*ELL
 P6=AC(3)*ELL

EN(1,1)=(AM(44)*BDE(12)-AM(45)*BDE(3))*AC(1)+AM(46)
 EN(1,2)=EN(1,1)
 EN(1,3)=(AM(44)*BDE(15)-AM(45)*BDE(6))*AC(2)+AM(46)
 EN(1,4)=EN(1,3)
 EN(1,5)=(AM(44)*BDE(18)-AM(45)*BDE(9))*AC(3)+AM(46)
 EN(1,6)=0.0
 EN(4,1)=EN(1,1)*EXP(P4)
 EN(4,2)=EN(1,2)*EXP(-P4)
 EN(4,3)=EN(1,3)*EXP(P5)
 EN(4,4)=EN(1,4)*EXP(-P5)
 EN(4,5)=EN(1,5)*COS(P6)
 EN(4,6)=EN(1,5)*SIN(P6)

EN(2,1)=(AM(48)*BDE(12)-AM(49)*BDE(3))*AC(1)+AM(50)
 EN(2,2)=EN(2,1)
 EN(2,3)=(AM(48)*BDE(15)-AM(49)*BDE(6))*AC(2)+AM(50)
 EN(2,4)=EN(2,3)
 EN(2,5)=(AM(48)*BDE(18)-AM(49)*BDE(9))*AC(3)+AM(50)
 EN(2,6)=0.0
 EN(5,1)=EN(2,1)*EXP(P4)
 EN(5,2)=EN(2,2)*EXP(-P4)
 EN(5,3)=EN(2,3)*EXP(P5)
 EN(5,4)=EN(2,4)*EXP(-P5)
 EN(5,5)=EN(2,5)*COS(P6)
 EN(5,6)=EN(2,5)*SIN(P6)

EN(3,1)=BDE(15)*AC(1)-BDE(3)*(BDE(14)*(AC(1)**2)+GAMB)+BDE(12)*(BDE(14)
 1E(83)*(AC(1)**2)+GAMA)
 EN(3,2)=EN(3,1)
 EN(3,3)=BDE(15)*AC(2)-BDE(6)*(BDE(14)*(AC(2)**2)+GAMB)+BDE(15)*(BDE(14)
 1E(83)*(AC(2)**2)+GAMA)
 EN(3,4)=-EN(3,3)
 EN(3,5)=0.0
 EN(3,6)=BDE(15)*AC(3)+BDE(18)*(BDE(13)*(AC(3)**2)-GAMA)-BDE(9)*(BDE(13)
 1E(84)*(AC(3)**2)-GAMB)
 EN(6,1)=EN(3,1)*EXP(P4)
 EN(6,2)=EN(3,2)*EXP(-P4)
 EN(6,3)=EN(3,3)*EXP(P5)
 EN(6,4)=EN(3,4)*EXP(-P5)
 EN(6,5)=-EN(3,6)*SIN(P6)
 EN(6,6)=EN(3,6)*COS(P6)

WRITE (9,2) (BDE(I), I=1,20)
 WRITE (9,3)

EKLIP23
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 EKLIP78
 EKLIP79
 EKLIP80
 EKLIP81W
 EKLIP82W


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DWU=-BDE(19)*DWA-BDE(20)*DWA-BDE(21)*WW*BT(20)/CAPCB
DDWU=-BDE(19)*DDWA-BDE(20)*DDWB-BDE(21)*DWW
DDDWU=-BDE(19)*DDDWA-BDE(20)*DDDWB-BDE(21)*DDWW
TAU=LOAD1-BDE(22)*DDWU-BDE(23)*DWW
DTAU=-BDE(22)*DDDWU-BDE(23)*DDWW
WPJ=LOAD2-BDE(24)*DDWB-BDE(25)*DTAU*BDE(23)*DWU*BDE(26)*WW
CALL RSLT (X)
CALL PRINT (X)
QA=(HA/12.)*TAU+GAMA*(WA+DWW)
QB=(HB/12.)*TAU+GAMB*(WB+DWW)
FQ=QA+QB
WRITE (9,6) QA,QB,FQ
IF (X.EQ.ABS(ELL/2.)) CALL VINSON (X,EA,EB,HA,HB,G3A,G3B,FNB)
1 CONTINUE
RETURN

2 FORMAT (1X,9H8DE ARE /10E12.5)
3 FORMAT (1X,20H EN(1,J) ARE . )
4 FORMAT (1X,7E11.4)
5 FORMAT (4(1X,2HV(11,2H)=E12.5)/2(1X,2HV(11,2H)=E12.5))
6 FORMAT (1X,20H AL(1) ARE )
7 FORMAT (6E12.5)
8 FORMAT (1X,3HQA=E12.5,2X,3HQB=E12.5,2X,5HQBAR=E12.5/)
END
SUBROUTINE EKLIP6
DIMENSION JIP(11)
COMMON DTEMP
COMMON TEMP,EA,EAC,NUA,NUAC,EB,EBC,NUB,NUBC,HA,HB,R,G3A,G3B,CA,CB,EKLIP 4
BDA,DR,H,CACB,CAPCB,NUMU,GAMA,GAMB,K(40),A(55),C(7),ST(30),FYTCA,FNEKLIP 5
2T0B,PDE(600),LOAD1,AM(50),LOAD2,EN(6,7),FNXA,FNXB,FMXA,FMXB,FNX,FMEKLIP 6
3X,FNCA,FNOB,AC(3),KLIP,ELL,DELP(20),V(6),AL(18),WW,DWW,DDWW,WB,DWEKLIP 7
4B,DDWB,DDWB,WA,DWA,WU,DWU,DDWU,DDWU,TAU,DTAU,WPJ,WHA,DHWA,WUA,DWEKLIP 8
3UA
REAL NUA,NUAC,NUR,NUBC,NUMU,K,LOAD1,LOAD2
CASE OF THREE REAL ROOTS,UNEQUAL,POSITIVE
BDE(1)=A(44)*AC(1)**3+A(45)*AC(1)
BDE(2)=A(41)*AC(1)**4+A(42)*AC(1)**2+A(43)
BDE(3)=BDE(1)/BDE(2)

BDE(4)=A(44)*AC(2)**3+A(45)*AC(2)
BDE(5)=A(41)*AC(2)**4+A(42)*AC(2)**2+A(43)
BDE(6)=BDE(4)/BDE(5)

BDE(7)=A(44)*AC(3)**3+A(45)*AC(3)
BDE(8)=A(41)*AC(3)**4+A(42)*AC(3)**2+A(43)
BDE(9)=BDE(7)/BDE(8)

BDE(10)=BDE(3)*A(21)*AC(1)**2+A(22))-A(24)*AC(1)
BDE(11)=A(23)*AC(1)**2)
BDE(12)=BDE(10)/BDE(11)

BDE(13)=BDE(6)*A(21)*AC(2)**2+A(22))-A(24)*AC(2)
BDE(14)=A(23)*AC(2)**2)
BDE(15)=BDE(13)/BDE(14)

BDE(16)=BDE(9)*A(21)*AC(3)**2+A(22))-A(24)*AC(3)
BDE(17)=A(23)*AC(3)**2)
BDE(18)=BDE(16)/BDE(17)
EKLIP 36

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P4=AC(1)*ELL	EKLIP37
P5=AC(2)*ELL	EKLIP38
P6=AC(3)*ELL	EKLIP39
EN(1,1)=(AM(44)*RDE(12)-AM(45)*BDE(3))*AC(1)+AM(46)	EKLIP40
EN(1,2)=EN(1,1)	EKLIP41
EN(1,3)=(AM(44)*RDE(15)-AM(45)*BDE(6))*AC(2)+AM(46)	EKLIP42
EN(1,4)=EN(1,3)	EKLIP43
EN(1,5)=(AM(44)*RDE(18)-AM(45)*BDE(9))*AC(3)+AM(46)	EKLIP44
EN(1,6)=EN(1,5)	EKLIP45
EN(4,1)=EN(1,1)*EXP(P4)	EKLIP46
EN(4,2)=EN(1,2)*EXP(-P4)	EKLIP47
EN(4,3)=EN(1,3)*EXP(P5)	EKLIP48
EN(4,4)=EN(1,4)*EXP(-P5)	EKLIP49
EN(4,5)=EN(1,5)*EXP(P6)	EKLIP50
EN(4,6)=EN(1,6)*EXP(-P6)	EKLIP51
EN(2,1)=(AM(48)*BDE(12)-AM(49)*BDE(3))*AC(1)+AM(50)	EKLIP52
EN(2,2)=EN(2,1)	EKLIP53
EN(2,3)=(AM(48)*BDE(15)-AM(49)*BDE(6))*AC(2)+AM(50)	EKLIP54
EN(2,4)=EN(2,3)	EKLIP55
EN(2,5)=(AM(48)*BDE(18)-AM(49)*BDE(9))*AC(3)+AM(50)	EKLIP56
EN(2,6)=EN(2,5)	EKLIP57
EN(5,1)=EN(2,1)*EXP(P4)	EKLIP58
EN(5,2)=EN(2,2)*EXP(-P4)	EKLIP59
EN(5,3)=EN(2,3)*EXP(P5)	EKLIP60
EN(5,4)=EN(2,4)*EXP(-P5)	EKLIP61
EN(5,5)=EN(2,5)*EXP(P6)	EKLIP62
EN(5,6)=EN(2,6)*EXP(-P6)	EKLIP63
EN(3,1)=BDE(85)*AC(1)-BDE(3)*(BDE(84)*(AC(1)**2)+GAMB)*BDE(12)*(BDE(83)*(AC(1)**2)+GAMA)	EKLIP64
EN(3,2)=EN(3,1)	EKLIP65
EN(3,3)=BDE(85)*AC(2)-BDE(6)*(BDE(84)*(AC(2)**2)+GAMB)*BDE(15)*(BDE(83)*(AC(2)**2)+GAMA)	EKLIP66
EN(3,4)=-EN(3,3)	EKLIP67
EN(3,5)=BDE(85)*AC(3)-BDE(9)*(BDE(84)*(AC(3)**2)+GAMB)*BDE(18)*(BDE(83)*(AC(3)**2)+GAMA)	EKLIP68
EN(3,6)=-EN(3,5)	EKLIP69
EN(6,1)=EN(3,1)*EXP(P4)	EKLIP70
EN(6,2)=EN(3,2)*EXP(-P4)	EKLIP71
EN(6,3)=EN(3,3)*EXP(P5)	EKLIP72
EN(6,4)=EN(3,4)*EXP(-P5)	EKLIP73
EN(6,5)=EN(3,5)*EXP(P6)	EKLIP74
EN(6,6)=EN(3,6)*EXP(-P6)	EKLIP75
AL(1)=-BDE(3)*V(1)	EKLIP76
AL(2)=BDE(3)*V(2)	EKLIP77
AL(3)=-BDE(6)*V(3)	EKLIP78
AL(4)=BDE(6)*V(4)	EKLIP79
AL(5)=-BDE(9)*V(5)	EKLIP80
AL(6)=BDE(9)*V(6)	EKLIP81
	EKLIP82
	EKLIP83
	EKLIP84
	EKLIP85
	EKLIP86
	EKLIP87
	EKLIP88
	EKLIP89
	EKLIP90
	EKLIP91
	EKLIP92
	EKLIP93
	EKLIP94
	EKLIP95
	EKLIP96

BDE(104)=BDE(103)*AC(1)

BDE(111)=-AC(1)*EXP(-P1)
BDE(112)=-BDE(111)*AC(1)
BDE(113)=-BDE(112)*AC(1)
BDE(114)=-BDE(113)*AC(1)

BDE(121)=EXP(P2)*(AC(2)*COS(P3)-AC(3)*SIN(P3))
BDE(122)=EXP(P2)*(AM(1)*COS(P3)-AM(2)*SIN(P3))
BDE(123)=EXP(P2)*(AM(3)*COS(P3)-AM(4)*SIN(P3))
BDE(124)=EXP(P2)*(AM(5)*COS(P3)-AM(6)*SIN(P3))

BDE(131)=EXP(P2)*(AC(3)*COS(P3)+AC(2)*SIN(P3))
BDE(132)=EXP(P2)*(AM(2)*COS(P3)+AM(1)*SIN(P3))
BDE(133)=EXP(P2)*(AM(4)*COS(P3)+AM(3)*SIN(P3))
BDE(134)=EXP(P2)*(AM(6)*COS(P3)+AM(5)*SIN(P3))

BDE(141)=-EXP(-P2)*(AC(2)*COS(P3)+AC(3)*SIN(P3))
BDE(142)=-EXP(-P2)*(AM(1)*COS(P3)+AM(2)*SIN(P3))
BDE(143)=-EXP(-P2)*(AM(3)*COS(P3)+AM(4)*SIN(P3))
BDE(144)=-EXP(-P2)*(AM(5)*COS(P3)+AM(6)*SIN(P3))

BDE(151)=-EXP(-P2)*(AC(3)*COS(P3)-AC(2)*SIN(P3))
BDE(152)=-EXP(-P2)*(AM(2)*COS(P3)-AM(1)*SIN(P3))
BDE(153)=-EXP(-P2)*(AM(4)*COS(P3)-AM(3)*SIN(P3))
BDE(154)=-EXP(-P2)*(AM(6)*COS(P3)-AM(5)*SIN(P3))

BDE(161)=AC(2)*EXP(P2)
BDE(162)=AC(2)*BDE(161)
BDE(163)=AC(2)*BDE(162)
BDE(164)=AC(2)*BDE(163)

BDE(171)=-AC(2)*EXP(-P2)
BDE(172)=-AC(2)*BDE(171)
BDE(173)=-AC(2)*BDE(172)
BDE(174)=-AC(2)*BDE(173)

BDE(181)=-AC(1)*SIN(P1)
BDE(182)=-AC(1)*2)*COS(P1)
BDE(183)=-AC(1)*3)*SIN(P1)
BDE(184)=-AC(1)*4)*COS(P1)

BDE(191)=AC(1)*COS(P1)
BDE(192)=-AC(1)*2)*SIN(P1)
BDE(193)=-AC(1)*3)*COS(P1)
BDE(194)=-AC(1)*4)*SIN(P1)

BDE(201)=-AC(2)*SIN(P2)
BDE(202)=-AC(2)*2)*COS(P2)
BDE(203)=-AC(2)*3)*SIN(P2)
BDE(204)=-AC(2)*4)*COS(P2)

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BDE(211)=AC(2)*COS(P2)	74
BDE(212)=- (AC(2)**2)*SIN(P2)	75
BDE(213)=- (AC(2)**3)*COS(P2)	76
BDE(214)= (AC(2)**4)*SIN(P2)	77
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BDE(221)=-AC(3)*SIN(P3)	82
BDE(222)=- (AC(3)**2)*COS(P3)	83
BDE(223)= (AC(3)**3)*SIN(P3)	84
BDE(224)= (AC(3)**4)*COS(P3)	85
	86
	87
BDE(231)=AC(3)*COS(P3)	88
BDE(232)=- (AC(3)**2)*SIN(P3)	89
BDE(233)=- (AC(3)**3)*COS(P3)	90
BDE(234)= (AC(3)**4)*SIN(P3)	91
	92
	93
BDE(241)=AC(3)*EXP(P3)	94
BDE(242)=AC(3)*BDE(241)	95
BDE(243)=AC(3)*BDE(242)	96
BDE(244)=AC(3)*BDE(243)	97
	98
	99
BDE(251)=-AC(3)*EXP(-P3)	100
BDE(252)=-AC(3)*BDE(251)	101
BDE(253)=-AC(3)*BDE(252)	102
BDE(254)=-AC(3)*BDE(253)	103
	104
RETURN	105
END	106-
SUBROUTINE POLYR (N,COEFF,ROOTS,D)	***** 1
DIMENSION A(51,3), IA(51,3), ROOTS(2,N), D(1), COEFF(1)	2
INTEGER DEGREE	3
DEGREE=N	4
N1=DEGREE+1	5
N=10	6
MMAX=15	7
DELTA=0.0001	8
EPSILON=0.000001	9
DO 1 I=1,N1	10
A(I,1)=COEFF(I)	11
IA(I,1)=0	12
CALL SCALE (A(I,1),IA(I,1))	13
1 CONTINUE	14
CALL RSSR (A,IA,ROOTS,DEGREE,M,MMAX,DELTA,EPSILON,D)	15
IF (N1-(DEGREE+1)) 3,3,2	16
2 RETURN	17
3 PRINT 4	18P
RETURN	19
	20
4 FORMAT (21HOSOME ROOTS NOT FOUND)	21
END	22-
SUBROUTINE RSSR (A,IA,ROOTS,DEGREE,M,MMAX,DELTA,EPSILON,D)	***** 1
DIMENSION A(51,3), IA(51,3), ROOTS(2,50), D(51), ROMCO(50), MROMCO	2
1(50), NONRT(50), MNONRT(50)	3
INTEGER DEGREE	4
N=DEGREE	5

IF (N) 1,1,3	6
1 DEGREE=NCUR	7
2 RETURN	8
3 N1=N+1	9
N2=N1+1	10
DO 5 I=1,N	11
K=N2-I	12
IF (A(K,1)) 6,4,6	13
4 J=N1-I	14
ROOTS(1,J)=0.0	15
ROOTS(2,J)=0.0	16
5 CONTINUE	17
DEGREE=0	18
GO TO 2	19
6 N1=K	20
N=K-1	21
NCUR=N	22
NL=4	23
7 CALL ROOTSQ (A,IA,NCUR,M)	24
CALL REALRT (A,IA,M,NCUR,DELTA,EPSILON,ROMOD,MROMOD,NONRT,MNONRT,N	25
ICO,ROOTS)	26
IF (NCO) 12,12,8	27
8 N1=NCUR+1	28
CALL COMPT (A,IA,ROMOD,ROOTS,M,MNONRT,NONRT,MROMOD,NCO,DELTA,EPSI	29
1LON,NCUR)	30
IF (NCUR) 12,12,9	31
9 IF (NL-NCUR) 11,11,10	32
10 NL=NCUR	33
GO TO 7	34
11 M=M+1	35
IF (MMAX-M) 1,7,7	36
12 CALL RECON (ROOTS,A(1,1),IA(1,1),D,DEGREE)	37
GO TO 1	38
END	39-
SUBROUTINE ROOTSQ (A,IA,NCUR,MM)	***** 1
DIMENSION A(51,3), IA(51,3)	RTSQ 2
N1=NCUR+1	RTSQ 3
DO 1 J=1,N1	RTSQ 4
A(J,2)=A(J,1)	RTSQ 5
IA(J,2)=IA(J,1)	RTSQ 6
A(J,3)=0.0	RTSQ 7
IA(J,3)=0	RTSQ 8
1 CONTINUE	RTSQ 9
DO 10 M=1,MM	RTSQ 10
DO 7 J=1,N1	RTSQ 11
K1=N1-J	RTSQ 12
K2=J-1	RTSQ 13
KM=XMINOF(K1,K2)	RTSQ 14
IF (KM) 2,5,2	RTSQ 15
2 DO 5 L=1,KM	RTSQ 16
LR=XMODF(L,2)	RTSQ 17
JL=J-L	RTSQ 18
JLP=J+L	RTSQ 19
IF (LR) 3,3,4	RTSQ 20
3 X=A(JL,2)+A(JLP,2)	RTSQ 21
IX=IA(JL,2)+IA(JLP,2)	RTSQ 22
CALL SCALE (X,IX)	RTSQ 23
CALL ADD (A(J,3),IA(J,3),X,IX,A(J,3),IA(J,3))	RTSQ 24
GO TO 5	RTSQ 25
4 X=A(JL,2)+A(JLP,2)	RTSQ 26

IX=IA(JL,2)+IA(JLP,2)	RTSQ 27
CALL SCALE (X,IX)	RTSQ 28
CALL SBTRY (A(J,3),IA(J,3),X,IX,A(J,3),IA(J,3))	RTSQ 29
5 CONTINUE	RTSQ 30
A(J,3)=2.0*A(J,3)	RTSQ 31
CALL SCALE (A(J,3),IA(J,3))	RTSQ 32
X=A(J,2)**2	RTSQ 33
IX=IA(J,2)+IA(J,2)	RTSQ 34
CALL SCALE (X,IX)	RTSQ 35
CALL ADD (A(J,3),IA(J,3),X,IX,A(J,3),IA(J,3))	RTSQ 36
JR=XMODF(J,2)	RTSQ 37
IF (JR) 6,6,7	RTSQ 38
6 A(J,3)=-A(J,3)	RTSQ 39
7 CONTINUE	RTSQ 40
IF (MM-M) 10,10,8	RTSQ 41
8 DO 9 J=1,N1	RTSQ 42
A(J,2)=A(J,3)	RTSQ 43
IA(J,2)=IA(J,3)	RTSQ 44
A(J,3)=0.0	RTSQ 45
IA(J,3)=0	RTSQ 46
9 CONTINUE	RTSQ 47
10 CONTINUE	RTSQ 48
RETURN	RTSQ 49
END	RTSQ 50-
SUBROUTINE REALRT (A,IA,M,NCUR,DELTA,EPSILON,ROMOD,MROMOD,MONRT,MN***** 1	***** 2
1 MONRT,NCU,ROUTS)	NRLRT 3
DIMENSION A(51,3), IA(51,3), ROOTS(2,50), ROMOD(50), MROMOD(50),	RLRT 4
1 MONRT(50), MNMONRT(50), RATIO(51), IPIV(51), ARED(50), IARED(50)	RLRT 5
RATIO(1)=1.0	RLRT 6
DO 6 I=2,NCUR	RLRT 7
I1=XMODF(I,2)	RLRT 8
IF (A(I,3)) 2,1,2	RLRT 9
1 RATIO(I)=0.0	RLRT 10
GO TO 6	RLRT 11
2 T=A(I,2)*A(I,2)	RLRT 12
IT=IA(I,2)+IA(I,2)	RLRT 13
CALL SCALE (T,IT)	RLRT 14
T=T/A(I,3)	RLRT 15
IT=IT-IA(I,3)	RLRT 16
IF (IT-2) 3,3,1	RLRT 17
3 IF (IT-2) 1,4,4	RLRT 18
4 CALL UNSCALE (T,IT)	RLRT 19
RATIO(I)=T	RLRT 20
IF (I1) 5,5,6	RLRT 21
5 RATIO(I)=-RATIO(I)	RLRT 22
6 CONTINUE	RLRT 23
RATIO(NCUR+1)=1.0	RLRT 24
IPIV(1)=1	RLRT 25
IPIV(NCUR+1)=1	RLRT 26
DO 9 I=2,NCUR	RLRT 27
X=ABS(RATIO(I)-1.0)	RLRT 28
IF (X-DELTA) 7,8,8	RLRT 29
7 IPIV(I)=1	RLRT 30
GO TO 9	RLRT 31
8 IPIV(I)=0	RLRT 32
9 CONTINUE	RLRT 33
NCUR=NCUR+1	RLRT 34
I1=0	RLRT 35
MULT=0	RLRT 36
I=1	

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14=1
10 I1=I1+1
    I2=I1+1
    MULT=PULT+1
    IF (IPIV(I2)) 10,10,11
11 ROMOD(I4)=A(I2,3)/A(I,3)
    IROMOD=IA(I2,3)-IA(I,3)
    CALL SCALE (ROMOD(I4),IROMOD)
    IF (IROMOD(I4)) 12,13,13
12 ROMOD(I4)=-ROMOD(I4)
13 CALL DOUBLOG (ROMOD(I4),IROMOD,XN,IXN)
    T=2**M*11
    XN=XN/T
    CALL SCALE (XN,IXN)
    CALL DLEXP (XN,IXN,ROMOD(I4),IROMOD)
    IF (IROMOD-74) 14,14,15
14 IF (IROMOD+74) 15,15,16
15 ROMOD(I4)=0.0
    IROMOD=0
    GO TO 17
16 CALL UNSCALE (ROMOD(I4),IROMOD)
17 MROMOD(I4)=MULT
    IF (NCUR+1-12) 19,19,18
18 I=I2
    I4=I4+1
    MULT=0
    I1=0
    GO TO 10
19 Q=0.0
    NCO=0
    DO 28 I=1,I4
        KL=I4+1-I
        W=-ROMOD(KL)
        IS=MROMOD(KL)
        DO 26 J=1,IS
            J=J
20 CALL TEST (A,IA,W,Q,NCUR,ROMOD(KL),EPSILON,K)
    IF (K) 23,23,21
21 ROOTS(1,NCUR)=-W
    ROOTS(2,NCUR)=0.0
    ARED(1)=A(I,1)
    IARED(1)=IA(I,1)
    DO 22 L=2,NCUR
        Y=ARED(L-1)*W
        IY=IARED(L-1)
        CALL SCALE (Y,IY)
        CALL SBTRT (A(L,1),IA(L,1),Y,IY,ARED(L),IARED(L))
        A(L,1)=ARED(L)
        IA(L,1)=IARED(L)
22 CONTINUE
    GO TO 25
23 IF (W) 24,27,27
24 W=-W
    GO TO 20
25 NCUR=NCUR-1
26 CONTINUE
    GO TO 28
27 NCO=NCO+1
    NONRT(NCO)=KL
    MNONRT(NCO)=I5+1-J

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28 CONTINUE	RLAT 97
RETURN	RLAT 98
END	RLAT 99-
SUBROUTINE COMPT (A,IA,ROMOD,ROOTS,M,MNONRT,NOVRT,MRMOD,NCU,DELT)	***** 1
IA, EPSILON, NCUR	***** 2
DIMENSION A(51,3), IA(51,3), ROMOD(50), ROOTS(2,50), SR(51,3), ISR	3
1(51,3), SRMOD(50), SROOTS(2,50), MNONRT(50), NOVRT(50), MSROMOD(5	4
20), NSONRT(49), MSNORT(49), MRMOD(50), U(2), R(2), B(49)	5
DO 28 I=1,NCU	6
JA=NONRT(I)	7
II=MNONRT(I)	8
II=II/2	9
IF (II) 1,1,2	10
1 II=1	11
2 IF (ROMOD(JA)) 3,28,3	12
3 Q=ROMOD(JA)	13
DO 27 J=1,II	14
CALL SUBRES (A,IA,NCUR,SR,ISR,Q)	15
IF (NCUR-4) 5,4,6	16
4 NSCUR=2	17
GO TO 7	18
5 NSCUR=1	19
J2=1	20
GO TO 8	21
6 NSCUR=NCUR-3	22
7 J2=NSCUR	23
8 LL=NSCUR+1	24
IF (NSCUR-1) 9,9,11	25
9 IF (SR(1,1)) 10,12,10	26
10 X=SR(1,1)	27
IX=ISR(1,1)	28
Y=SR(2,1)	29
IY=ISR(2,1)	30
CALL UNSCALE (X,IX)	31
CALL UNSCALE (Y,IY)	32
SROOTS(1,1)=-Y/X	33
NSCUR=0	34
GO TO 13	35
11 CALL ROOTSO (SR,ISR,NSCUR,M)	36
CALL REALT (SR,ISR,M,NSCUR,DELTA,EPSILON,SRMOD,MSROMOD,NSONRT,MS	37
1NORT,NSCO,SROOTS)	38
IF (J2-NSCUR) 12,12,13	39
12 SROOTS(1,J2)=J.J	40
13 SROOTS(1,J2)=SROOTS(1,J2)*ROMOD(JA)	41
T=ROMOD(JA)*ROMOD(JA)	42
IF (SROOTS(1,J2)-T) 14,21,21	43
14 W=SROOTS(1,J2)	44
WE=ROMOD(JA)*ROMOD(JA)	45
CALL TEST (A,IA,W,WE,NCUR,ROMOD(JA),EPSILON,K)	46
IF (K) 20,20,15	47
15 ROOTS(1,NCUR)=-W/2.0	48
T=4.0*WE	49
U=W*W	50
T=T-U	51
IF (T) 16,16,17	52
16 T=-T	53
U=SQRT(T)	54
ROOTS(1,NCUR)=ROOTS(1,NCUR)-U/2.0	55
ROOTS(1,NCUR-1)=-(-W-U)/2.0	56
ROOTS(2,NCUR)=0.0	57

ROOTS(2,NCUR-1)=0.0	58
GO TO 18	59
17 U=SQRT(T)	60
ROOTS(2,NCUR)=U/2.0	61
ROOTS(1,NCUR-1)=ROOTS(1,NCUR)	62
ROOTS(2,NCUR-1)=-ROOTS(2,NCUR)	63
18 D(1)=W	64
D(2)=WE	65
CALL QUADIV (NCUR,A,IA,R,D,B)	66
JX=NCUR-1	67
DO 19 JY=1,JX	68
A(JY,1)=B(JY)	69
IA(JY,1)=0	70
CALL SCALE (A(JY,1),IA(JY,1))	71
19 CONTINUE	72
NCUR=NCUR-2	73
GO TO 27	74
20 W=-W	75
CALL TEST (A,IA,W,WE,NCUR,ROMOD(JA),EPSILON,K)	76
IF (K) 21,21,15	77
21 IF (J2-1) 28,22,24	78
22 IF (J2-1) 26,28,23	79
23 J2=J2-1	80
SROOTS(1,J2)=0.0	81
GO TO 14	82
24 IF (SROOTS(1,J2)-SROOTS(1,J2-1)) 25,26,25	83
25 J2=J2-1	84
GO TO 13	85
26 J2=J2-1	86
GO TO 21	87
27 CONTINUE	88
28 CONTINUE	89
RETURN	90
END	91-
SUBROUTINE TEST (A,IA,W,Q,N,ROMOD,EPSILON,K)	***** 1
DIMENSION A(51,3), IA(51,3), B(3), IB(3), T(2), E(2), C(51)	TEST 2
B(1)=0.0	TEST 3
IX=0	TEST 4
IW=0	TEST 5
IB(1)=0	TEST 6
B(2)=A(1,1)	TEST 7
IB(2)=IA(1,1)	TEST 8
DO 2 I=1,N	TEST 9
X=W*B(2)	TEST 10
IX=IB(2)	TEST 11
CALL SCALE (X,IX)	TEST 12
Y=Q*B(1)	TEST 13
IY=IR(1)	TEST 14
CALL SCALE (Y,IY)	TEST 15
CALL ADD (X,IX,Y,IY,2,IZ)	TEST 16
CALL SBTRT (A(I+1,1),IA(I+1,1),2,IZ,B(3),IB(3))	TEST 17
IF (N-1) 2,2,1	TEST 18
1 B(1)=B(2)	TEST 19
IB(1)=IB(2)	TEST 20
B(2)=B(3)	TEST 21
IB(2)=IB(3)	TEST 22
2 CONTINUE	TEST 23
KOUNT=1	TEST 24
CEPSIL=EPSILON	TEST 25
T(1)=0.0,T(2)=0.0	TEST 26

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N1=N+1
X=2.C*EPSILON
Y=X*ROMOD
E(1)=ROMOD*Y
E(2)=ROMOD*CEPSIL*ROMOD
DO 6 I=1,N1
IF (A(I,1)) 3,4,4
3 C(1)=-A(I,1)*IC*TA(I,1)
GO TO 5
4 C(1)=A(I,1)*IC*TA(I,1)
5 CALL UNSCALE (C(1),IC)
T(1)=T(1)+E(1)*C(1)
T(2)=T(2)+E(2)*C(1)
6 CONTINUE
DIF=T(1)-T(2)
IF (D) 10,7,10
7 IF (R(3)) 8,9,9
8 B(3)=-B(3)
9 IF (IB(3)-74) 10,10,12
10 IF (IR(3)+74) 12,11,11
11 CALL UNSCALE (B(3),IB(3))
IF (DIF-B(3)) 12,12,17
12 K=0
IF (KOUNT-2) 13,14,16
13 IF (J) 15,14,15
14 IF (M) 15,16,16
15 SENSE LIGHT 2
KOUNT=KOUNT+1
16 RETURN
17 K=1
GO TO 16
18 IF (IB(2)-74) 19,19,12
19 IF (IB(2)+74) 12,20,20
20 CALL UNSCALE (B(2),IB(2))
IF (IB(3)-74) 21,21,12
21 IF (IR(3)+74) 12,22,22
22 CALL UNSCALE (B(3),IB(3))
X=Q*B(2)*B(2)
Y=M*B(2)*B(3)
Z=B(3)*B(3)
V=X-Y+Z
IF (V) 23,17,24
23 V=-V
24 DIF=DIF+DIF
IF (DIF-V) 12,17,17
END
SUBROUTINE SURRES (A,IA,N,SR,ISR,ROMOD)
DIMENSION A(51,3), IA(51,3), SR(51,3), ISR(51,3), C(51), A(50,3)
N1=N+1
T=1.0
DO 1 I=1,N
J=N1-I
T=T*ROMOD
C(I)=A(I,J)*T
IC=IA(I,J)
CALL UNSCALE (C(I),IC)
1 CONTINUE
C(N1)=A(N1,1)
IC=IA(N1,1)
CALL UNSCALE (C(N1),IC)

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IF (N-2) 17,17,2	15
2 N2=N-2	16
DO 3 I=1,N2	17
B(1,1)=0.0	18
B(1,2)=0.0	19
3 CONTINUE	20
I=2	21
B(1,2)=C(1)	22
4 B(1,3)=C(1)-B(1,1)	23
DO 5 J=2,N2	24
B(J,3)=-B(J-1,2)-B(J,1)	25
5 CONTINUE	26
IF (N-(3+1)) 8,6,6	27
6 I=I+1	28
DO 7 J=1,N2	29
B(J,1)=B(J,2)	30
B(J,2)=H(J,3)	31
7 CONTINUE	32
GO TO 4	33
8 IF (N-4) 19,9,13	34
9 IF (N-(2+1)) 12,10,10	35
10 I=I+1	36
DO 11 J=1,2	37
B(J,1)=B(J,2)	38
B(J,2)=B(J,3)	39
11 CONTINUE	40
GO TO 4	41
12 B(3,3)=-B(2,2)	42
SR(3,1)=-C(5)+B(1,3)	43
ISR(3,1)=0	44
SR(2,1)=B(2,3)	45
ISR(2,1)=0	46
SR(1,1)=B(3,3)	47
ISR(1,1)=0	48
CALL SCALE (SR(1,1),ISR(1,1))	49
CALL SCALE (SR(2,1),ISR(2,1))	50
CALL SCALE (SR(3,1),ISR(3,1))	51
GO TO 16	52
13 SR(N2,1)=C(N)-B(1,3)	53
ISR(N2,1)=0	54
SR(N2-1,1)=-C(N1)-B(2,3)	55
ISR(N2-1,1)=0	56
CALL SCALE (SR(N2,1),ISR(N2,1))	57
CALL SCALE (SR(N2-1,1),ISR(N2-1,1))	58
IF (N2-2) 16,16,14	59
14 DO 15 J=3,N2	60
K=N2+1-J	61
SR(K,1)=-B(J,3)	62
ISR(K,1)=0	63
CALL SCALE (SR(K,1),ISR(K,1))	64
15 CONTINUE	65
16 RETURN	66
17 SR(1,1)=C(1)	67
ISR(1,1)=0	68
SR(2,1)=-C(2)	69
ISR(2,1)=0	70
18 CALL SCALE (SR(1,1),ISR(1,1))	71
CALL SCALE (SR(2,1),ISR(2,1))	72
GO TO 16	73
19 SR(1,1)=-C(4)	74

SR(2,1)=C(3)-C(1)	75
ISA(1,1)=0	76
ISR(2,1)=0	77
GO TO 10	78
END	79-
SUBROUTINE RECON (ROOTS,A,IA,D,M)	***** 1
DIMENSION ROOTS(2,50), D(51)	2
X=A	3
IX=IA	4
CALL UNSCALE (X,IX)	5
DO 1 I=1,N	6
D(I)=0.0	7
1 CONTINUE	8
D(N+1)=1.0	9
I=1	10
NL=N-1	11
2 IF (ROOTS(2,1)) 3,7,3	12
3 T=ROOTS(1,1)*ROOTS(1,1)	13
U=ROOTS(2,1)*ROOTS(2,1)	14
T=T+U	15
U=2.0*ROOTS(1,1)	16
DO 5 J=1,NL	17
IF (I+J-N) 5,4,4	18
4 D(J)=D(J+2)+T*D(J)	19
D(J)=D(J)-U*D(J+1)	20
5 CONTINUE	21
D(N)=T*D(N)	22
D(N)=D(N)-U*D(N+1)	23
D(N+1)=T*D(N+1)	24
I=I+2	25
6 IF (N-I) 10,2,2	26
7 DO 9 J=1,N	27
IF (J+I-N) 9,9,8	28
8 D(J)=D(J+1)-D(J)*ROOTS(1,1)	29
9 CONTINUE	30
D(N+1)=-D(N+1)*ROOTS(1,1)	31
I=I+1	32
GO TO 6	33
10 NS=N+1	34
DO 11 I=1,NS	35
D(III)=D(III)*X	36
11 CONTINUE	37
RETURN	38
END	39-
SUBROUTINE QUADIV (N,A,IA,R,D,B)	***** 1
DIMENSION A(51,3), IA(51,3), R(2), D(2), B(49)	2
B(1)=A(1,1)	3
IB=IA(1,1)	4
CALL UNSCALE (B(1),IB)	5
IF (N-2) 4,4,1	6
1 AA=A(2,1)	7
IAA=IA(2,1)	8
CALL UNSCALE (AA,IAA)	9
B(2)=AA-R(1)*D(1)	10
IF (N-3) 4,4,2	11
2 NT=N-1	12
DO 3 I=3,NT	13
XN=R(I-1)*D(1)	14
YN=B(I-2)*D(2)	15
AA=A(I,1)	16

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12=1Y	4
2 RETURN	5
3 IF (1Y) 5,4,5	6
4 Z=X	7
12=1X	8
GO TO 2	9
5 IDIFF=1X-1Y	10
IF (IDIFF) 6,7,7	11
6 1A=1Y	12
A=Y	13
B=X	14
IDIFF=-IDIFF	15
GO TO 8	16
7 1A=1X	17
A=X	18
B=Y	19
8 IF (16-IDIFF) 9,9,10	20
9 Z=A	21
12=1A	22
GO TO 2	23
10 IF (IDIFF) 11,13,11	24
11 DO 12 1=1,IDIFF	25
B=B/64.0	26
12 CONTINUE	27
13 CONTINUE	28
Z=A+B	29
12=1A	30
CALL SCALE (Z,12)	31
GO TO 2	32
END	33-
SUBROUTINE SBTR (X,1X,Y,1Y,Z,1Z)	***** 1
W=-Y	2
CALL ADD (X,1X,W,1Y,Z,1Z)	3
RETURN	4
END	5-
SUBROUTINE SCALE (X,1X)	***** 1
REC64=1.0/64.0	2
IF (1X) 1,11,2	3
1 Y=-X	4
GO TO 3	5
2 Y=X	6
3 IF (64.0-Y) 4,5,5	7
4 Y=Y/64.0	8
1X=1X+1	9
GO TO 3	10
5 IF (Y-REC64) 6,7,7	11
6 Y=Y+64.0	12
1X=1X-1	13
GO TO 5	14
7 IF (1X) 8,9,9	15
8 X=-Y	16
GO TO 10	17
9 X=Y	18
10 RETURN	19
11 1X=0	20
GO TO 10	21
END	22-
SUBROUTINE UYSCALE (X,1X)	***** 1
IF (1X+R4) 1,2,2	2
1 X=0.0	3

IX=0	4
GO TO 6	5
2 IF (IX-94) 4,4,3	6
3 X=1.0E+153	7
IX=0	8
PRINT 7	9P
GO TO 6	10
4 IF (IX) 5,6,5	11
5 X=X*64.0**IX	12
IX=0	13
6 RETURN	14
	15
7 FORMAT (25HOEXP. OVERFLOW IN UNSCALE)	16
END	17-
SUBROUTINE CHECK1 (V,AC,PP1,PP2,PP3)	***** 1
DIMENSION V(6), AC(3)	CHECK 2
V(1)=V(1)*EXP(PP1)	CHECK 3
V(2)=V(2)*EXP(-PP1)	CHECK 4
AV3=EXP(PP2)*(V(3)*COS(PP3)+V(4)*SIN(PP3))	CHECK 5
AV4=EXP(PP2)*(V(4)*COS(PP3)-V(3)*SIN(PP3))	CHECK 6
AV5=EXP(-PP2)*(V(5)*COS(PP3)+V(6)*SIN(PP3))	CHECK 7
AV6=EXP(-PP2)*(V(6)*COS(PP3)-V(5)*SIN(PP3))	CHECK 8
V(3)=AV3	CHECK 9
V(4)=AV4	CHECK 10
V(5)=AV5	CHECK 11
V(6)=AV6	CHECK 12
	CHECK 13
RETURN	CHECK 14
END	CHECK 15-
SUBROUTINE CHECK2 (V,AC,PP1,PP2,PP3)	***** 1
DIMENSION V(6), AC(3)	2
AV1=V(1)*COS(PP1)+V(2)*SIN(PP1)	3
AV2=-V(1)*SIN(PP1)+V(2)*COS(PP1)	4
V(1)=AV1	5
V(2)=AV2	6
AV3=EXP(PP2)*(V(3)*COS(PP3)+V(4)*SIN(PP3))	7
AV4=EXP(PP2)*(V(4)*COS(PP3)-V(3)*SIN(PP3))	8
AV5=EXP(-PP2)*(V(5)*COS(PP3)+V(6)*SIN(PP3))	9
AV6=EXP(-PP2)*(V(6)*COS(PP3)-V(5)*SIN(PP3))	10
V(3)=AV3	11
V(4)=AV4	12
V(5)=AV5	13
V(6)=AV6	14
RETURN	15
END	16-
SUBROUTINE CHECK4 (V,AC,PP1,PP2,PP3)	***** 1
DIMENSION V(6), AC(3)	CHK4 2
V(1)=V(1)*EXP(PP1)	CHK4 3
V(2)=V(2)*EXP(-PP1)	CHK4 4
AV3=V(3)*COS(PP2)+V(4)*SIN(PP2)	CHK4 5
AV4=-V(3)*SIN(PP2)+V(4)*COS(PP2)	CHK4 6
AV5=V(5)*COS(PP3)+V(6)*SIN(PP3)	CHK4 7
AV6=-V(5)*SIN(PP3)+V(6)*COS(PP3)	CHK4 8
V(3)=AV3	CHK4 9
V(4)=AV4	CHK4 10
V(5)=AV5	CHK4 11
V(6)=AV6	CHK4 12
RETURN	CHK4 13
END	CHK4 14-
SUBROUTINE CHECK5 (V,AC,PP1,PP2,PP3)	***** 1

DIMENSION V(6), AC(3)	CHK5 2
V(1)=V(1)*EXP(PP1)	CHK5 3
V(2)=V(2)*EXP(-PP1)	CHK5 4
V(3)=V(3)*EXP(PP2)	CHK5 5
V(4)=V(4)*EXP(-PP2)	CHK5 6
AV5=V(5)*COS(PP3)+V(6)*SIN(PP3)	CHK5 7
AV6=-V(5)*SIN(PP3)+V(6)*COS(PP3)	CHK5 8
V(5)=AV5	CHK5 9
V(6)=AV6	CHK5 10
RETURN	CHK5 11
END	CHK5 12-
SUBROUTINE RICE (ROOT,AC,KLIP)	00000 1
DIMENSION IC(6), IM(6), IRE(6)	RICE 2
DIMENSION ROOT(2,6), AC(3)	RICE 3
COMMON /KLP/ IC1,IC2,IC3,IC4,IC5,IC6,IM1,IM2,IM3,IM4,IM5,IM6,IR1,IR2,IR3,IR4,IR5,IR6	RICE 4
EQUIVALENCE (IC1,IC(1)), (IM1,IM(1)), (IR1,IRE(1))	RICE 5
WRITE (9,20)	RICE 6E
WRITE (9,21) ((ROOT(I,J),I=1,2),J=1,6)	RICE 7W
DO 1 J=1,6	RICE 8W
IC(1)=0	RICE 9
IM(1)=0	RICE 10
IRE(1)=0	RICE 11
1 CONTINUE	RICE 12
IR=3	RICE 13
IC=0	RICE 14
IM=0	RICE 15
DO 5 J=1,6	RICE 16
IF (ABS(ROOT(1,J)).LT.1.E-12) ROOT(1,J)=0.0	RICE 17
IF (ABS(ROOT(2,J)).LT.1.E-12) ROOT(2,J)=0.0	RICE 18
IF (ROOT(2,J).EQ.0.0) GO TO 4	RICE 19
IF (ROOT(1,J)) 2,3,2	RICE 20
2 IC=IC+1	RICE 21
IC(1)=J	RICE 22
GO TO 5	RICE 23
3 I=I+1	RICE 24
IM(1)=J	RICE 25
GO TO 5	RICE 26
4 IR=IR+1	RICE 27
IRE(1)=J	RICE 28
5 CONTINUE	RICE 29
WRITE (9,22) IR,11,IC	RICE 30
IF (IR.EQ.2.AND.IC.EQ.4) GO TO 6	RICE 31W
IF (IR.EQ.2.AND.IM.EQ.4) GO TO 7	RICE 32
IF (IM.EQ.6) GO TO 9	RICE 33
IF (IP.EQ.2.AND.IC.EQ.4) GO TO 16	RICE 34
IF (IR.EQ.4.AND.IM.EQ.2) GO TO 17	RICE 35
IF (IR.EQ.6) GO TO 19	RICE 36
GO TO 19	RICE 37
	RICE 38
	RICE 39
	RICE 40
6 KLIP=1	RICE 41
	RICE 42
AC(1)=ABS(ROOT(1,IR1))	RICE 43
AC(2)=ABS(ROOT(1,IC1))	RICE 44
AC(3)=ABS(ROOT(2,IC1))	RICE 45
RETURN	RICE 46
7 KLIP=2	RICE 47
	RICE 48
AC(1)=ABS(ROOT(1,IR1))	RICE 49

AC(2)=ABS(ROOT(2,IM1))	RICF 50
IF (AC(2).NE.ABS(ROOT(2,IM2))) GO TO 8	RICF 51
AC(3)=ABS(ROOT(2,IM3))	RICF 52
RETURN	RICF 53
8 AC(3)=ABS(ROOT(2,IM2))	RICF 54
RETURN	RICF 55
9 KLIP=3	RICF 56
AC(1)=ABS(ROOT(2,IM1))	RICF 57
IF (ABS(ROOT(2,IM1)).NE.ABS(ROOT(2,IM2))) GO TO 10	RICF 58
AC(2)=ABS(ROOT(2,IM3))	RICF 59
IF (ABS(ROOT(2,IM4)).NE.AC(1).AND.ABS(ROOT(2,IM4)).NE.AC(2)) GO TO 12	RICF 60
12	RICF 61
GO TO 14	RICF 62
10 AC(2)=ABS(ROOT(2,IM2))	RICF 63
IF (ABS(ROOT(2,IM3)).EQ.AC(1)) GO TO 11	RICF 64
IF (ABS(ROOT(2,IM3)).EQ.AC(2)) GO TO 13	RICF 65
AC(3)=ABS(ROOT(2,IM4))	RICF 66
RETURN	RICF 67
11 IF (ABS(ROOT(2,IM4)).EQ.AC(2)) GO TO 14	RICF 68
12 AC(3)=ABS(ROOT(2,IM4))	RICF 69
RETURN	RICF 70
13 IF (ABS(ROOT(2,IM4)).NE.AC(1)) GO TO 15	RICF 71
14 AC(3)=ABS(ROOT(2,IM5))	RICF 72
RETURN	RICF 73
15 AC(3)=ABS(ROOT(2,IM4))	RICF 74
RETURN	RICF 75
16 KLIP=4	RICF 76
AC(1)=ABS(ROOT(2,IM1))	RICF 77
AC(2)=ABS(ROOT(1,IC1))	RICF 78
AC(3)=ABS(ROOT(2,IC1))	RICF 79
RETURN	RICF 80
17 KLIP=5	RICF 81
AC(1)=ABS(ROOT(1,IR1))	RICF 82
AC(3)=ABS(ROOT(2,IM2))	RICF 83
IF (ABS(ROOT(1,IR2)).EQ.AC(1)) GO TO 18	RICF 84
AC(2)=ABS(ROOT(1,IR2))	RICF 85
RETURN	RICF 86
18 AC(2)=ABS(ROOT(1,IR3))	RICF 87
19 KLIP=1776	RICF 88
RETURN	RICF 89
20 FORMAT (10X,10H ROOTS)	RICF 90
21 FORMAT (1X,E12.5,12X,E12.5)	RICF 91
22 FORMAT (10X,3HIR=14,3X,3HII=14,3X,3HIC=14/)	RICF 92
END	RICF 93
SUBROUTINE FINGLE (EN,V)	RICF 94
DIMENSION FEN(3,3), VP(3)	RICF 95
DIMENSION DEN(3,3), EN(6,7), V(6), DUDU(6), C(3,1)	RICF 96
DO 1 I=1,3	RICF 97
DO 1 J=1,3	RICF 98
DEN(I,J)=0.0	RICF 99
1 CONTINUE	RICF 100-
DO 2 I=1,3	***** 1
	2
	3
	4
	5
	6
	7
	8
	9

```

DEN(1,1)=EN(1+3,1)
DEN(1,2)=EN(1+3,3)
DEN(1,3)=EN(1+3,4)
C(1,1)=EN(1+3,7)
2 CONTINUE
WRITE (9,9) ((DEN(I,J),J=1,3),I=1,3)
CALL LEQ (DEN,C,3,1,3,3,DET)
V(1)=C(1,1)
V(3)=C(2,1)
V(4)=C(3,1)
WRITE (9,10) V(1),V(3),V(4)
DO 3 I=1,3
DO 3 J=1,3
FEN(I,J)=0.0
3 CONTINUE
DO 4 I=1,3
FEN(I,1)=EN(I,2)
FEN(I,2)=EN(I,5)
FEN(I,3)=EN(I,6)
4 CONTINUE
C(1,1)=EN(1,7)-EN(1,1)*V(1)-EN(1,3)*V(3)-EN(1,4)*V(4)
C(2,1)=EN(2,7)-EN(2,1)*V(1)-EN(2,3)*V(3)-EN(2,4)*V(4)
C(3,1)=EN(3,7)-EN(3,1)*V(1)-EN(3,3)*V(3)-EN(3,4)*V(4)
WRITE (9,9) ((FEN(I,J),J=1,3),I=1,3)
WRITE (9,6) C(1,1),C(2,1),C(3,1)
CALL LEQ (FEN,C,3,1,3,3,DET)
V(2)=C(1,1)
V(5)=C(2,1)
V(6)=C(3,1)
WRITE (9,11) V(2),V(5),V(6)
DO 5 I=1,6
DUUU(I)=EN(1,1)*V(1)+EN(1,2)*V(2)+EN(1,3)*V(3)+EN(1,4)*V(4)+EN(1,5)
1)*V(5)+EN(1,6)*V(6)-EN(1,7)
5 CONTINUE
WRITE (9,7) (DUUU(I),I=1,6)
RETURN

6 FORMAT (1X,25H FOR THE FEN MATRIX /10X,7HC(1,1)=E12.5,5X,7HC(
12,1)=E12.5,5X,7HC(3,1)=E12.5/)
7 FORMAT (1X,36H SOLUTION CHECK BY SUBSTITUTION IS /(6E12.5))
8 FORMAT (1X,20H FEN(I,J) BY ROWS IS/(3E15.7))
9 FORMAT (1X,20H DEN(I,J) BY ROWS IS/(3E15.7))
10 FORMAT (1X,5HV(1)=E12.5,2X,5HV(3)=E12.5,2X,5HV(4)=E12.5)
11 FORMAT (1X,5HV(2)=E12.5,2X,5HV(5)=E12.5,2X,5HV(6)=E12.5)
END
SUBROUTINE LEQ (A,B,NEQS,NSOLNS,IA,IB,DET)
LINEAR EQUATIONS SOLUTIONS FORTRAN II VERSION
SOLVE A SYSTEM OF LINEAR EQUATIONS OF THE FORM AX=B BY A MODIFIED
GAUSS ELIMINATION SCHEME

NEQS = NUMBER OF EQUATIONS AND UNKNOWNNS
NSOLNS = NUMBER OF VECTOR SOLUTIONS DESIRED
IA = NUMBER OF ROWS OF A AS DEFINED BY DIMENSION STATEMENT ENTRY
IB = NUMBER OF ROWS OF B AS DEFINED BY DIMENSION STATEMENT ENTRY
ADET = DETERMINANT OF A, AFTER EXIT FROM LEQ

DIMENSION A(IA,IA), B(1B,1B)
NSIZ=NEQS
NBSIZ=NSOLNS
NORMALIZE EACH ROW BY ITS LARGEST ELEMENT. FORM PARTIAL DETERNT

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10
11
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15W
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20W
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29
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31
32
33W
34W
35
36
37
38
39W
40
41
42
43
44W
45
46
47
48
49
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51
52
53
54-
***** 1
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9
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15

```

DET=1.0	16
DO 6 I=1,NSIZ	17
BIG=A(I,1)	18
IF (NSIZ-1) 17,17,1	19
1 DO 3 J=2,NSIZ	20
IF (ABS(BIG)-ABS(A(I,J))) 2,3,3	21
2 BIG=A(I,J)	22
3 CONTINUE	23
BG=1.0/BIG	24
DO 4 J=1,NSIZ	25
4 A(I,J)=A(I,J)*BG	26
DO 5 J=1,NSIZ	27
5 B(I,J)=B(I,J)*BG	28
DET=DET*BIG	29
6 CONTINUE	30
START SYSTEM REDUCTION	31
NUMSYS=NSIZ-1	32
DO 16 I=1,NUMSYS	33
SCAN FIRST COLUMN OF CURRENT SYSTEM FOR LARGEST ELEMENT	34
CALL THE ROW CONTAINING THIS ELEMENT, ROW NBGRW	35
NN=I+1	36
BIG=A(I,1)	37
NBGRW=I	38
DO 8 J=NN,NSIZ	39
IF (ABS(BIG)-ABS(A(J,1))) 7,8,8	40
7 BIG=A(J,1)	41
NBGRW=J	42
8 CONTINUE	43
BG=1.0/BIG	44
SWAP ROW I WITH ROW NBGRW UNLESS I=NBGRW	45
IF (NBGRW-I) 9,12,9	46
SWAP A-MATRIX ROWS	47
9 DO 10 J=I,NSIZ	48
TEMP=A(NBGRW,J)	49
A(NBGRW,J)=A(I,J)	50
10 A(I,J)=TEMP	51
DET=-DET	52
SWAP B-MATRIX ROWS	53
DO 11 J=1,NSIZ	54
TEMP=B(NBGRW,J)	55
B(NBGRW,J)=B(I,J)	56
11 B(I,J)=TEMP	57
ELIMINATE UNKNOWNNS FROM FIRST COLUMN OF CURRENT SYSTEM	58
12 DO 15 K=NN,NSIZ	59
COMPUTE PIVOTAL MULTIPLIER	60
PMULT=-A(K,1)/BG	61
APPLY PMULT TO ALL COLUMNS OF THE CURRENT A-MATRIX ROW	62
DO 13 J=1,NSIZ	63
13 A(K,J)=PMULT*A(I,J)+A(K,J)	64
APPLY PMULT TO ALL COLUMNS OF MATRIX B	65
DO 14 L=1,NSIZ	66
14 B(K,L)=PMULT*B(I,L)+B(K,L)	67
15 CONTINUE	68
16 CONTINUE	69
DO BACK SUBSTITUTION	70
WITH A-MATRIX COLUMN = NCOLB	71
17 DO 22 NCOLB=1,NSIZ	72
DO FOR ROW = NROW	73
DO 21 I=1,NSIZ	74
NROW=NSIZ+1-I	75


```

SUBROUTINE KSLT (X)
COMMON DTEMP
COMMON TEMP,EA,EAC,NUA,NUAC,EB,EBC,NUB,NURC,HA,HR,R,G3A,G3B,CA,CB,RSLT
10A,OB,H,CACH,CAPCN,NUNU,GAMA,GAMB,K(40),A(55),G(7),RT(30),FITOA,FNRSLT
2TOR,RDE(60),LOAD1,AM(50),LOAD2,EN(6,7),FNXA,FNXR,FNXX,FNXX,FNX,FNRSLT
3X,FNCA,FNCB,AC(3),KLIP,ELL,DELPH(20),V(6),AL(18),WW,DWW,DDWW,WR,DWRSLT
4B,DDWR,DDDDW,WA,UBA,WU,DWU,DDWU,DDDDW,TAU,DFAU,WPJ,WWA,DWWA,NUA,DNRSLT
5UA
COMMON /SHEAR/ QA,QR,FQ
REAL NUA,NUAC,NUB,NURC,NUNU,K,LOAD1,LOAD2
WWA=WW*RT(9)
DWWA=DWW
NUA=WU*(HA/2.)*WA+(HR/2.)*WR
DWUA=DWU*(HA/2.)*WA+(HR/2.)*DWW
FNXA=CA*DWA+(CA*NUA/R)*(WWA)-RT(5)/(1.-NUA)*RT(21)
FNXB=CB*DWU+(CB*NUR/R)*(WWA)-RT(6)/(1.-NUB)*RT(22)
FNX=FNXA+FNXB
FMXA=UA*DWA-RT(12)/(1.-NUA)*RT(18)
FMXB=OB*DWU-RT(13)/(1.-NUB)*RT(19)
FMX=FMXA+FMXB+(HA/2.*HR/2.)*FNXA
QA=(HA/12.)*TAU*GAM*(WA+DWW)
QB=(HR/12.)*TAU*GAM*(WB+DWW)
FQ=QA+QB
RETURN
END
SUBROUTINE BNGO (EN,V)
DIMENSION DUMMY(6)
DIMENSION EN(6,7),REN(6,6),C(6,1),V(6)
DO 1 I=1,6
DO 1 J=1,6
REN(I,J)=EN(I,J)
C(I,1)=FN(I,7)
1 CONTINUE
CALL LEQ (REN,C,6,1,6,6,DET)
DO 2 I=1,6
V(I)=C(I,1)
2 CONTINUE
DO 3 I=1,6
DUMMY(I)=EN(I,1)*C(I,1)+EN(I,2)*C(I,2)+EN(I,3)*C(I,3)+EN(I,4)*C(I,4)+
11)*EN(I,5)*C(I,5)+EN(I,6)*C(I,6)-EN(I,7)
3 CONTINUE
WRITE (9,4)
WRITE (9,5) (DUMMY(I),I=1,6)
RETURN
4 FORMAT (40H SOLUTION CHECK BY SUBSTITUTION)
5 FORMAT (1X,40HAR(1)=E12.5,1X,40HAR(2)=E12.5,1X,40HAR(3)=E12.5,
11X,40HAR(4)=E12.5,1X,40HAR(5)=E12.5,1X,40HAR(6)=E12.5//)
END

```


APPENDIX B
Data

TABLE B.1
AVERAGE MATERIAL PROPERTIES FOR PG AND ATJ GRAPHITE

	<u>PG</u>	<u>ATJ</u>
E	3.1×10^6 psi	2.26×10^6 psi
ν	-0.21	+ 0.30
ν_c	+ 0.90	+ 0.25
R	varies with case	30 in.
L	40 in.	40 in.
α	$1.43 \times 10^{-6} \frac{\text{in}}{\text{in} - ^\circ\text{F}}$	$4.25 \times 10^{-6} \frac{\text{in}}{\text{in} - ^\circ\text{F}}$
α_c	$13.1 \times 10^{-6} \frac{\text{in}}{\text{in} - ^\circ\text{F}}$	$4.25 \times 10^{-6} \frac{\text{in}}{\text{in} - ^\circ\text{F}}$

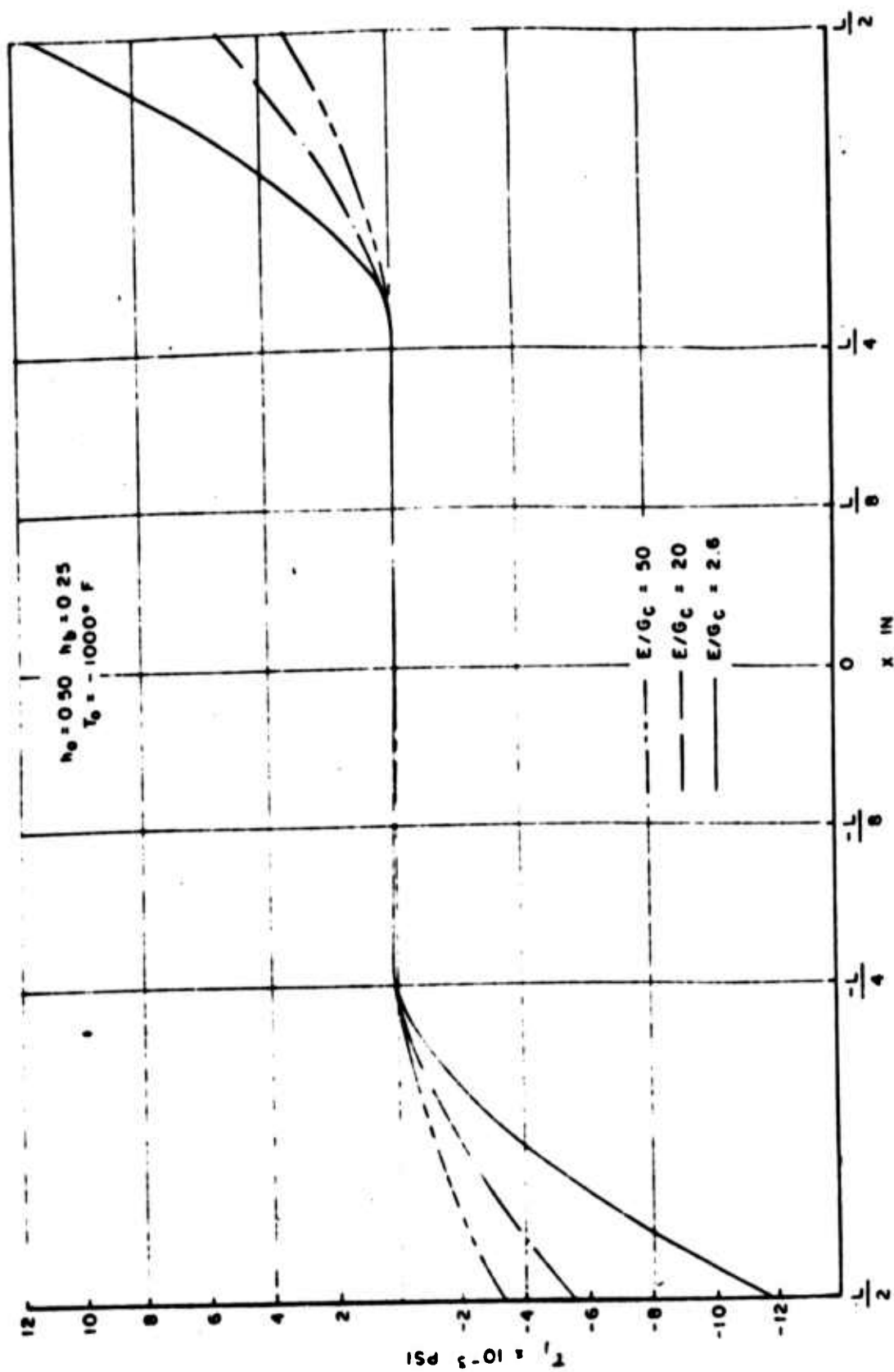


Figure B.1 Effect of E/G_c on Joint Shear Stress.

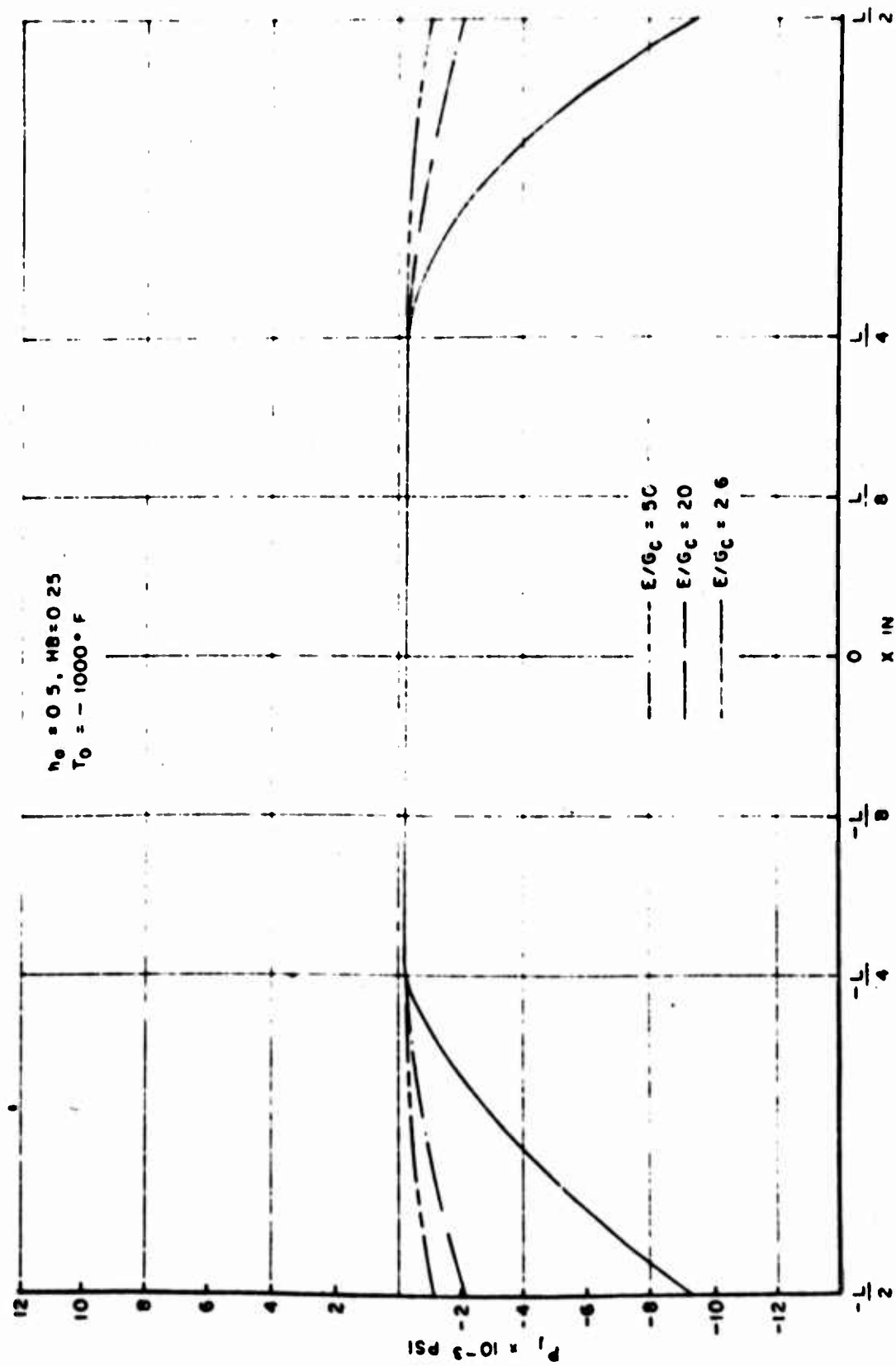


Figure B 2 Effect of E/G_c on Joint Normal Stress

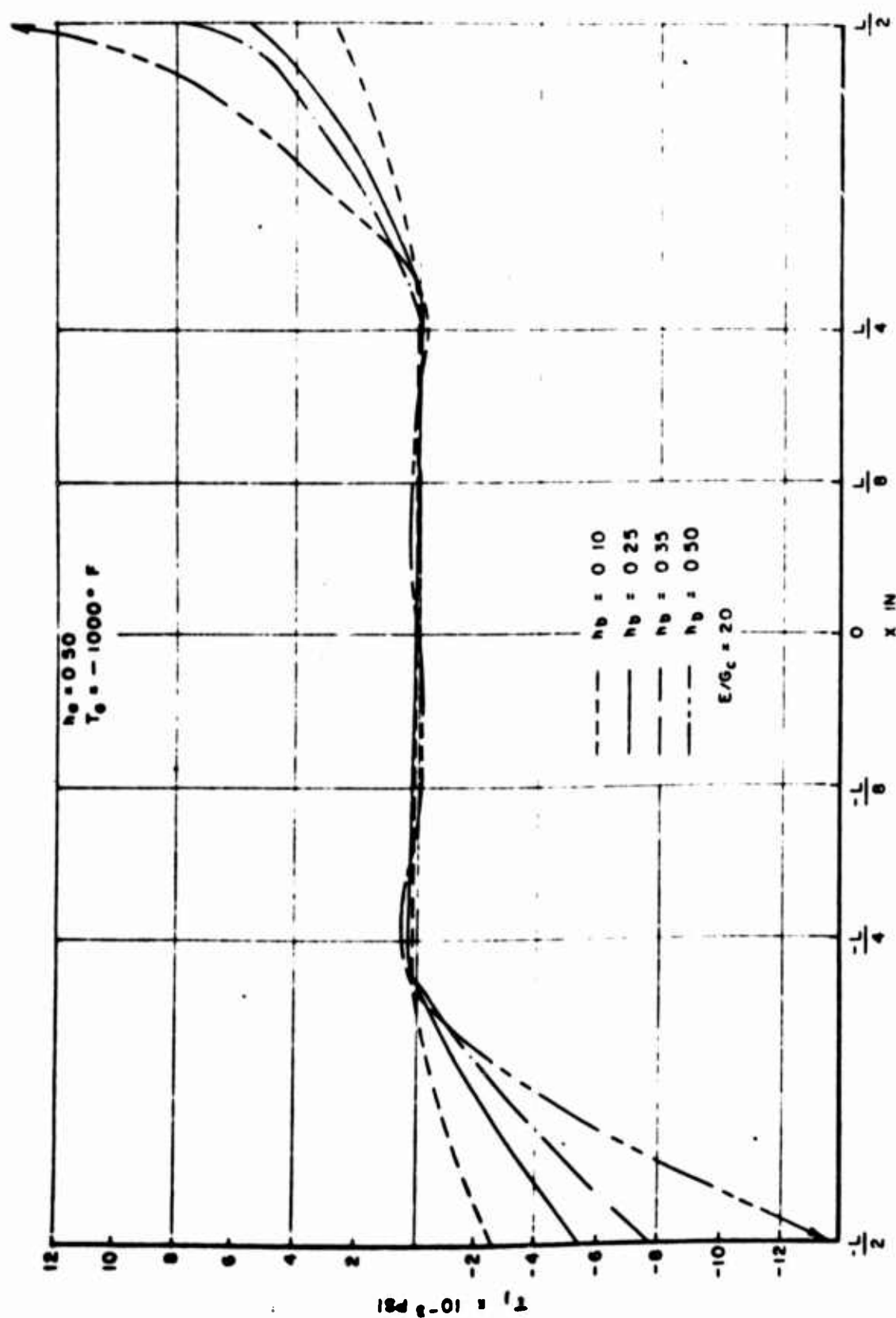


Figure B.3 Behavior of Joint Shear Stress With Varying Mandrel Thickness

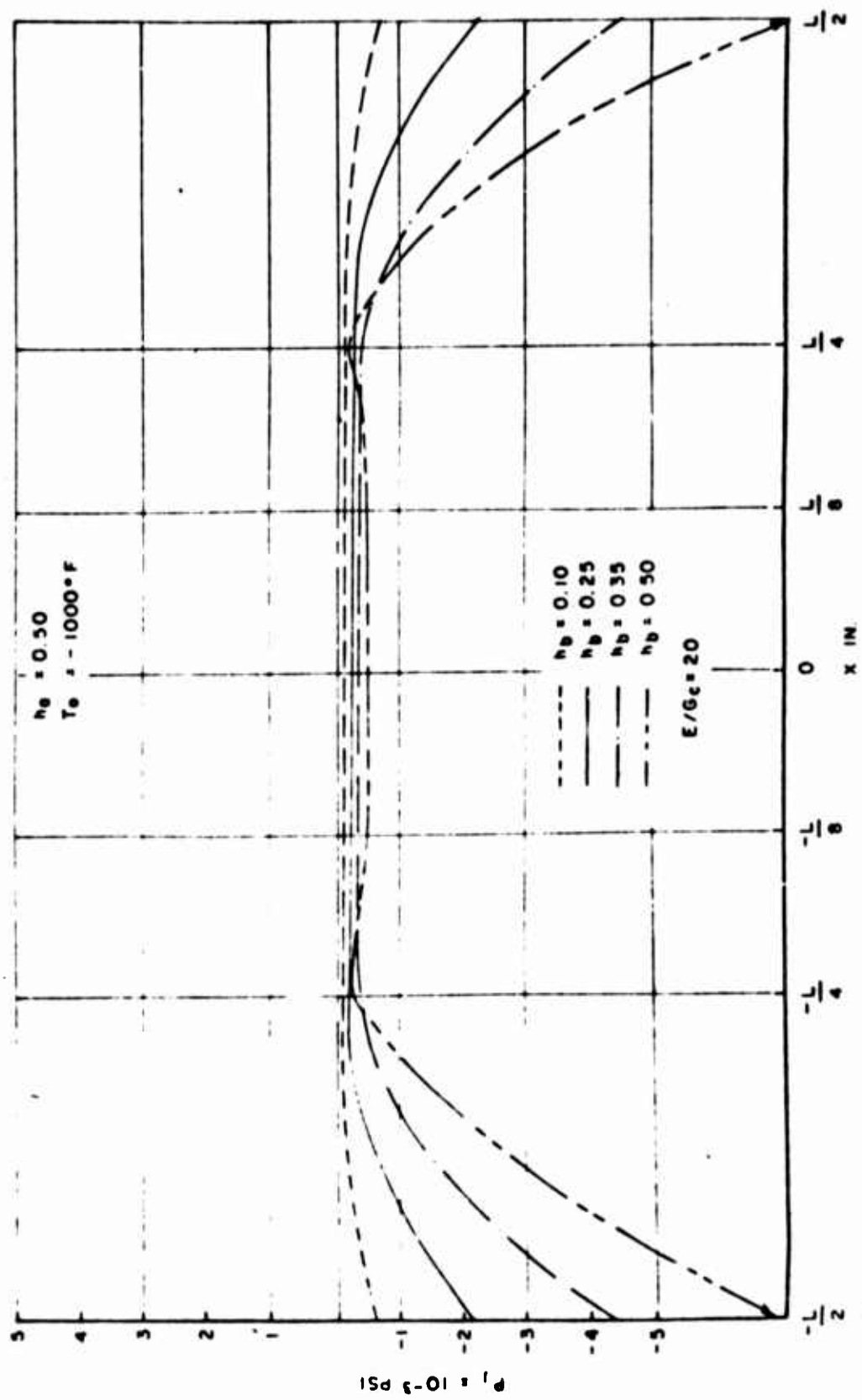


Figure B 4 Behavior of Joint Normal Stresses With Varying Mandrel Thickness

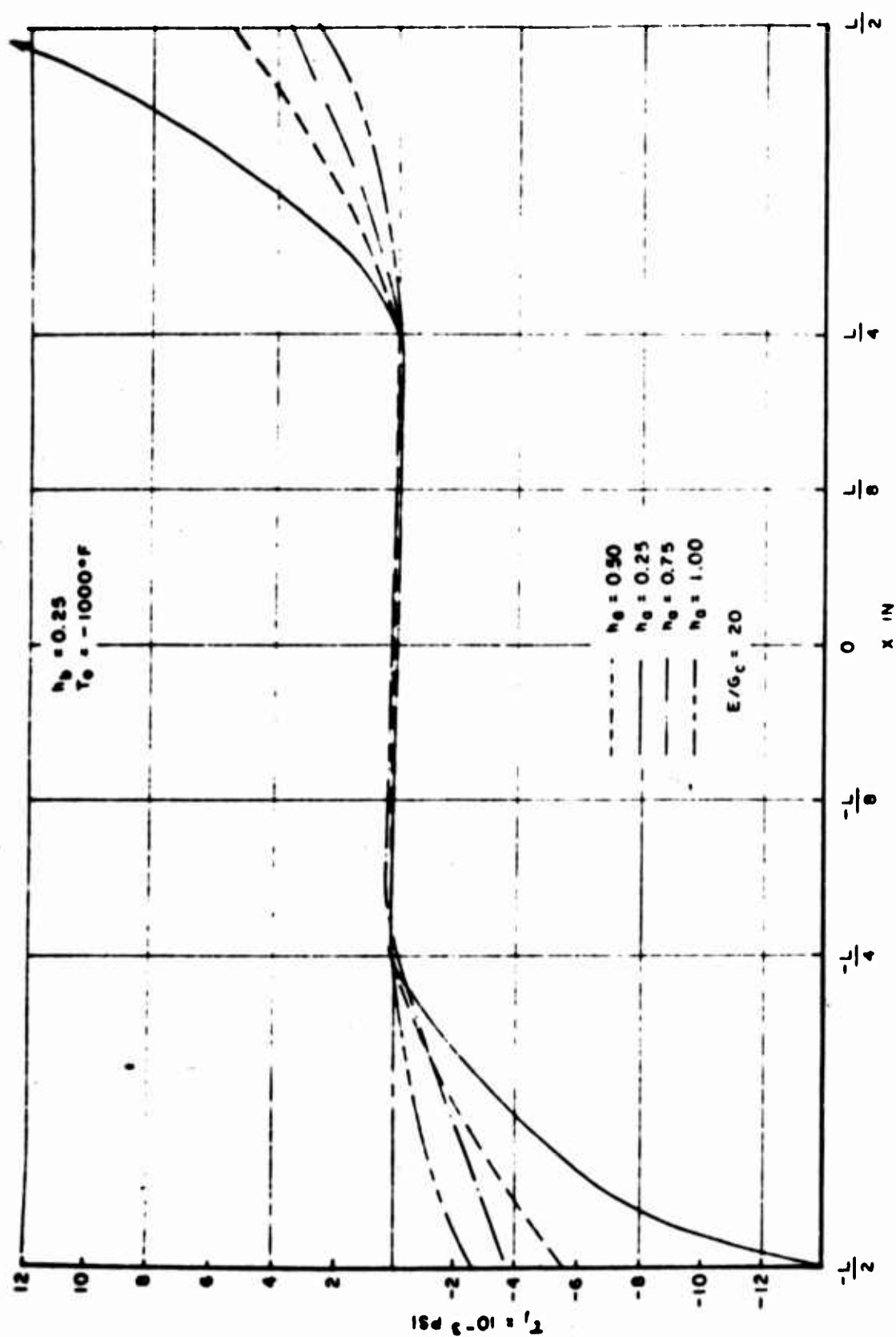


Figure B.5 Variation of Joint Shear Stresses With Increasing PG Deposition Thickness

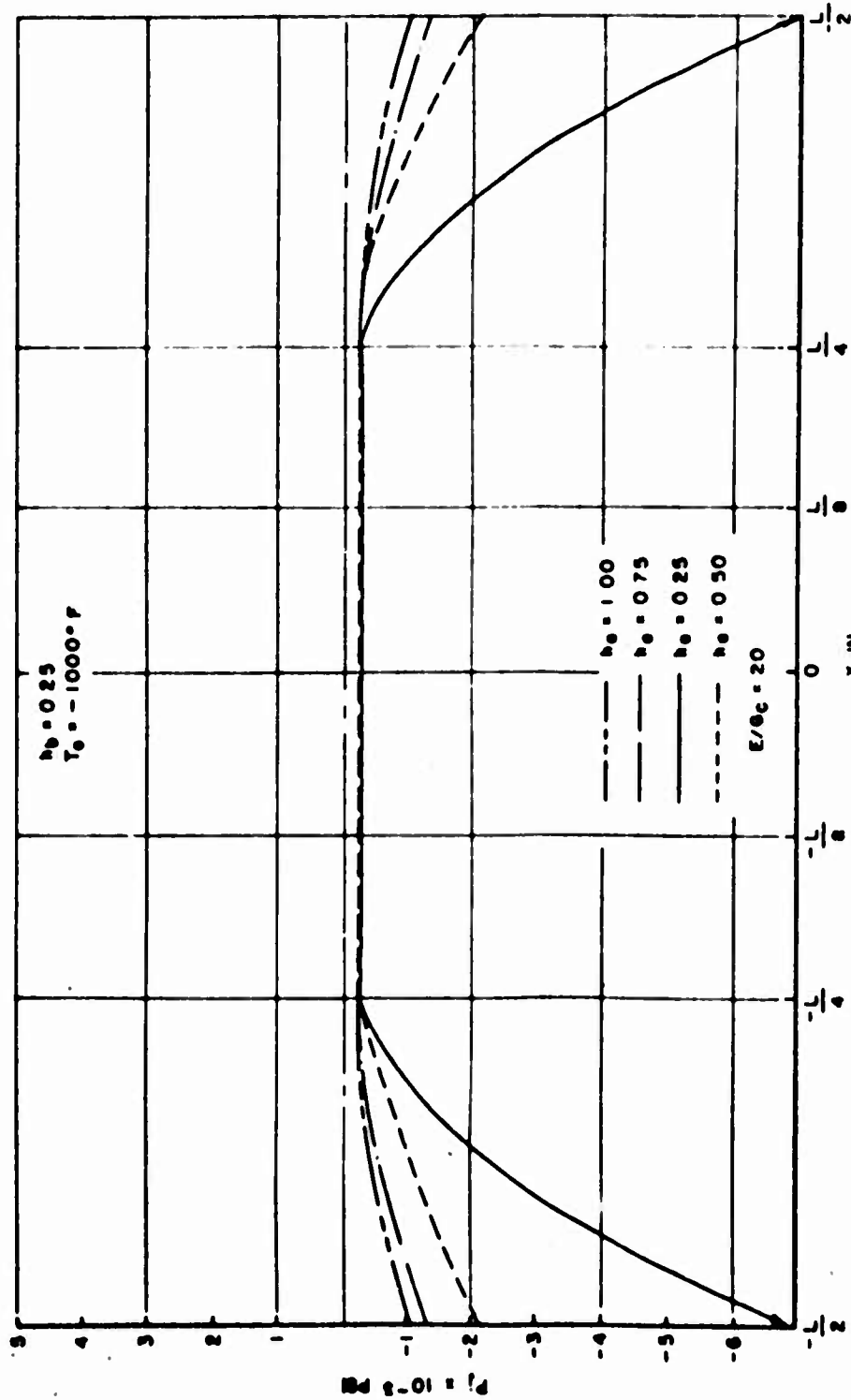


Figure B 6 Variation of Joint Normal Stresses With Increasing PG Deposition Thickness.

APPENDIX C
DEFINITION OF TERMS

$$Z_1 = (C_a + C_b) D \{(\alpha_{17} + \alpha_{27} + \pi_1 + \pi_2) - \frac{h}{2} D (\alpha_{47} + \pi_4)\} \\ + \left\{ \frac{h}{2} C_b D^2 + \left(\frac{C_a v_a + C_b v_b}{R_a R_b} \right) \right\} (\alpha_{37} + \alpha_{47} + \pi_3 + \pi_4)$$

$$Z_2 = (C_a + C_b) \{ \alpha_{67} + \pi_6 - \frac{5}{12} h_b (\alpha_{47} + \pi_4) \} \\ + \frac{5}{12} h_b C_b (\alpha_{37} + \alpha_{47} + \pi_3 + \pi_4)$$

$$Z_3 = (C_a + C_b) \{ \alpha_{57} + \pi_5 - \frac{5}{12} h_a (\alpha_{47} + \pi_4) \} \\ + \frac{5}{12} h_a C_b (\alpha_{37} + \alpha_{47} + \pi_3 + \pi_4)$$

$$Z_4 = a_{31} Z_1 - (a_{11} D^2 + a_{15}) Z_2$$

$$Z_5 = a_{23} D^2 Z_3 - (a_{31} D^2 + a_{32}) Z_2 \quad (C.1)$$

$$a_{11} = D_a (C_a + C_b) + \frac{C_a C_b h_a h}{4}$$

$$a_{12} = - \frac{C_a C_b h_a}{2} \left(\frac{v_a}{R_a} - \frac{v_b}{R_b} \right)$$

$$a_{13} = D_b (C_a + C_b) + \frac{C_a C_b h_b h}{4}$$

$$a_{14} = \frac{h_b}{h_a} a_{12}$$

$$a_{15} = \frac{C_a C_b h}{2} \left(\frac{v_a}{R_a} - \frac{v_b}{R_b} \right)$$

$$a_{16} = \left(\frac{C_a v_a + C_b v_b}{R_a R_b} \right)^2 - \left(\frac{C_a + C_b}{R_a R_b} \right)^2$$

$$a_{21} = D_b (C_a + C_b) + \frac{5}{24} C_a C_b h_b^2$$

$$a_{22} = - (C_a + C_b) \frac{5}{6} G_c^b h_b$$

$$a_{23} = \frac{5}{24} C_a C_b h_a h_b$$

$$a_{24} = \frac{5}{12} C_a C_b h_b \left(\frac{v_a}{R_a} - \frac{v_b}{R_b} \right) + (C_a + C_b) \frac{5}{6} G_c^b h_b$$

$$a_{31} = D_a (C_a + C_b) + \frac{5}{24} C_a C_b h_a^2$$

$$a_{32} = - (C_a + C_b) \frac{5}{6} G_c^a h_a$$

$$a_{33} = a_{23}$$

$$a_{34} = \frac{5}{12} C_a C_b h_a \left(\frac{v_a}{R_a} - \frac{v_b}{R_b} \right) - \frac{5}{6} G_c^a h_a (C_a + C_b) \quad (c.2)$$

$$b_{11} = a_{13}a_{23} - a_{11}a_{21}$$

$$b_{12} = a_{14}a_{23} - a_{11}a_{22} - a_{21}a_{12}$$

$$b_{13} = -a_{12}a_{22}$$

$$b_{14} = a_{15}a_{23} - a_{11}a_{24}$$

$$b_{15} = a_{16}a_{23} - a_{12}a_{24}$$

$$b_{21} = a_{23}a_{23} - a_{31}a_{21}$$

$$b_{22} = -(a_{31}a_{22} + a_{32}a_{21})$$

$$b_{23} = -a_{32}a_{22}$$

$$b_{24} = a_{23}a_{34} - a_{31}a_{24}$$

$$b_{25} = -a_{32}a_{24}$$

(C.3)

$$g_1 = \begin{matrix} b & b \\ 21 & 14 \end{matrix} - \begin{matrix} b & b \\ 11 & 24 \end{matrix}$$

$$g_2 = \begin{matrix} b & b \\ 21 & 15 \end{matrix} + \begin{matrix} b & b \\ 22 & 14 \end{matrix} - \begin{matrix} b & b \\ 12 & 24 \end{matrix} - \begin{matrix} b & b \\ 11 & 25 \end{matrix}$$

$$g_3 = \begin{matrix} b & b \\ 22 & 15 \end{matrix} + \begin{matrix} b & b \\ 23 & 14 \end{matrix} - \begin{matrix} b & b \\ 12 & 25 \end{matrix} - \begin{matrix} b & b \\ 13 & 24 \end{matrix}$$

$$g_4 = \begin{matrix} b & b \\ 23 & 15 \end{matrix} - \begin{matrix} b & b \\ 13 & 25 \end{matrix} \quad (C.4)$$

$$L_I(x) = \left(\begin{matrix} b & D^4 \\ 21 & \end{matrix} + \begin{matrix} b & D^2 \\ 22 & \end{matrix} + \begin{matrix} b \\ 23 \end{matrix} \right) \bar{z}_4 - \left(\begin{matrix} b & D^4 \\ 11 & \end{matrix} + \begin{matrix} b & D^2 \\ 12 & \end{matrix} + \begin{matrix} b \\ 13 \end{matrix} \right) \bar{z}_5$$

$$L_{II}(x) = z_4$$

$$L_{III}(x) = z_2$$

$$L_{IV}(x) = 1/(C_a + C_b) (\alpha_{37} + \alpha_{47} + \pi_3 + \pi_4)$$

$$L_V(x) = \alpha_{47} + \pi_4$$

$$L_{VI}(x) = \alpha_{27} + \pi_2 \quad (C.5)$$

$$k_1 = C_a h_a / 2(C_a + C_b)$$

$$k_2 = C_a h_b / 2 (C_a + C_b)$$

$$k_3 = 1/R (C_a v_a + C_b v_b) / (C_a + C_b)$$

$$k_4 = C_b$$

$$k_5 = k_8 = C_b v_b / R_b$$

$$k_6 = C_b h_b^2 / 12$$

$$k_7 = h_b / 2$$

$$k_9 = C_b / R_b^2 \quad (C.6)$$

$$m = 1/3 \left\{ 3 \left(\frac{g_3}{g_1} \right) - \left(\frac{g_2}{g_1} \right)^2 \right\}$$

$$n = 1/27 \left\{ 2 \left(\frac{g_2}{g_1} \right)^3 - 9 \left(\frac{g_2}{g_1} \right) \left(\frac{g_3}{g_1} \right) + 27 \left(\frac{g_4}{g_1} \right) \right\} \quad (C.7)$$

$$A = \left\{ -n/2 + n^2/4 + m^3/27 \right\}^{1/3}$$

$$B = \left\{ -n/2 - n^2/4 + m^3/27 \right\}^{1/3} \quad (C.8)$$

$$\lambda_1 = A + B - 1/3 (g_2/g_1)$$

$$\lambda_2 = +1/2 (A + B) + 1/3 (g_2/g_1) \quad (C.9)$$

$$\lambda_3 = \frac{\sqrt{3}}{2} (A - B)$$

$$r = (\lambda_2)^2 + (\lambda_3)^2$$

$$\tan \theta_1 = -\lambda_3 / \lambda_2$$

$$\tan \theta_2 = \lambda_3 / \lambda_2 \quad (C.10)$$

$$c_1 = (\lambda_1)^{1/2}$$

$$c_2 = r^{1/4} \cos (\theta_1/2) \quad (C.11)$$

$$c_3 = r^{1/4} \sin (\theta_1/2)$$

$$q = \left| \frac{n^2}{4} + \frac{m^3}{27} \right|^{1/2} \quad (C.12)$$

$$c_4 = (-n)^{1/3}$$

$$c_5 = -1/2 \left\{ c_4 + 3 (2q)^{1/3} \right\} \quad (C.13)$$

$$c_6 = -1/2 \left\{ c_4 - 3 (2q)^{1/3} \right\}$$

APPENDIX D

Derivation of Integrated Shear Stress Resultant

The required shear stress-strain relation can be developed by weighted integration in order to obtain the factor 5/6 that is generally accepted for isotropic plates and shells. The method follows that in reference (15). For convenience, the procedure for cylindrical shells is reproduced here.

Expressions for normal stress distributions with z can be obtained by replacing the strains in the stress-strain relations (1) by the approximate forms (8) and neglecting $(z/R)_i$ in comparison to unity in the same terms. It is to be understood that the following hold for each lamina.

$$\sigma_x = \frac{E}{1-\nu^2} \left(u_0' + \frac{\nu w}{R} + z\beta' \right) - \frac{E\alpha T}{1-\nu} + \frac{\nu E\bar{w}}{R(1-\nu^2)} \quad (D.1)$$

$$\sigma_\theta = \frac{E}{1-\nu^2} \left(\nu u_0' + \frac{w}{R} + \nu z\beta' \right) - \frac{E\alpha T}{1-\nu} + \frac{E\bar{w}}{R(1-\nu^2)} \quad (D.2)$$

Expressions for the stress distributions in terms of stress resultants and couples can be obtained by using equations (11) in (D.1) and (D.2).

$$\sigma_x = \frac{N_x}{h} + \frac{N_{Tx}}{(1-\nu)h} - \frac{\nu E}{Rh(1-\nu^2)} \int_{-\frac{h}{2}}^{\frac{h}{2}} \bar{w} dz + \frac{12}{h^3} z \left[M_x + \frac{M_{Tx}}{1-\nu} - \frac{E}{R(1-\nu^2)} \int_{-\frac{h}{2}}^{\frac{h}{2}} \bar{w} z dz \right] - \frac{E\alpha T}{1-\nu} + \frac{\nu E\bar{w}}{R(1-\nu^2)} \quad (D.3)$$

$$\sigma_{\theta} = \frac{N_{\theta}}{h} + \frac{N_{T\theta}}{(1-\nu)h} - \frac{E}{Rh(1-\nu^2)} \int_{-\frac{h}{2}}^{\frac{h}{2}} \bar{w} dz + \frac{12}{h^3} z \left[M_{\theta} + \frac{M_{T\theta}}{1-\nu} - \frac{E}{Rh(1-\nu^2)} \int_{-\frac{h}{2}}^{\frac{h}{2}} \bar{w} z dz \right] - \frac{E\alpha T}{1-\nu} + \frac{E}{R(1-\nu^2)} \bar{w} \quad (D.4)$$

The shear stress is related to the normal stress by the equilibrium equations (2). If the first of these is integrated with respect to z , the following is obtained.

$$\sigma_{xz} - \tau_{2i} = \frac{1}{R} \int_{-\frac{h}{2}}^z \frac{\partial}{\partial x} (R\sigma_x) dz \quad (D.5)$$

Using (D.3) in (D.5)

$$\sigma_{xz} - \tau_{2i} = -\frac{1}{Rh} \int_{-\frac{h}{2}}^z \left\{ RN'_x + \frac{12}{h^2} z RM'_x \right\} dz + \frac{1}{R} \int_{-\frac{h}{2}}^z \Omega dz \quad (D.6)$$

where

$$\Omega = \frac{R}{1-\nu} \frac{\partial}{\partial x} \left\{ \frac{N_{Tx}}{h} + \frac{12z}{h^3} M_{Tx} - E\alpha T - \frac{\nu E}{(1+\nu)R} \left[\frac{1}{h} \int_{-\frac{h}{2}}^{\frac{h}{2}} \bar{w} dz + \frac{12z}{h^3} \int_{-\frac{h}{2}}^{\frac{h}{2}} \bar{w} z dz - \bar{w} \right] \right\}$$

Referring to the integrated equilibrium equations (10) and replacing the normal stress resultants and couples in (D.5) by their

equivalents in terms of the shear stress resultant.

$$\sigma_{xz} - \tau_{2i} = \frac{1}{h} \int_{-\frac{h}{2}}^z \left[-\frac{12z}{h^2} Q + \left(1 + \frac{6z}{h}\right) \tau_{1i} - \left(1 - \frac{6z}{h}\right) \tau_{2i} \right] dz + \frac{1}{R} \int_{-\frac{h}{2}}^z \Omega dz \quad (D.8)$$

Performing the indicated integration

$$\begin{aligned} \sigma_{xz} - \tau_{2i} = & \frac{3}{2} \left[1 - \left(\frac{2z}{h} \right)^2 \right] \frac{Q}{h} - \frac{\tau_{1i}}{4} \left[1 - \frac{4z}{h} - 3 \left(\frac{2z}{h} \right)^2 \right] - \frac{\tau_{2i}}{4} \left[1 + \frac{4z}{h} \right. \\ & \left. - 3 \left(\frac{2z}{h} \right)^2 \right] + \frac{1}{R} \int_{-\frac{h}{2}}^z \Omega dz \end{aligned} \quad (D.9)$$

From (D.6) and equations (9), it can readily be shown that

$$\int_{-\frac{h}{2}}^{\frac{h}{2}} \Omega dz = 0$$

(D.10)

$$\int_{-\frac{h}{2}}^{\frac{h}{2}} \Omega z dz = 0$$

The shear stress distribution (D.9) satisfies shear stress boundary conditions and the definition of the shear stress resultant. To prove the latter it is necessary to make use of (D.10) and

$$\int_{-\frac{h}{2}}^{\frac{h}{2}} \int_{-\frac{h}{2}}^z f(y) dy dz = \int_{-\frac{h}{2}}^{\frac{h}{2}} \left(\frac{h}{2} - z\right) f(z) dz \quad (D.11)$$

McDonough (15) points out that the effect of the last term in (D.9) is to modify the classical quadratic shear stress distribution but not the magnitude of the shear stress resultant. The modification is due to the nonlinearity of the normal stress distribution, primarily its distribution with x .

Proceeding with the weighted integration as in (15), the shear stress-strain relation of the set (1) is multiplied by the weighting function $\left[1 - \left(\frac{2z}{h}\right)^2\right]$ and then integrated through the thickness of the shell to yield:

$$\int_{-\frac{h}{2}}^{\frac{h}{2}} \left[1 - \left(\frac{2z}{h}\right)^2\right] \sigma_{xz} dz = 2G_c \int_{-\frac{h}{2}}^{\frac{h}{2}} \left[1 - \left(\frac{2z}{h}\right)^2\right] G_{xz} dz \quad (D.12)$$

The integral on the left hand side is evaluated using (D.9) with the aid of (D.10) and (D.11). The integral on the right hand side is evaluated by using the shear strain given in equation (8). After integration, rearrangement and simplification, equation (11A) results:

$$Q = \frac{\bar{m}}{6} + \frac{5}{6} G_c h (\beta_i + w_i') + \frac{5}{4} \int_{-\frac{h}{2}}^{\frac{h}{2}} \left[1 - \left(\frac{2z}{h} \right)^2 \right] G_c \bar{w}' dz + \frac{5}{4R} \int_{-\frac{h}{2}}^{\frac{h}{2}}$$

$$\left(\frac{2z}{h} \right)^2 \int_{-\frac{h}{2}}^z \Omega dy dz \quad (D.13)$$

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13. ABSTRACT

A theory for the analysis of stresses in laminated circular cylindrical shells subjected to arbitrary axisymmetric mechanical and thermal loadings has been developed. This theory, specifically for use with pyrolytic graphite type materials, differs from the classical thin shell theory in that it includes the effects of transverse shear deformation and transverse isotropy, as well as thermal expansion through the shell thickness.

Solutions in several forms are developed for the governing equations. The form taken by the solution function is governed by geometric considerations. A range in which the various solution forms occur was determined numerically.

As a sample problem, the slow cooling of pyrolytic graphite deposited onto a commercial graphite mandrel was considered. Investigation of normal and shear stress behavior at the pyrolytic graphite - mandrel interface showed that these stresses decrease in magnitude with increasing E/G ratio and increasing deposit to mandrel thickness (h_a/h_b) ratio. This implies that a thin mandrel and a material weak in shear are desirable to minimize the possibilities of flaking and delamination of the pyrolytic graphite.

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